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Effects of Inlet Distortion on the Development of Secondary Flows in a Subsonic Axial Inlet Compressor Rotor

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EFFECTS OF INLET DISTORTION ON THE DEVELOPMENT

OF SECONDARY FLOWS IN A SUBSONIC AXIAL INLET

COMPRESSOR ROTOR

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Detailed flow measurements were taken inside an isolated axial compressor rotor operating subsonically near peak efficiency. Laser Anemometer measurements were made with two inlet velocity profiles. One profile consisted of an unmodified baseline flow, and the second profile was distorted by placing axisymmetric screens on the hub and shroud well upstream of the rotor. A primary flow is defined in the rotor and deviations from this primary flow for each inlet flow condition are identified. A comparison between the two flow deviations is made to assess the development of a passage vortex due to the distortion of the inlet flow. A comparison of experimental results with computational predictions from a Navier-Stokes solver showed good agreement between predicted and measured flows. Measured results indicate of a distorted inlet profile has a minimal effect on the development that the flow in the rotor passage and the resulting passage vortex.

ORIGINAL CONTAINS
COLOR ILLUSTRATIONS

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NOMENCLATURE

ANGLE Absolute Flow Angel (Figure 12) Confidence Interval C_ Specific Heat C_ d Buleev Length Scale D Jameson dissipative Term DSPD Design Speed Total Energy Probe Volume Interference Frequency fa FA, a Glow Angle, Absolute (Figure 27) FAdf, a Relative Flow Angle Differences (Figure 33) k Kinetic Energy N Number of Laser Measurements Number of Blades $N_{\mathbf{b}}$ Number of Windows per Passage N P,p Pressure Prandl Number Pr r, θ, z Cylindrical Coordinate System Radius R,r Re Reynolds Number Normal Distance from Wall, Buleev Length Scale Gas Constant Time T Temperature

- U, V, W Contravariant Velocity Components
- u,v,w Velocity components from Squire and Winter
- V_{ax} Axial Velocity (Figure 21)
- Vt. a Tangential Velocity (Figure 23)
- V_{to} Absolute Total Velocity (Figure 12)
- V_{to, r} Total Velocity, Relative (Figure 32)
- V_{θ} Tangential Velocity
- W Relative Velocity
- x,y,z Cartesian Coordinate System
- Z Desired Confidence Interval
- γ Angle of Turning
- δ Standard Second Difference Operators
- ϵ Smoothing Parameters
- R Probe Volume Laser Beams Crossing Angle
- λ Laser Beam Light Wavelength
- λ_c Courant Number
- μ Viscosity
- ν Kinematic Viscosity
- ξ,η,ζ Generalized Body Fitted Coordinates, from Chima
- ξ Streamwise Vorticity
- ρ Density
- σ Standard Deviation
- **Ω** Vorticity
- W Rotor Angular Rotation

Superscripts

- ' Relative Reference Frame, from Chima
- Vector

Subscripts

av Average

bi Biased

m Meridional Direction (Figure 19)

o Stagnation

pav Passage Average

p Primary Flow

 r, θ, Z Cylindrical Coordinate System (Figure 19)

s Secondary Flow

1,2 Inlet, Outlet Conditions

I. INTRODUCTION

Current turbomachinery design techniques are based, to a large extent, on the very large body of empirical data accumulated over the past 50 years. In a normal design procedure these data are used in conjunction with a steady state, axisymmetric through-flow calculation to produce an initial design. This design is then analyzed using a Quasi-3D or full-3D Euler solver with or without a boundary layer approximation. After this process, the design is normally fabricated and then tested, first alone and then in its operating environment. Modifications are made as a result of the tests. This cycle is then repeated, as necessary, until the machine reaches its design specifications. While this process has provided a number of excellent designs, it tends to be difficult, lengthy, and expensive. Further, since it makes extensive use of previous designs and the empirical relations derived from these results, new machines tend to be similar to previous designs, regardless of possible improvements that might result in more radical design departures. A better understanding of the flow processes occurring inside a turbomachine would not only improve the reliability of the design process but would also improve the performance of the machine. Further, the rapid growth in computational capabilities over the past decade has allowed the development of a number of fully three-dimensional viscous flow solutions. These solutions promise greater understanding of the flow physics, improvements in performance, and increased design procedure efficiency.

Detailed data of benchmark quality is essential for a better understanding of flow physics and to provide experimental verification of the advanced codes currently becoming operational.

The present study was designed to explore in detail the flow inside a compressor rotor. Specifically, it endeavored to examine certain "secondary" flows generated in the rotor. The ultimate goals were two-fold: (1) to provide detailed quality data for an improved understanding of the flow physics and (2) to explore the effects of inlet distortion on the development of a compressor rotor flow field.

The term "Secondary Flows" encompasses many of the flow processes inside a compressor rotor that result in lowered efficiencies and range. These flows can be exceedingly complex and difficult to analyze. Secondary flow is usually defined as the difference between the actual flow and the idealized axisymmetric flow. Wood [1] includes cascade secondary flows, scraping effects, tip leakage flows, and radial flows in his definition of secondary flow. These flows are shown schematically in Fig. 1.

The primary focus of this research was the cascade secondary flow or the "Passage Vortex." This type of flow results when a nonuniform inlet velocity profile is turned by a row of blades. This can be visualized by considering the simple example of flow through the bend of a rectangular channel, as shown in Fig. 2. For this example, it has been assumed that the flow is incompressible and that viscous effects can be neglected. Further, consider an inlet velocity profile that varies in the spanwise direction. This variation ranges from some low

velocity near the endwalls to freestream velocity near midspan. If the distance from an endwall to where the velocity approximates the freestream value is of the order of magnitude of the cross channel distance or less, the freestream static pressure will be nearly the same as the static pressure in the region of lower velocity. Simply put, the static pressure is nearly constant across the passage inlet. This type of velocity profile is common at the inlets of the middle stages of axial compressors. It results from the increasing thickness of the endwall boundary layers and the gradual convergence of the passage endwalls. However, it is not important here how this velocity deficit occurs. This profile will be influenced by the blade-to-blade pressure gradient that results from the turning of the flow as it moves through the passage. There will then be a pressure gradient that exists between the suction and pressure surfaces. This pressure gradient can closely be approximated by [2]:

$$\frac{\partial P}{\partial y} = \frac{\rho V}{R} \tag{1}$$

Equation 1 demonstrates that any reduction in the fluid velocity,

V, must be matched by an equivalent reduction in the radius of curvature

of the flow. This results in the generation of the cascade secondary

flow shown in Figs. 1 and 2.

The passage vortex, then, is caused by an axisymmetric distortion of the inlet total pressure profile. It is most often caused by endwall boundary layers, which can be quite substantial in the middle and aft stages of an axial flow compressor. To date, this effect of inlet

distortion on the middle stage compressor rotors has only been approximated by empirical relations.

This thesis presents the results of an experimental study of the flow phenomena resulting from the presence of these flow distortions on a transonic inlet compressor rotor operating subsonically at 60% of design speed.

Two operating conditions were observed. One was a "uniform" or baseline inlet velocity profile with no unusually thick boundary layers. The other flow condition was at the same operating point with artificially enhanced endwall boundary layers. These boundary layers were generated upstream of the rotor using screens that extended partially into the inlet flow passage. The differences in the generated velocity profiles are indicative of the size and structure of the passage vortex that was generated by the inlet flow distortion.

These two flow conditions were compared and their differences were calculated. These results were compared with a state-of-the-art three-dimensional Navier-Stokes solver. This solver includes endwall and blade boundary layer effects and has been used at the NASA Lewis Research Center to accurately model the velocity fields of several compressor designs. The LDV data acquired during this experiment and its follow on program should greatly aid in the validation of this and other codes for analyzing turbomachinery fluid flows.

II. LITERATURE SURVEY

A. INTRODUCTION

Research accomplished in the area of secondary flow or even just the "Passage Vortex" is quite extensive and it would be impractical, if not impossible, to adequately describe all of the previous work in this area. Therefore, the following section will merely highlight work in the specific area of this study.

The analytical/numerical modeling of the passage vortex will be reviewed first and then the experimental work done will be considered.

Literature on secondary flows dates back at least to 1877 when J.

Thompson [3] published an article discussing the flow of water in a river bend. Fifty years later, W. R. Dean [4] published a study of the secondary flows generated in laminar pipe flows.

B. ANALYTICAL/NUMERICAL WORK

The first "modern" publication in the area was the now classic 1949 paper by Squire and Winter [5]. They analyzed the secondary flow generated in an inviscid fluid with a nonuniform inlet profile moving through a cascade channel. Fig. 3 shows the channel they analyzed.

This early effort was very narrow in scope and was certainly restricted by the lack of computational power. Accordingly, a number of simplifications were made which included:

(1) a nonrotating reference frame,

- (2) a perfect fluid,
- (3) steady flow,
- (4) no body forces,
- (5) deflection without expansion or contraction,
- (6) passage width small compared to the radius of turn (a<<R_{bo}, Fig. 3),
- (7) small approach-flow velocity non-uniformity, and
- (8) two-dimensional approach flow, i.e., the approach velocity varies only in the direction of the blade span (b).

The equations of motion:

$$\nabla \times (\nabla \times \overline{\Omega}) = 0 \tag{2}$$

along with the conservation of vorticity:

$$\nabla \cdot \mathbf{\Omega} = 0 \tag{3}$$

were used to develop a simplified description of the downstream streamwise vorticity:

$$\xi_2 = \frac{\partial w_2}{\partial \beta_2} - \frac{\partial v_2}{\partial z_2} = -2\gamma \frac{\partial U_1}{\partial z_1} \tag{4}$$

W. R. Hawthorne has published numerous papers on the subject of secondary flow. He derived a general equation for inviscid, incompressible flow in a cascade [6]. Beginning with Eqs. (2) and (3), and continuity, Hawthorne showed that:

$$(\xi/V)_2 - (\xi/V)_1 = 2\int_1^2 |\nabla P_o| \sin\gamma \frac{d\delta}{\rho V^2}$$
 (5)

This equation does not, as written, require the Bernoulli surface distortion to be small or a uniform spanwise pressure distribution. Further, it allows velocity variations in other than the spanwise direction.

Both of these papers approached the problem of the passage vortex by considering it as a phenomenon developing in a fluid flowing in a closed channel. This "Channel Theory" is not, however, the only way to look at this phenomenon. It can be viewed as the result of flow over an airfoil. In 1951, Lieblien and Ackley [2] took this approach when they analyzed the secondary flows in cascades and inlet guide vanes in this manner.

L. H. Smith must also be mentioned in this section for the seminal work he did using the airfoil approach to the problem. In 1955, he published two papers [7, 8] which recommended a definite method for accounting for the secondary velocities generated by the passage vortex.

These early publications set the direction for analytical work for the next twenty five years. Secondary flows were assumed small and then were "decoupled" from the main through-flow. Much was done in this direction.

These concepts were broadened and generalized in the 1960's and 1970's and culminated in several numerical algorithms for accounting for secondary flow effects [9-11]. During this same time frame, another

more general approach to describing flows in a compressor rotor passage was developing. This approach was based on the solution of the equations of motion on the digital computer.

The ultimate goal for any analytical formulation is, of course, the solution of the complete Navier-Stokes Equations. However, even with the current capabilities available, the solution of these equations can be extremely time consuming for even relatively simple geometries such as turning ducts. Thus, the initial use of digital computers was to solve simple models of flow through compressors.

Although two-dimensional blade-to-blade solutions were being published in the late 1940's [12], perhaps the major event paving the way for the analysis of turbomachinery flow consisted of the publishing of a series of papers by Wu [13-17]. In these papers, Wu developed a "General Theory" where the stream function equations are solved on two intersecting families of stream surfaces; a blade-to-blade stream surface and a hub-to-shroud stream surface (Fig. 4). He postulated a procedure where the stream functions are solved iteratively on each series of surfaces until a converged solution is obtained. A detailed derivation of the stream function equations used in Wu's analysis may be found in Ref. [15].

There have been numerous attempts to use Wu's stream function surfaces to solve for the flow in turbomachines. They are, however, two-dimensional solutions and cannot, directly, provide information on the various secondary flow phenomena that occur in turbomachinery.

To begin to effectively explore the development of secondary flows

computationally, a three-dimensional solution is required. In 1969, MacCormack [18] published a three-dimensional Euler solution that provided second order accuracy in both time and space. Thomkins [19] used the MacCormack algorithm in a full three-dimensional code. This method can be applied to any general compressor blade shape.

Denton [20-22] has written Euler codes using the control volume approach with quadrilateral elements in two dimensions and six sided elements in three-dimensions. This approach results in simpler expressions for surface fluxes. He has improved convergence by using "multigridding" techniques where the flux balances are applied over large grids concurrently with the finer mesh. Denton uses upwind differencing in the streamwise direction for the mass and momentum flux with downwind differencing for pressure. Central differencing is used for all quantities in the pitchwise and spanwise directions.

Recently, Denton and Wood [23] have added a simple one-dimensional boundary layer calculation to the original code.

A great deal of effort is currently being directed towards the development of fully three-dimensional solutions to the Navier-Stokes equations.

An operational code developed by Hah [24-26] is currently being used at the Lewis Research Center to analyze the flow field inside turbomachinery.

Just becoming available at the center is a Navier-Stokes solver developed by Chima [27] for use in the analysis of turbomachinery flow fields. This code represents an operational 3-D Navier-Stokes solver

for use in turbomachinery and has been used in the current effort to evaluated the data acquired and to aid in accurately describing the flow phenomena observed. This code will be discussed in more detail in a later section (Section III, part F).

C. EXPERIMENTAL WORK

There are two broad types of turbomachines: compressors and turbines. The differing purposes of each, of course, result in greatly differing designs. Both of these broad classifications can be further divided, for experimental purposes, into three subdivisions of increasing complexity and difficulty: (1) two-dimensional cascades, (2) annular cascades, stators, and inlet guide vanes, and (3) rotors. While all three areas have contributed to the understanding of secondary flows, the great extent of the published work requires this survey to be largely restricted to the last of these subcategories.

Smith [8] examined the secondary flows that were generated inside an axial compressor stage consisting of inlet guide vanes, a rotor and a stator. While he did not specifically look at the effects of a variable boundary layer thickness, he did make detailed full span measurements through both the hub and shroud boundary layers and, for the inlet guide vanes and stator, across the pitch. Unfortunately, since only "conventional" instrumentation was available, he obtained only pitchwise average information behind the rotor. No information was gathered

between the blades of either the stator or the inlet guide vanes. Smith unsuccessfully attempted to obtain secondary flow information on the rotor by attaching tufts of wool to the trailing edge of the blades and observing them with a stroboscopic light. Regrettably, centrifugal effects nullified this attempt. Commenting on the data obtained, he reflects that "in view of the experimental difficulties involved, the coarseness of the data, and the sensitivity of the method to small changes, we can only say there appears to be no conflict between the calculated and measured values. Thus, no clear-cut experimental verification of the secondary flow theories has been made."

In the 1950's, turbomachinery development efforts began in earnest at several research organizations. Most notably, the engine companies such as General Electric at Evandale and Pratt and Whitney at Hartford (UTRC) and research institutes such as Vrije Universiteit Brussel, VKI, Pennsylvania State University, and Escole Centrale de Lyon began to obtain detailed measurements of the flow fields generated in axial rotors.

One set of extensive data on the development of secondary flow was published by McKensie [28] at Rolls Royce in England. Dixon and Horlock [29] used the experimental data generated by McKensie to test the accuracy of their method of predicting secondary flow development.

McKensie's compressor consisted of four axial stages. Dixon and Horlock show boundary layers which occupy roughly one-half the passage. The information presented consists entirely of pitch averaged axial velocities and flow angles at the exits of the inlet guide vanes and the

first stage rotor exit. They concluded that while their technique for calculating secondary flow development works well behind the inlet guide vanes, the accuracy of the technique is greatly impaired by the boundary layer growth through the rotor. They do propose, however, that their calculations would be useful for predictions in an "ultimate steady flow" occurring downstream of a number of compressor stages.

De Ruyck, Hirsch, and Kool [30-32] published several papers in the late 1970's showing the three-dimensional flow behind a relatively low speed compressor rotor and comparing it to A. G. Smith's [33] theoretical treatment of secondary flow development. As such, they assume the correctness of Smith's assumptions. Their data were obtained with a hot wire probe placed behind the rotor. In their analysis, they assume a primary flow in the direction of the blade angles and ignore the tip leakage. The agreement between their hot wire measurements and Smith's predictions was, at best, not very good. However, when they included trailing shed and trailing filament vortices, they were able to obtain "qualitive agreement." They concluded that the trailing shed and trailing filament vortices must be included in the analysis and that tip leakage effects must also be considered.

The General Electric Company has supported an extensive experimental program at their Low Speed Research Compressor Facility (LSRC) in Evandale, Ohio. This low speed compressor has a 1.52M tip diameter, a 1.07M hub diameter and can operate at a tip speed of over 60 m/s.

Smith's 1970 presentation [34] at a symposium on blading research contributed extensive pitchwise-average data on 10 separate compressor configurations tested in the LSRC facility. These different "builds" each consisted of four identical stages. The program varied spacing, clearance height, gap, and pitch. Smith concluded that there was an "ultimate steady flow" profile that developed in a multistage machine and that this pitchwise-average flow could be analyzed in a manner similar to that suggested by Mellor and Strong [35], among numerous other authors.

Smith, along with Adkins [36] and Koch [37] continued to publish results from the General Electric LSRC at Evandale, Ohio. Smith and Adkins paper represents a rather extensive effort to analyze all of the secondary flows occurring in rotors. They present only pitch-averaged spanwise data taken for two different configurations at General Electric They also analyze data taken by Larguier [38] at ONERA, consisting of spanwise averaged pitchwise exit flow angles. These analyses are based on Smith's early work at Johns Hopkins.

The LSRC data showed relatively small secondary flow fields primarily because of the high blade aspect ratios (2.8) and it should be noted that the data was acquired primarily to study tip clearance flows.

Smith and Adkins concluded:

 that secondary flows are important to the design of axial flow compressors,

- (2) that for compressors operating at the design point, secondary flows can be adequately calculated using inviscid analysis, and
- (3) that downstream mixing processes are quite complex and current modeling techniques are quite crude and empirical.

Work at General Electric has continued in the 1980's on the "ultimate steady flow" concept with Wisler et al. [39] publishing an important paper attempting to clarify the relative importance of classic secondary flow with respect to turbulent diffusion in the spanwise mixing in axial compressors.

This problem was considered in earlier papers by Gallimore [40] and Gallimore and Cumpsty [41]. This work, done at the Whittle Lab in England indicated that turbulent diffusion was the dominant mixing mechanism in axial flow compressors. These results were based on observations of two four stage axial flow compressors. A tracer gas technique [42] was used where ethylene was injected upstream of the first stage rotor and its density was sampled downstream of the fourth stage stator.

Wisler et al.'s work made use of the same gas tracing technique, in conjunction with hot wire anemometry, in the LSRC at Evandale. These extensive measurements included ethylene injection in front of the third stage stator and through static taps on both the suction and pressure surface of the third stage stator. The trace gas was sampled downstream of the third stage stator and the fourth stage rotor. Cross channel

plots were presented showing the spread of the trace gas with respect to the third stage stator that indicate that "in addition to being mixed by turbulent diffusion, the low momentum fluid in the endwall is transported radially along the stator vane edges by secondary flow. The passing rotor then chops, turns and transports this low momentum fluid, spreading it circumferentially across the stator passage" [41].

Note that both of these programs looked at "ultimate steady flow" in compressor stages. Here both rotor and stator effects were considered and information was gathered between compressor components.

Lebeouf and Papailiou et al. [43-45] have published a number of reports on the secondary flows in cascades and compressor rotors. This work was performed on a transonic axial compressor rotor and was conducted at the high speed compressor test facility at the Ecole Centrale de Lyon. Their 1977 and 1978 papers describe the results of tests conducted using a SNECMA compressor. In these studies, the flow angles, static and total pressures and temperatures upstream and downstream of the rotor were measured. A 1977 report describes extensive pitchwise average data at these axial locations for two operating conditions: "nominal" and near surge. In this report the experimental results were compared with a simple radial equilibrium analysis and the following conclusions were reached:

(1) that data reduction using radial equilibrium will require the use of curvature terms, and

(2) that difficulties exist when an analytical representation of the velocity profiles is sought in both the absolute and the relative systems of reference.

It was determined in the report that these conclusions favor a differential formulation of the endwall boundary layers.

In a later 1982 work, Lebeouf and Papailiou et al. [45] dealt with a single stage axial transonic compressor consisting of inlet guide vanes, a rotor, and a stator. The theoretical analysis used was based on a Quasi-3D-inviscid computation using a method described by Katsanis [46] for the blade-to-blade solution and a meridional solution by Lebeouf combined with a viscous endwall solution based on the work of Mellor and Wood [47].

Data was taken for this report using conventional instrumentation.

Circumferential radial surveys at two axial locations between the inlet guide vanes and the rotors and pitch averaged radial surveys at two axial locations behind the rotor were all acquired.

In comparing the results behind the rotor with theoretical computations, the report notes that, between 20% and 85% of the blade height, the prediction of average total pressure losses and outlet flow angles are "well reproduced." The secondary flow (endwall boundary layer) model presented in the report reproduces the hub β angles well. The theoretical reproduction of the flow in the tip area is less exact because the model fails to account for tip clearance and moving wall

effects. Small discrepancies in total temperature near the hub were also highlighted in the paper.

The conclusion section states that "in spite of the usefulness of global methods for the computation of secondary flows, (such as the one developed by Mellor and Wood), it is believed that the use of detailed flow computation, ---, may result to improvements in the efficiency of modern machines through the gain obtained in better understanding the flow behaviour."

Under the guidance of Lakshiminarayana, an extensive program exploring, both experimentally and computationally, turbomachinery flows is being conducted at the Pennsylvania State University. The experimental work is being done in a low speed axial flow compressor [48] to include work in both rotors and stators. The work in rotors has examined not only general internal flows [49] but has also studied wake structures [46-49], annulus boundary layers [54] and tip clearance effects.

Lakshiminarayana and his associates have published well over one hundred papers on internal turbomachinery flows. A number of articles, however, are particularly pertinent to the current research. For example, the team at Pennsylvania State University placed conventional pressure probes and hot wire anemometers on the hub of a compressor rotor. As a result, they were able to measure the velocities and turbulence levels of the wake behind [51-53] and the pressures [55] inside this lightly loaded, low speed compressor. The results were compared with the predictions on Katsanis and McNally [56-57], obtaining

"very good" agreement with the exception of the region near the tip of the rotor. The report theorizes that this may be the result of tip clearance effects. This data, taken for one inlet velocity profile at one operating condition covers the region from 10 to 90% pitch at six axial locations and five radial locations.

Some of the conclusions that resulted from these tests are:

- (1) in the inviscid regions of the rotor passage, the radial velocity is small;
- (2) relative velocity develops from a linear variation from blade-to-blade at the inlet to nearly uniform at the exit;
- (3) "metal blockage" from the blades has an appreciable effect on the axial velocity;
- (4) passage averaged relative total pressure remains constant with respect to the axial distance in the inviscid regions of the blade passage;
- (5) static pressure varies appreciably in the blade wake;
- (6) with the exception of the wake region, all properties are uniform at the exit: and
- (7) the quasi 3D predictions of Katsanis and McNally [56-58] are very good except near the blade leading edges and blade tip regions.

The United Technologies Research Center, a Division of Pratt and Whitney, has maintained a strong experimental program studying the flows within axial compressors.

Their test facility, the "Large Scale Rotating Rig" (LSRR) is 1.52 m in diameter. The rotating gear speed is roughly 500 RPM which gives an operating Reynolds number of approximately 5x10⁵.

In two separate papers, Dring, Josyln, and Hardin
[59, 60] detailed what they considered a "benchmark" study of the
nonstrip theory (secondary flow) effects in a compressor rotor. Data
were obtained inside the rotor using flow visualization techniques and
static pressure taps. Also acquired were detailed three-dimensional
pressure and angle distributions behind almost two full passages of a
single stage rotor. The inlet conditions in this study were uniform
with the exception of the thin inlet endwall boundary layers.

All velocity components, C_{ps} , C_{pt} and angle distribution plots at four axial positions aft of the blade are presented. Data is provided at four flow settings:

axial velocity/mean tip speed = 0.65, 0.75, 0.85, 0.95.

The overall conclusion from these reports was that "radial flow effects (on the blade surface) can have a major impact on the nature of the wake of a compressor rotor." It was noted that the accumulation of low energy fluid in the suction surface corner is inhibited by radial flow effects. Thus, corner stall can be inhibited. Further, mid-span performance closely models that of a two-dimensional cascade [61].

Later, Wagner et al. [62] specifically studied the effects of a nonuniform inlet profile on the performance of a rotor identical to the machine described in the previous report. A mechanical device was able to generate a thickened boundary layer on both the hub and the shroud of

about 37% of the span. Then the rotor was surveyed at the same operating points that were studied in the previous work, i.e., the same four flow coefficients.

In the summary of results the following observations were noted:

- (1) increased loss in the midspan region and decreased loss near the hub and tip regions relative to previous results,
- (2) a general increase in radial displacement of the flow with increased loading,
- (3) a rise of static pressure compared with the previous flow,
- (4) only a very small rotation of the flow due to the relative eddy,
- (5) increased tip losses and increased hub and tip blockage occurring with increased loading, and
- (6) exit relative flow angles were only "weakly" affected by the inlet profile.

These, then, are the major experimental programs in the area of secondary flows in compressors. The important features of each can be summarized as follows:

(1) Researchers at General Electric, have attempted to create a comprehensive system to account for secondary flow effects. This work has been greatly concerned with tip clearance effects. They have developed an "ultimate steady state" model to help describe the flow in axial compressors. They consider secondary flows important media for spanwise fluid transport in compressors.

- (2) LeBeouf et al. have worked primarily with a transonic compressor. The data is pitch averaged and taken behind the rotor.
- (3) Lakshiminarayana et al. have used hot wires and conventional pitot static instrumentation in a low speed compressor.

 Their work includes secondary flow measurements inside the rotor passage. The overall program has looked at all potential secondary flow effects.
- (4) Dring et al. have specifically examined the effects of a nonuniform inlet profile and the effects it has on the generation of the passage vortex and other flow effects. Their work was performed on a low speed compressor rotor. They obtained tangential surveys by means of a rotating pitot static instrumentation system positioned behind the rotor.

A summary of the performance parameters and measurement capabilities of the various facilities previously discussed is presented in Table I.

III. APPARATUS AND FACILITIES

A. INTRODUCTION

There are five broad areas that will be presented in this section:

- (A) the Laser Anemometer system which includes the optics, flow seeding, and probe volume positioning system.
- (B) the data Acquisition/Reduction system which includes the conventional pitot static instrumentation.
 - (C) the test facility.
 - (D) the compressor rotor, its' design and specifications.
- (E) and the Computational Development for the Navier-Stokes Solver.

Previous publications have discussed, to some extent, all of the areas listed above and these previous publications will be noted where necessary.

B. THE LASER ANEMOMETER SYSTEM

1. Optical System

The Laser Anemometer measures the velocity of a fluid by measuring the velocities of particles immersed in that fluid, the assumption being that the particles observed are moving at a velocity near that of the fluid. The Laser Anemometer makes use of the coherence, monochromicity, and nondivergence characteristics of the typical laser beam. It makes its measurements nonintrusively and has become, in the last decade, one of the preferred methods of studying turbomachinery flows.

There are several types of laser anemometer systems. The system used at the Lewis Research Center's single stage axial flow compressor facility is a "fringe type" system and was developed in-house.

There are a number of excellent publications [63-65] describing the optical operation of the laser-fringe anemometer (laser doppler velocimeter). Simply put, the system operates by measuring the heterodyne frequency of light reflected (or emitted) from a particle illuminated simultaneously by two incident laser beams.

The fringe type system splits a single laser beam into two beams. These two beams are focused and cross at a desired point in the flow-called the "Probe Volume." Using the fringe model for explanation, it is said that in the region of beam intersection, a pattern of constructive and destructive interference is created (Fig. 5). A particle in the flow traversing the probe volume will alternately be illuminated or not, depending on the region of the probe volume it is in. This intermittent illumination of the particle can be observed using a photomultiplier tube (PMT). This explanation of the operation of a laser anemometer is the "fringe model."

The PMT signal takes the form of a sine wave of varying amplitude.

The frequency of this sine wave can be related to the measured velocity

of the particle using the relation [65]:

$$V = \frac{f_D \lambda}{2 \times \sin \kappa} \tag{6}$$

If the particle is small, generally less than a micron in diameter, the particle velocity and the fluid velocity can be assumed to be the same.

Particles observed in this investigation were measured and were virtually all less than one half micron in diameter. This size is considered small enough to accurately reflect the velocities of the fluid [64, 66].

Of critical importance here is the obvious fact that the intersection of these two beams results in the measurement of only one component of velocity (Fig. 5). This component lies along a line formed by the intersection of two planes. One of these is the plane in which the two incoming laser beams lie. The other plane is perpendicular to the line bisecting these two beams and is in the probe volume. Figure 5 illustrates the probe volume and the measured velocity component.

During the normal operation of the laser system the incoming laser beams are oriented in two directions about the bisecting line. This allows the resolution of two of the three orthogonal velocity components. In turbomachnes, these two components are usually the axial (through-flow) and the tangential components. Normally the radial components are considered small and are not resolved.

The laser anemometer system used currently at the Lewis Research Center has been highly modified from the system described in Refs. [66-68]. It remains a fringe type, on axis, backscatter system using light at a frequency of 5145Å. The velocity components measured are in or near the axial/tangential planes and the components resolved are the tangential and the axial velocity components.

The modifications to the original Lewis Research Center fringe system that are detailed below were initiated principally to allow the inclusion of a frequency shifter to the system.

The optical system used can be characterized as a two component, one color, three beam system. Table II and the schematic in Fig. 6 provide detailed information concerning the order in which the system was assembled, and Refs. [65, 66, 69] provide specific information on each commercially purchased system part.

This three beam system was designed to create two separate probe volumes. The beam pairs were perpendicular resulting in measurements of velocity components that were perpendicular to each other. During the current program, the frequency shifted beam (Fig. 5d) was common to both probe volumes. A mechanical shutter blocked one of the two unshifted beams, allowing only one probe volume in the target volume at any given moment. Therefore, only one component of velocity was measured at any given time.

Although the complete system originally was designed to measure two velocity components simultaneously, the velocity and turbulence intensities of the flow inside the test rotor precluded this mode of operation. For example, at the -5% chord location in the passage, the axial velocity and the absolute tangential velocities were of the same order of magnitude at a significant number of radial/tangential locations. These locations are near the leading edge of the rotor blades and reflect the approaching blade. The similarity of velocity

magnitudes precludes the effective separation of velocity component information taken simultaneously.

This region near the blade leading edge was not the only region where the two orthogonal measured velocity components were nearly equal. Regions in the vicinity of the trailing edge also showed similar velocity magnitudes.

The argon-ion laser was operated in the tem, mode (the lowest Transverse Electric Magnetic mode) at a power level varying between 1.5 and 2.5 W to generate a beam at a frequency of 514.5 nm. The variation in output power of the laser over the test period was the result of variations in input cooling water temperature and input electric power. The tem mode of operation provides the highest available power output for the laser used during this investigation. Even so, losses in the optical system reduced the actual power delivered to the probe volume to less than 0.8 W. These losses result from the unavoidable imperfect transmission of light through the various components of the optical train, both passive and active. This resulted in marginal operation in a number of regions. Specifically, areas near the hub where blade blockage reduced the light collected by the collection optics. Also, near the trailing edge where flow conditions have spread the seed and reduced the seed particle density. And finally, where the rear lip of the observation window reduced the viewing solid angle.

A brief discussion of the components used in this Laser Anemometer system has been included here to help clarify its assembly and

operation. More detailed information on the purposes and specifications are included in Ref. [65].

The laser is positioned below the optical package (Fig. 6) and its beam is directed through a vertical 180° turn up to the optics level of the Table. The beam then passes through a collimator which allows the laser beam waist to be positioned at the probe volume. This insures that the wave fronts are flat and, therefore, the resulting interference pattern is constant throughout the probe volume. This is important since the calculated particle velocity is a function of the heterodyne frequency which remains constant only if the wave fronts remain planar. Following the collimator are a series of beam splitters and polarization rotators.

The polarization rotators insure that the beams are properly polarized. It is important, to achieve a strong interference pattern, that the polarization of all three beams is the same. This was achieved within ±4°. It is also important to achieve an equal power split among all three beams. Unfortunately, the available equipment precluded achieving both the proper power split and the same polarization simultaneously. Therefore, since the proper polarization has proved more important than an equal power split in achieving high contrast fringes, polarization was optimized in lieu of power split. This resulted in a Bragg shifted beam with approximately 0.1 W less power (0.25 versus 0.35 W.). This number varied somewhat with the available laser power output and with adjustment of the Bragg cell.

The frequency shifter changes the middle beam frequency by 40 mHz, causing the interference pattern to "move" in the probe volume and thereby adding a velocity offset to the emitted signal. For example, if a particle were stationary in the probe volume, the movement of the pattern over the particle would create an apparent velocity that is associated with the 40 mHz signal. There are several advantages to this technique. The bias eliminates the problem of directional ambiguity. Laser Anemometry provides only a line along which the velocity is observed and a velocity magnitude is measured. In flows with low or zero velocities, the direction of particle motion, even if observed, may be uncertain. This problem is circumvented by using frequency shifting.

The use of the frequency shifting technique also eliminates the problem of angle biasing. Angle biasing is a bias in the measurements that result from a variation in the actual flow angle. The signal processor used in this present work requires that a seed particle cross at least 8 fringes as it passes through the probe volume. Therefore, more valid measurements per unit time occur when the flow is parallel to the fringe normal. In an unshifted probe volume, the error is proportional to the angle between the fringe normal and the velocity vector. In a Bragg shifted system, virtually any particle entering the probe volume will cross 8 fringes since the fringes are moving at relatively high speed.

Another advantage to frequency shifting is that, since the fringe pattern moves through the probe volume with a frequency of 40 mHz, it allows the probe volume to be focused to its minimum size and still

retains the required minimum number of fringes (8) for signal processing. This maximizes power density and minimizes position uncertainties.

The beam stop blocks out extraneous laser beams generated by the frequency shifter and the beam steering module allows the two separate probe volumes to be accurately superimposed.

The beam pair selector consists of a high speed shutter that selects two of three laser beams generated in the system. They are orientated as if at the corners of a right angle isosceles triangle (Fig. 5e). The triangle is orientated so that one of the right angle sides is parallel to the axial direction and one is parallel to the tangential direction. The beam at the right angle corner (the vertex of the triangle) is frequency shifted and is the common beam to both of the two sets of beams. The beam selector blocks one of the nonfrequency shifted beams. These two sets of beams create two orthogonal probe volumes that measure the axial and the tangential velocities.

The 2.27X beam expander is used to effectively reduce the size of the probe volume. Since the probe volume linear dimensions are a function of $(1/D_{\rm e-2})$, an increase in the beam diameter $(D_{\rm e-3})$ by 2.27 results in a corresponding reduction in the probe volume length and width. The principle advantages in using a beam expander are the reduction in the depth of the measuring volume and the increase in the power density in this volume.

As discussed in section 3 below, the light fluoresced by the particles in the probe volume is orange in color. The orange band-pass

filter stops light of frequencies other than orange from entering the photomultiplier tube. This can improve the signal-to- noise ratio by eliminating extraneous reflected light. However, during the current investigation, the orange pass filter was used only near the rotor endwalls, since the fluoresced light intensity is only approximately 10% that of the light scattered in the direct backscatter mode. Since signal strength was marginal in the direct reflection mode in most regions, operation in the fluorescent mode was normally not feasible.

The final optical element in the beam's path is a 3.2 mm thick window made of chemically strengthened glass and extends from 1.5 blade chord lengths upstream to 1 chord length downstream of the rotor blades. The window covers a circumferential width of 20° or two rotor pitches. It conforms with the outer flow path in both the axial (streamwise) and tangential directions.

An inhouse code [70] was used to verify that refraction errors created by observing the flow through this window would not seriously degrade the signal acquired.

Further, during the daily operation, the probe volume position was accurately determined prior to the start of the days data acquisition phase.

All optical elements were either bolted to the optical table with clamps or attached with permanent magnets.

The probe volume is an ellipsoidal shaped region approximately 125 μm wide and 2 mm long.

To obtain data the light emitted or reflected from the probe volume is collected by a focusing lens. It is then directed through an orange pass filter and another focusing lens that focuses it through a 125 μ m diameter pinhole onto the face of a photomultiplier tube.

The combination of the 160 mm and the 200 mm focal lengths of the transmitting and receiving lenses actually results in a pinhole image diameter of 125 μ m at the probe volume. The beam waist diameter at the probe volume dictates the selection of the image diameter and, hence, the pin-hole diameter.

The lenses are cemented doublet, commercially available, corrected to obtain "negligible" spherical aberration.

Table III provides the design probe volume specifications and the calculated probe volume specifications from the measured crossing angles.

2. Positioning System

The entire optical system is mounted on top of a mobil metal cart.

This cart can be rolled away from the compressor facility for easy

access and maintenance. When rolled into position, the cart is bolted

to the floor to insure a firm base for the laser anemometer.

Mounted on the cart are three commercially available translating stages. These stages provide more than 45 cm of movement in three

orthogonal directions. This movement insures complete coverage of a rotor passage.

3. Flow Seeding

Liquid seed particles, nominally 0.5 μ m in diameter, are injected into the flow through a 6 mm diameter tube located 35 cm upstream of the rotor.

In order to minimize the problem of a noisy signal produced from light scattering off solid surfaces, the fluorescent dye technique [67] was used near rotor endwalls. Here, fluorescent seed particles absorb the incident laser light and emit light at a different wavelength. An optical filter in the receiving optics blocks the unwanted reflected light scattered from surfaces near the probe volume. The selected seed material was a 0.02 molar solution of Rhodamine 6G in a 50-50 mixture by volume of ethylene glycol and benzyl alcohol. This material fluoresces orange when it absorbs green laser light. As was mentioned earlier, beam power limitations restricted operations in the fluorescent mode to regions immediately adjacent to the endwalls. As will become apparent from a review of the provided output, the data obtained in this mode of operation is of far poorer quality.

In the passage areas away from the endwall regions, direct light scattering from the seed was observed. In these regions of low

background light levels, the greater intensity of the scattered light greatly enhanced the data acquisition rates.

C. DATA ACQUISITION AND REDUCTION

The data acquisition system consisted of the optical system to acquire the signal, a signal processor to do preliminary processing and validation of the data, a shaft angle encoder to provide information on the circumferential (or tangential) position of the compressor rotor and a minicomputer to control the data acquisition and do preliminary post processing. A laser buffer interface received the signal from the signal processor and the shaft angle encoder, combined them and sent them to the minicomputer. The signal processor, shaft angle encoder, and minicomputer are described in greater detail below.

1. The Minicomputer

The data acquisition and online data reduction was greatly enhanced by the addition of a larger, more capable minicomputer and the implementation of additional software.

The current minicomputer, a VAX-11/750 using the VMS operating system is described in some detail in Ref. [72]. The machine contains sixteen 32-bit general purpose registers. The computer has hardware floating point multiply-divide capability. It also has dual cartridge magnetic disk storage with a total capacity of 360 million 16-bit words.

The minicomputer terminal has a CRT display for presentation of both alphanumeric and graphic information.

Typical data acquisition sessions required the use of the machine simultaneously in at least three ways: (1) data acquisition, (2) probe volume positioning, and (3) preliminary data screening and observation.

2. The Signal Processor

PMT signal bursts were processed by a commercially available counter type processor [73]. This processor contains a 250 mHz clock accurate to within ±1 nanosecond.

Invalid measurements will result from any of the following conditions:

- (1) amplifier saturation,
- (2) amplitude limit exceeded,
- (3) an end of burst condition before eight cycles are detected,
- (4) more than 254 cycles per burst,
- (5) failure to meet the comparison criteria:

$$t_8 \cdot \frac{8}{5} - 2 \cdot 10^{-9} \text{ sec } \le t_5 \le t_8 \cdot \frac{8}{5} + 2 \cdot 10^{-9} \text{ sec}$$
 (7)

where t_8 = time measured for 8 cycles and t_5 = time measured for 5 cycles,

(f) time measured exceeds 224-2 counts.

A validated signal representative of the time to cross eight fringes and, thus, the appropriate velocity component, is sent to the minicomputer along with a signal from the electronic shaft angle encoder. This latter signal defines the tangential position of the rotor when the PMT signal burst was processed.

Previously provided to the minicomputer is information concerning the velocity component measured, probe volume axial position, and fluid temperature.

3. Electronic Shaft Angle Encoder

This device, developed in part at the Lewis Research Center, provides the current angular position independent of the rotor speed with the only required input being an accurate once-per-rev pulse (OPR). The encoded angular position of the rotor is produced by a counter that is clocked by a frequency synthesizer. The synthesizer frequency is adjusted as necessary each revolution so that the number of counts per revolution remains constant.

An operational requirement of the shaft angle encoder is that the fractional change in the angular velocity of the rotor for each revolution be small compared to the desired resolution in the angular position expressed as a fractional change in the angular position expressed as a fractional part of one revolution.

During the current project, the rotor had 36 blades and a design operating speed of 17189 RPM. It was operated at 60% design speed or

approximately 10313 RPM (then corrected to standard day conditions).

For a desired count of 200 per blade passage, the counts per revolution are 7200. The long term (1 sec) speed drift in the facility is about 0.3% (22 counts per 7200). However, the rev-to-rev speed changes are less than one count.

For the data taken during this study, the encoder was divided by four to yield a resolution of 50 angular positions per blade passage (suction surface to suction surface).

Note that the velocity measurements do not occur at discrete shaft positions, but rather are made anywhere within an interval between adjacent shaft positions marked by the shaft angle encoder. This interval is along an arc in the flow passage at a given region radial/axial location. For the data taken in this experiment, this arc length varies between 0.621 mm at the hub and 0.896 mm at the rotor tip.

At each radial and axial position surveyed, data are recorded at 1100 different shaft positions. These shaft positions are distributed as 50 positions per passage across 22 consecutive passages. The velocity and flow angle are calculated at each position from runs made at the two different beam orientations. The velocity distribution across the measured 22 passages is considered to be 22 separate observations of the flow in an average blade passage.

Velocities and flow angles at corresponding points relative to the blade in each individual blade passage are averaged together to yield a spatially-ensemble averaged blade-to-blade velocity and flow angle distribution; i.e. an "average passage" velocity field.

A typical run for the laser data acquisition system consists of collecting approximately 5000 measurements at a given axial/radial position, yielding an average of about 5 measurements at each of the 1100 measured shaft angle positions. Run times varied from 1 to 30 min for each position. Generally, the closer to the endwalls and the farther downstream in the rotor, the longer the data acquisition time needed and the poorer the quality of the data.

4. Conventional Instrumentation

The conventional pitot-static instrumentation available in the facility was used to set and monitor the rotor operating conditions. It was also used to provide information for the analytical solution that was subsequently used for comparison and analysis of the experimental data.

This conventional data and the rig operating conditions were monitored and recorded using the center-wide "Escort" data acquisition system [74]. This system is an interactive, real time data acquisition, display, and recording system which is used for steady state measurements. The system consists of a remote acquisition microprocessor (RAMP), data input and output peripherals, and a minicomputer. The minicomputer coordinates and executes all real time processing. The RAMP acquires the data from the facility instruments, sends the data to the minicomputer, and distributes the processed data from the minicomputer to the display device.

Surveys of flow conditions upstream and downstream were made on a regular basis to insure the stability of the flow conditions.

The rotor casing was modified for this test to include substantial additional instrumentation. This instrumentation includes static taps on the casing and hub, and surveys upstream and downstream of the rotor. Table IV shows the locations of the surveys. Table V provides the positions of the static taps.

The survey instrumentation consisted of total pressure, static pressure, total temperature, and flow angle. Survey data was taken at 9 radial locations and two axial locations.

Total pressure, total temperature, and flow angle were measured with a combination cobra probe similar to the one shown in Fig. 7a. The static pressure was measured with an 8° "C-shaped" wedge probe similar to that shown in Fig. 7b. Each probe was positioned with a null-balancing, stream-directional-sensitive control system that automatically aligned the probe to the direction of flow. The material used for the thermocouples was iron-constantan. The wedge probes were calibrated in a low speed air tunnel. The total pressure and thermocouple probes were calibrated to a freestream Mach number of 0.9 in a high speed wind tunnel. Two combination and two static wedge probes were used both upstream and downstream of the rotor.

A calibrated flat-plate orifice was used to determine the flow rate and an electronic speed counter, in conjunction with a magnetic pickup, was used to measure rotative speed (RPM).

Data uncertainties due to the inherent errors of the instrumentation and recording systems are given in Table VI.

D. TEST FACILITY

The entire investigation was carried out at the Lewis Research Center single stage compressor test facility. A schematic of this research facility is shown in Fig. 8. It is described in detail in Refs. [75, 76].

The drive motor for the system consists of a 3000 HP electric motor with a variable-frequency power supply. Motor speed can vary from 400 to 3600 RPM. The motor is coupled to the compressor rotor through a 5.52 ratio gear box that increases the compressor rotor speed to an approximate maximum of 19850 RPM (the program rotor design speed was 17189 RPM). Ambient, unconditioned air was the working fluid and was drawn in from the roof of the building and exhausted through the basement of the building. The amount of airflow was measured across a thin plate orifice and was controlled using the downstream collector valve.

E. RESEARCH COMPRESSOR

The aerodynamic and mechanical design of the compressor used for this experiment is presented in detail in Ref. [75]. Aerodynamic performance is presented in Ref. [77]. The design parameters are rotor for an eight stage compressor. The first stage design pressure ratio was 1.82. The blades were designed to be multiple circular arc blade profiles. Figure 9 shows a meridional plane view of the compressor flow path and Fig. 10 shows the rotor blade sections at three radial locations: near the hub, midspan, and near the tip. Table VIII provides axial and radial geometry for both the endwalls and the blade leading and trailing edges. Rotor design tip clearance was 0.5mm.

F. COMPUTATIONAL DEVELOPMENT

1. Introduction

Recently a fully three-dimensional Navier-Stokes code has become available at the Lewis Research Center. This program, developed inhouse by Dr. R. V. Chima, was developed for the analysis of turbomachinery blade rows and other internal flows [27].

It was used for the analysis of both the baseline and the enhanced endwall test configurations of this investigation. The information presented below was taken from Refs. [27] and [78]. A more detailed presentation is available in these publications.

2. Computational Grid

This Navier-Stokes solver requires the generation of a number of computational grids spanning various portions of the solution space.

Initially, a coarse, equally spaced, meridional grid, Fig. 9, is generated between the supplied hub and shroud contours. The blade geometry is interpolated onto this coarse grid.

Next a series of two dimensional blade-to-blade grids are generated along the meridional grid lines using a code developed by Sorenson [79]. This series of two-dimensional C-type grids are reclustered spanwise to form a fully three-dimensional grid. The two-dimensional grids are arranged along the radial stacking line and stretched so that the blade shape remains constant and the angular pitch of the outer periodic boundary remains constant. This grid is shown in Fig. 11. Once the grid is generated, the coordinates are transformed from cylindrical to cartesian coordinates for the solution.

Finally, because the C-type grid generated does not extend far enough upstream, the grid generation program also generated an H-type grid for the upstream solution space.

3. Governing Equations

The Navier-Stokes equations were written in a Cartesian coordinate system rotating with an angular velocity, w, about the x-axis. This

rotation introduces source terms into the y- and z- momentum equations. These governing equations are mapped onto a general body-fitted (ξ,η,ζ) coordinate system; the ξ -coordinate direction is assumed to follow the flow direction. The thin-layer approximation is used to eliminate all viscous terms in the flow direction while retaining all viscous terms in the cross channel plane. The resulting equations are:

$$\partial_{t}q + J[\partial_{\xi}\hat{E} + \partial_{\eta}\hat{P} + \partial_{\zeta}\hat{G} - \frac{1}{R\Theta} \left(\partial_{\eta}\hat{P}_{V} + \partial_{\zeta}\hat{G}_{V}\right)] = H \tag{8}$$

where:

$$\hat{q} = J^{-1}[\rho, \rho u, \rho v, \rho w, e]^{T}
H = [0, 0, -\omega \rho w, \omega \rho v, 0]^{T}
\hat{E} = J^{-1}[\rho U', \rho u U' + \xi_{p} \rho, \rho v U' + \xi_{p} \rho, \rho w U' + \xi_{p} \rho, e U' + \rho U]^{T}
\hat{F} = J^{-1}[\rho V', \rho u V' + \eta_{p} \rho, \rho v V' + \eta_{p} \rho, \rho w V' + \eta_{p} \rho, e V' + \rho V]^{T}
\hat{G} = J^{-1}[\rho W', \tau u W' + \zeta_{p} \rho, \rho v W' + \zeta_{p} \rho, \rho w W' + \zeta_{p} \rho, e W' + \rho W]^{T}$$
(9)

It should be noted that the velocities in Eq. (9) are absolute with respect to a coordinate system that is fixed to the blade. Relative velocities (denoted by prime) are given by:

$$u' = u$$

$$v' = v - \omega z$$

$$w' = w + \omega y$$
(10)

Further, the relative contravariant velocity components are given by:

$$U' = \xi_{x}u' + \xi_{y}v' + \xi_{z}w'$$

$$V' = \eta_{x}u' + \eta_{y}v' + \eta_{z}w'$$

$$W' = \zeta_{x}u' + \zeta_{y}v' + \zeta_{z}w'$$
(11)

where it should be observed that u'=u but U'≠U.

The energy and static pressure are given respectively by:

$$e = \rho \left[c_{\nu} T + (u^2 + v^2 + v^2) / 2 \right] \tag{12}$$

$$p = (\gamma - 1) \left[e - \rho \left(u^2 + v^2 + w^2 \right) / 2 \right] \tag{13}$$

Using the Stokes' hypothesis, $\lambda = -\frac{\pi}{4}\mu$, the viscous flux can be written:

$$\hat{F} = J^{-1}\mu \left[0, F_2, F_3, F_4, F_5\right]^T \tag{14}$$

where

$$F_{2} = C_{3}\partial_{\eta}u + C_{2}\eta_{x} + C_{3}\partial_{\zeta}u - C_{4}\eta_{x} + C_{5}\zeta_{x}$$

$$F_{3} = C_{1}\partial_{\eta}v + C_{2}\eta_{y} + C_{3}\partial_{\zeta}v - C_{4}\eta_{y} + C_{5}\zeta_{y}$$

$$F_{4} = C_{1}\partial_{\eta}w + C_{2}\eta_{x} + C_{3}\partial_{\zeta}w - C_{4}\eta_{x} + C_{5}\zeta_{x}$$

$$F_{5} = \frac{\mu\gamma}{P_{T}}\left[C_{1}\partial_{\eta}\left(C_{v}T\right) + C_{3}\partial_{\zeta}\left(C_{v}T\right)\right] + uF_{2} + vF_{3} + wF_{4}$$
(15)

and

$$C_{1} = \eta_{x}^{2} + \eta_{y}^{2} + \eta_{z}^{2}$$

$$C_{2} = \frac{1}{3} (\eta_{x} \partial_{\eta} u + \eta_{y} \partial_{\eta} v + \eta_{z} \partial_{\eta} w)$$

$$C_{3} = \eta_{x} \zeta_{x} + \eta_{y} \zeta_{y} + \eta_{z} \zeta_{z}$$

$$C_{4} = \frac{2}{3} (\zeta_{x} \partial_{\zeta} u + \zeta_{y} \partial_{\zeta} v + \zeta_{z} \partial_{\zeta} w)$$

$$C_{5} = \eta_{x} \partial_{\zeta} u + \eta_{y} \partial_{\zeta} v + \eta_{z} \partial_{\zeta} w$$

$$(16)$$

Terms multiplied by C_1 and C_2 lead to nonmixed second derivative viscous terms, e.g., $u_{\eta\eta}$. On the other hand, terms multiplied by C_3 , C_4 , and C_5 lead to mixed derivative terms such as $u_{\eta\zeta}$. The viscous flux vector G can be written similarly, with directions η and ζ everywhere interchanged.

Metric terms are defined using the following relations.

$$\begin{bmatrix} \xi_{x} & \eta_{x} & \zeta_{x} \\ \xi_{y} & \eta_{y} & \zeta_{y} \\ \xi_{z} & \eta_{z} & \zeta_{z} \end{bmatrix} = J \begin{bmatrix} y_{\eta} z_{\zeta} - y_{\zeta} z_{\eta} & y_{\zeta} z_{\xi} - y_{\xi} z_{\zeta} & y_{\xi} z_{\eta} - y_{\eta} z_{\xi} \\ x_{\zeta} z_{\eta} - x_{\eta} z_{\zeta} & x_{\xi} z_{\zeta} - x_{\zeta} z_{\xi} & x_{\eta} z_{\zeta} - x_{\xi} z_{\eta} \\ x_{\eta} y_{\zeta} - x_{\zeta} y_{\eta} & x_{\zeta} y_{\xi} - x_{\xi} y_{\zeta} & x_{\xi} y_{\eta} - x_{\eta} y_{\xi} \end{bmatrix}$$

$$(17)$$

where

$$J = (x_{\xi} y_{\eta} z_{\zeta} + x_{\zeta} y_{\xi} z_{\eta} + x_{\eta} y_{\zeta} z_{\xi} - x_{\xi} y_{\zeta} z_{\eta} - x_{\eta} y_{\xi} z_{\zeta} - x_{\zeta} y_{\eta} z_{\xi})^{-1}$$
(18)

The equations are nondimensionalized by arbitrary reference quantities. The Reynolds number, Re, and the Prandtl number, Pr, are defined in terms of these quantities. It is assumed that the specific heats C_p and C_v and the Prandtl number are constant, that Stokes' hypothesis is valid, and that the effective viscosity for turbulent flows may be written as:

$$\mu_{eff} = \mu_{lam} + \mu_{curb} \tag{19}$$

where the laminar viscosity is calculated using a power law function of temperature:

$$\frac{\mu_{lam}}{\mu_{ref}} = \left(\frac{T}{T_{ref}}\right)^n \tag{20}$$

with n=% for air.

4. Turbulence Model

The Baldwin-Lomax algebraic two-layer eddy viscosity model [80] is applied on cross-channel (η,ζ) planes. Two modifications to the

standard model are made to account for the endwall boundary layer, the blade boundary layer and wake, and their interactions at the corners.

First, the distance from the wall is calculated using the Buleev [81] length scaled:

$$d = \frac{2s_{\eta}s_{\zeta}}{s_{\eta} + s_{\zeta} + (s_{\eta}^{2} + s_{\zeta}^{2})^{\frac{1}{2}}}$$
 (21)

where s_{η} and s_{ζ} are normal distances from the walls in the η - and ζ directions, respectively. This length scale has the desirable property
that d approaches the normal distance from one wall at large distances
from the other wall.

Secondly, the turbulent viscosities are calculated across each boundary layer or wake separately and then the total turbulent viscosity is taken as the vector sum of the components. This assumption has the following desirable properties: (1) that outside of one viscous layer, $\mu_{\rm turb}$ takes on values calculated for the other layer, (2) that it goes to zero in the core flow, and (3) that near the corners it accounts for both walls.

5. Boundary Conditions

At the inlet, total temperature $T_{\mathfrak{G},\mathbf{ref}}$ is specified as a constant. A ζ -distribution of total pressure $(P_{\mathfrak{G}}/P_{\mathfrak{G},\mathbf{ref}})$ is specified, as a constant or as appropriate for an inlet boundary layer with a given

thickness and a power-law profile. The inlet whirl distribution, rv_{θ} , is specified.

For both test cases, measured survey data as shown in Fig. 12 was input to the solver.

The hub exit static pressure is specified and $(\rho, \rho u, \rho v, \rho w)$ are extrapolated. The exit radial pressure distribution is found by integrating the axisymmetric radial momentum equation:

$$\frac{dp}{dt} = \frac{\rho v_0^2}{I} = \frac{\rho}{r^3} (vz - wy)^2 \tag{22}$$

Sidewalls and the trailing edge are treated as periodic boundaries.

On the blade surface V'= U'= W'= 0 for viscous flows such as the current test cases. Blade surface pressures are found from the normal momentum equation. On the hub $\zeta=1$ and on the tip $\zeta=\zeta_{max}$:

On the blades ($\eta=1$) the normal momentum equation can be found from Eq. 23 by replacing (everywhere by η and V' by W'.

6. Multistage Runge-Kutta Algorithm

The governing equations are discretized using a node-centered finite difference scheme. Second order central differences are used throughout.

The multistage Runge-Kutta scheme developed by Jameson, Schmidt, and Turkel [82] is used to advance the flow equations in time from an initial guess to steady state. If Eq. (8) is rewritten as

$$\partial_e q = -J[R_I - (R_V + D)] \tag{24}$$

where $R_{\rm I}$ is the inviscid residual including the source term, $R_{\rm V}$ is the viscous residual, and D is an artificial dissipation term described in the next section, then the multistage Runge-Kutta algorithm can be written as follows:

$$\begin{aligned}
Q_{0} &= Q_{n} \\
Q_{1} &= Q_{0} - \alpha_{1} J \Delta t [R_{I} Q_{0} - (R_{V} + D) Q_{0}] \\
&\cdot = \cdot \\
\vdots &\vdots \\
Q_{k} &= Q_{0} - \alpha_{k} J \Delta t [R_{I} Q_{k-1} - (R_{V} +) Q_{0}] \\
Q_{n+1} &= Q_{k}
\end{aligned} (25)$$

For efficiency both the physical and artificial dissipation terms are calculated only at the first stage, then held constant for subsequent stages.

7. Artificial Dissipation

The dissipative term D in Eq. (24) is a nonconservative version of that used by Jameson et al. [75]. It is given by:

$$D_{q} = (D_{\xi} + D_{q} + D_{\zeta}) q \tag{26}$$

where the ξ -direction operator is given by:

$$D_{\xi}q = C(V_2q_{\xi\xi} - V_4q_{\xi\xi\xi\xi}) \tag{27}$$

where

$$C = \frac{1}{J\Delta t} \tag{28}$$

is a coefficient that cancels similar terms in Eq. (25). To minimize the artificial dissipation in viscous regions, C is reduced linearly across several grid points to zero at the walls. The terms V_2 and V_4 in Eq. (27) are given by:

$$V_2 = \mu_2 \max(v_{i+1}, v_i, v_{i-1}) V_4 = \max(0, \mu_4, V_2)$$
 (29)

where

$$v_{i,j} = \frac{|P_{i+1,j} - 2P_{i,j} + P_{i-1,j}|}{|P_{i+1,j} + 2P_{i,j} + P_{i-1,j}|}$$
(30)

and

$$\mu_2 = O(1) \mu_4 = O(\frac{1}{16})$$
 (31)

In smooth regions of the flow the dissipative terms are of third order and do not detract from the formal second-order accuracy of the scheme.

8. Three-Dimensional Stability Limit

Applying a linear stability analysis to the inviscid form of the governing equation gives the following expression for the time step: where

$$\Delta t \le \frac{CFL}{1_{x} (1 + 1_{y} |v| + 1_{y} |w| + c\sqrt{1_{x}^{2} + 1_{y}^{2} + 1_{x}^{2} + \omega^{2}}}$$
(32)

$$\begin{aligned}
 I_x &= K_x + m_x + K_y \\
 I_y &= K_y + m_y + K_y \\
 I_x &= K_x + m_x + K_y
 \end{aligned}
 \tag{33}$$

The Courant limit for a particular multistage scheme depends on the number of stages and the choice of coefficients, a_i , of Eq. (25).

To accelerate convergence to the steady state, the maximum permissible time step at each point is used so that the Courant number is constant everywhere. The time step is calculated once based on the initial conditions. It is stored and not updated during the calculations.

9. Implicit Residual Smoothing

Residual smoothing was introduced by Lerat [83] for use with the Lax-Wendroff scheme and was later applied to Runge-Kutta schemes by Jameson [84]. The technique involves replacing the residual calculated in Eq. (24) with a value that has been smoothed by an implicit filter such as:

$$(1 - \epsilon_{\xi} \delta_{\xi\xi}) (1 - \epsilon_{\eta} \delta_{\eta\eta}) (1 - \epsilon_{\zeta} \delta_{\zeta\zeta}) \overline{R} = R$$
(34)

where $\delta_{\xi\xi}$, $\delta_{\eta\eta}$, and $\delta_{\zeta\zeta}$ are standard second difference operators and ϵ_{ξ} , ϵ_{η} , and ϵ_{ζ} are smoothing parameters.

Linear stability analysis has shown that the Runge-Kutta scheme with implicit residual smoothing may be made unconditionally stable if the ϵ smoothing parameters are made sufficiently large. In one dimension:

$$\epsilon \geq \frac{1}{4} \left[\left(\frac{\lambda_c}{\lambda_c^*} \right)^2 - 1 \right] \tag{35}$$

gives unconditional stability if λ^* is the Courant limit of the unsmoothed scheme, and λ is a larger operating Courant number. In three dimensions different ϵ 's may be used in each direction, and their magnitudes may be often reduced below the value given by Eq. (35).

10. Application of the Solver

Experimental information of the inlet flow conditions was used as inputs to the solver. These initial experimental conditions are density, the velocity vector, and the internal energy at the measured inlet radial locations. All input conditions were nondimensionalized by the inlet stagnation speed of sound and stagnation density. The inlet endwall boundary layers were assumed fully turbulent.

The code was then solved for a number of mass flows to generate an operating map for the compressor for both test configurations. Flow points on the computed maps were matched to the test operating conditions and results at the same relative position on the flow map were compared. Mass flows for the baseline test case, 10.727 Kg/sec and

the enhanced endwall, 10.545 Kg/sec compared favorably with the mass flows calculated using the Chima solver, 10.661 Kg/sec for the baseline and 10.904 Kg/sec for the enhanced endwall.

Mass was conserved to within 0.1% between inlet and exit conditions for all calculations for both inlet test cases. Both test cases were carried through at least 2040 iterations, during which maximum residuals decreased by approximately two orders of magnitude. Both maximum and average residual values had reached minimum values by 1700 iterations and solutions were considered final by 2040 iterations.

Next, the ASCII files containing the two computed solutions were downloaded to the MicroVax from the Cray YMP for additional post processing. The information in these files was interpolated from the solution "C" grid to the locations where data was acquired using a spline fitting routine. The information was then plotted using the same plotting routine that was used to plot the acquired experimental results.

Since the computed mass flows do not match exactly, slight differences in velocity magnitudes and flow angles can be expected.

Finally, it should be noted that the solver, in it's current form, contains no provisions for tip clearance. Therefore, differences between computed flow solutions and measured flow conditions can be expected in regions near the shroud where clearance effects might be observed. However, since data was not successfully acquired beyond 90% span, this discrepancy is not expected to adversely affect the comparisons between experimental and predicted results.

IV. EXPERIMENTAL PROCEDURE

A. INTRODUCTION

The primary goal of the current research was to explore the effects of axisymmetric inlet flow radial distortion on the performance of a typical compressor rotor. To accomplish this, it was necessary to minimize the possibility of extraneous flow features developing in the passage due to facility instabilities which could be produced by small variations in rotor speed. Operation at a flow rate slightly lower than the peak efficiency point would tend to stabilize the flow and still minimize off-design flow abnormalities. Therefore, the flow condition analyzed was specified to be near the 60% design RPM peak efficiency point (60% DSPD). Operation at 60% eliminated shock induced secondary flows. As an additional benefit, data acquisition rates were greatly improved and noise levels were correspondingly reduced.

To further isolate the secondary flow effects caused by the inlet distortion, the data was acquired for two different inlet conditions. The first, or nominal, flow condition was specified to be at the near peak efficiency, "smooth" inlet velocity profile, 60% DSPD flow condition. Then, for the second flow configuration, axisymmetric screens were placed on the hub and shroud, approximately 25 cm upstream of the rotor. These screens had the effect of generating a nonuniform inlet profile, thereby developing the thicker inlet endwall boundary layers that enhance the passage vortex. These blocking screens consisted of a coarse mesh (four wires per 2.54 cm mesh) extending 3.05 cm from each endwall into the flow. A finer mesh (eight wires per

2.54 cm) screen was laid on top of the coarse screens and these extended from the endwalls approximately 0.76 cm into the flow.

The mesh design used was based on previous work done at the Lewis Research Center. The design has been shown to produce a smoothly varying inlet velocity profile. The selected screen coarseness was chosen to provide an inlet profile severe enough to provide substantial difference between the two test cases and mild enough to allow stable operation of the rotor. Further, the velocity profiles needed to be mild enough to allow a computational solution to be obtained.

Figure 12 shows both profiles of the absolute total velocity, V_{to}, and absolute flow angle, angle, as surveyed at the upstream aerodynamic survey location. The profiles were selected to be different enough to generate a measurable passage vortex but moderate enough to allow smooth operation of the rotor and successful modeling on the available analysis codes. Flow conditions were matched by adjusting the flow coefficients, as discussed by Wagner et al. [64]. Here, the area average inlet axial velocity was used to define the flow coefficient.

Aerodynamic data consisting of total pressures, static pressures, and temperatures were acquired at two locations; one upstream and one downstream of the rotor. The upstream location was 2.54 cm upstream of the hub/rotor leading edge intersection and the downstream location was at 10.668 cm downstream of the hub/rotor leading edge intersection.

Data acquired at the aerodynamic survey location was used for several purposes. First, this data provided the inlet pressure profile and exit static pressure required as inputs for the flow solver used in the

analysis portion of this project. Second, these profiles were used to adjust the calculated flow coefficients and, lastly, regular surveys during the data acquisition period were used to help maintain constant rig flow conditions.

The inlet surveys show the average total velocity, V_{to}, and absolute flow angle at a position well upstream of the rotor. A comparison between the two velocity profiles shows a difference of over 5 m/sec at the midspan of the passage with the thickened or artificially enhanced boundary layer (AEBL) velocities higher. The situation is reversed closer to the endwalls with the baseline profile over 6 m/sec larger near the hub endwall and over 4 m/sec higher on the shroud side of the passage. This difference in the inlet velocity profile is the desired difference to generate the passage vortex that is the object of this thesis. The differences here are of the order of 10% of the freestream velocities.

Differences in the inlet flow angle measurements between the two velocity profiles is small except near the shroud. There is some indication that the enhanced boundary layer is turned more at the midspan than the baseline with the converse being true at the endwalls. However, with the exception of the two measurements nearest the shroud, the differences are very small; less than 0.3°. More interesting is the fact that there seems to be, for both profiles, a turning of the inlet flow from axial near the hub to about 2° near the tip. The cause of this is unclear. There is nothing upstream of the rotor to cause any

turning of the flow. Further, while laser anemometer measurements indicate some turning of the flow upstream of the rotor, the profile is of a more constant magnitude from hub to shroud rather than one of increasing magnitude. It seems unlikely that the rotor would affect the flow this far upstream. Therefore, the most likely cause of this profile is a small, systematic error in the measurement of the flow.

Overall results of the surveys at the aerodynamic exit are shown in Fig. 12b. These results show the enhanced boundary layer case with only slightly higher exit velocities, less than 3 m/sec, over most of the flow passage. They also show slightly higher turning of less than 2° over most of the passage.

It is difficult to see any great differences between the two test flow conditions. However, the enhanced boundary layer case seems to be slightly more turned near the endwalls than at the midspan than the baseline case. Maximum differences are of the order of 2°. This increased turning would be consistent with the generation of the passage vortex.

Laser Anemometer data was acquired at eight different chord locations, the farthest upstream was at the upstream aero-survey location and the farthest downstream was at the 105% chord location. At each of these locations, data was taken at ten spanwise locations varying from 5% span (from the hub) to 90% span. No usable data was obtained at any chord location at 95% span. Figure 13 shows a schematic

of the locations and Table IX gives the chord, span, axial, and radial position of each location.

Little usable data was acquired at the 105% chord line due to extremely high noise levels. Table X lists the points where the data acquired was, at best, marginal. Data acquisition was attempted at the 95% span location but was unsuccessful due to high noise levels and to a rapid buildup of particle seed on the surface of the observation window.

The test program began with rig "shakedown" runs in September of 1987. Aerodynamic surveys of the rotor were accomplished in October of 1987. Laser Anemometer shakedown runs began in February of 1988 and data acquisition began in mid-September of that year. A facility wide shutdown forced the termination of data acquisition on November 1 of 1988. It is important to note that data was acquired over a six week period from mid-September to the end of October. During this time frame, stagnation temperatures at the inlet varied from approximately 25° C to below freezing and relative humidities from 20% to saturation conditions. A typical data acquisition period would start at 9:00 AM and would terminate near midnight.

To minimize the effect of varying atmospheric conditions, every effort was made to acquire the data for one chord location during one test period. Further, rig operating conditions were continuously monitored and the operating point of the rig was regularly corrected to standard day conditions.

The installation and removal of the screens for generating the enhanced endwall boundary layers required an extensive disassembly of the test compressor. For this reason, all "baseline" data was obtained first and then the screens were installed. After the screen installation, all the thickened endwall boundary layer data was secured. This strategy reduced the number of compressor assemblies to two.

The "growth" of the rotor, due to rotational forces and blade aerodynamic loading, was measured to insure the proper positioning of the probe volume. Due to the relative shortness of the blade, little growth occurred between 1% and 60% speed. This radial growth was less than 0.3 mm. Growth between 60% and design speed was about 0.3 mm. No measurable blade "untwist" occurred between 1% and 60% design speed.

B. RUN PERIOD SETUP AND DATA ACQUISITION PROCEDURES

Prior to each daily data acquisition period, the laser anemometer and the probe volume position with respect to the rotor were checked. Laser Anemometer alignment was checked by attempting to acquire signal at an upstream location at low flow conditions. Approximately 25% of the time a system realignment was required. To realign, a small wire was placed in the probe volume and the pinhole positioned to optimize PMT output signal. The wire simulated the presence of particles in the flow and provided a reflected image or signal. This image was superimposed on the pinhole and the PMT signal optimized. On these occasions, beam trueness (parallelness with respect to the other beams),

laser power, beam polarization, and probe volume coincidence (both probe volumes occupy the same space) were all checked. Of these variables, pinhole position and laser output power were most likely to require optimization.

After the laser anemometer alignment was optimized, the probe volume position with respect to the rotor was determined. This was done by positioning the probe volume on the front lip of the rotor disk. This position was accurately known with respect to the blade surfaces. Once the probe volume was accurately positioned on the rotor, software offsets in the laser anemometer positioning code were corrected to insure accurate positioning of the probe volume.

Aerodynamic surveys were taken upstream and downstream of the rotor at the start, finish, and at regular intervals of each data taking session. Flow conditions were continuously monitored during the run period to minimize run point drift. Estimates in the run point drift have been previously given in the facilities section of this thesis.

Prior to each day of testing, an inhouse computer program was run to determine the axial and radial positions for each point in the flow field to be surveyed. The passage was generally surveyed at one chord location during each data acquisition period. The data acquisition began at the closest to hub location to be surveyed (5% span) and moved radially outward towards the tip. This program also generated an optimum offset from radial that would minimize the blade blockage for runs inside the blade.

The aerodynamic design of the rotor blades resulted in a blade shape that "leaned" off a radial line. Figure 14a shows a schematic of a typical area blocked by the blade shadowing. This lean leads to a shadowing of portions of the blade passage. This blockage was minimized by moving the line of sight off radial while remaining in a plane perpendicular to the axis of machine rotation (Fig. 14b). This was accomplished by mounting turning mirror 18, shown in Fig. 6, on a goniometer cradle. It must be noted that when this was done, it caused a rotation of the beam planes equal in magnitude to the size of the radial offset.

The goniometer offset was increased and decreased for a number of runs in an unsuccessful attempt to obtain data nearer the blade surfaces (Fig. 14c). The maximum off-radial goniometer setting used was 6°.

This off-radial positioning of the goniometer was calculated using the inhouse computer code mentioned above. The calculation yielded the setting that would minimize the effects of blade shadowing. The size of the off-radial movement of the optical package was determined using an inhouse computer code that geometrically traced the optical paths of the incoming laser beams and calculated an off radial line-of-sight that would minimize the effects of blade blockage. This leads to a maximum velocity error of less than 0.6% for the tangential component. This maximum offset was used in only two locations. All velocities were corrected for goniometer offsets used. Table IX lists the various goniometer offsets used.

All laser anemometer data acquisition was done through the test cell minicomputer. Each radial/axial data point was manually entered into the software on the computer. This software then directed the proper positioning of the laser table.

Additional software acquired and processed the raw PMT signal to present a real time velocity or frequency histogram and ensemble averaged blade-to-blade velocity information to the operator. Example output is shown is Figs. 15 and 16.

The information in these two figures, and in Fig. 18, needs some clarification. The data acquisition software was designed to operate not only in turbomachinery but also in wind tunnels. Therefore, position is displayed in terms of cartesian coordinates. For this experiment, X and Xpos refer to axial location while Y and Ypos refer to the radial location. 2 and 2pos were not used and the position measurements were provided in inches. Zt or "beam rotation" refer to the measured velocity component where 0° indicates the axial throughflow component and 90° indicates the tangential velocity component. The term RT or "beam deflection" indicates the deviation from zero of the observation angle of the laser anemometer package, i.e., the off-radial goniometer angle. The total temperature of the flow is presented in degrees Rankine and "nmeas" presented in Fig. 16 refers to the total number of processed measurements acquired at that axial/radial location. *Run* of *Run nos* indicates which data runs are being presented. numbers are the first or original run numbers given in Table XI.

In Fig. 15, nplot is the number of plotted measurements. Lclip and hclip indicate the number of measurements not plotted because they exist beyond the plotted range. Vmean is the mean measured frequency and stdev is the standard deviation. In Fig. 15, the abscissa is the frequency in mHz. The data acquired has been artificially shifted by forty mHz. This offset has been removed for this figure. The negative frequencies reflect measurements of particles with negative tangential velocities. These velocities result in frequencies less than forty mHz.

In Figs. 16 and 18, the abscissa represents the ensemble average window in which a velocity measurement was acquired.

The laser anemometer data acquisition technique is presented in more detail in Ref. [68].

At each axial/radial location, an attempt was made to acquire at least 5000 data measurements. A reduced number of measurements were obtained at 75%, 95%, and 105% chords due to excessively low data acquisition rates in these regions.

Two orthogonal velocity components were obtained at each radial/axial location.

One of the input laser beams was frequency shifted by 40 mHz, thereby effectively eliminating the flow angle biasing problem and the directional uncertainty problem.

The laser anemometer survey locations in Fig. 12 represent, in actuality, arcs in space at these locations. During a data acquisition sequence, data began to be recorded on reception of a specific once-per-rev signal transmitted by the shaft angle encoder. Data was then

accepted over a period of time during which the rotor accomplished 0.6111 revolutions (22 of 36 rotor passages). During this time interval, 22 passages of the rotor passed through the stationary position of the probe volume. Whenever a valid signal was acquired, it was stored, along with the specific time interval that started after the arrival of the once-per-rev signal. Each velocity signal was then assigned a position in one of 1100 "windows" or bins that together made up the arc that spanned the 22 passages of the rotor over which data was acquired. This is 50 windows per passage. In the post processing phase of the project, these 22 passages were further combined into one "average" passage. At each axial/radial location in this average passage, an arc spanning the passage from suction surface to suction surface was divided into 50 windows. Each contained an average velocity of data acquired in that average arc segment from data acquired in each of the 22 passages at the geometrically equivalent arc segment.

This consolidation of data acquired over these 22 passages was justified because there were no appreciable differences in the data acquired in any of the passages. This indicated that, aerodynamically, the passages were equivalent and could be viewed as identical.

The time required to complete a revolution (the remaining 14 of 36 blade passages) was used by the software to store and perform preliminary processing of the acquired data.

PMT voltage was normally set at 1800 V for florescence and 1400 V when the seed particles were observed from direct scattering. This

voltage was varied to optimize data rate with a minimum of adverse noise effects.

The laser was operated at maximum power at all times. This varied from 1.6 to 2.4 W. Losses in the optical system typically reduced the power delivered to the probe volume for two beams by 75%, i.e., for the 2.4 W case only 0.6 Watts was delivered to the probe volume. Data acquisition with laser power levels less than 2.0 W was marginal or impossible in some locations and power levels of less than 1.6 W made data acquisition impossible at all locations. The variation in power output from the laser was the result of variations in the input power from the commercial power system. Since the laser power supply was closely tuned to the voltage level of the input power, small variations resulting from load variations on the commercial power grid caused substantial variations in the laser output power. The laser output power also depended upon the fine orientation of the laser reflector mirrors. These were hand adjusted, if necessary, prior to the days run and it was not always possible to achieve a perfect orientation.

C. DATA REDUCTION

At the end of the data acquisition phase, there were 960 sets of ensemble averaged velocity measurements. A set of measurements consisted of all the measurements of a single velocity component taken in each of the 1100 shaft positions at one axial/radial position. To obtain the complete measured velocity vector at an axial/radial

location, two sets of velocity measurements must be combined: one axial velocity and one tangential measurement. At each location, at least two sets of data, one for each velocity component, were taken for each flow configuration. The flow configurations were the baseline or uniform inlet configuration and the enhanced endwall boundary layer (AEBL) configuration. At many locations, more than two sets of data were acquired. At a minimum, 320 sets of laser anemometer data are required to complete the test (80 locations times 2 configurations times 2 velocity components).

The initial step in the data reduction process involved visually comparing all data sets at each location to determine the most noise free data at each location for additional processing. Acceptable runs with the same spatial location, goniometer setting, velocity orientation, and inlet configuration were combined into single sets. At least one set for each velocity component, regardless of data quality, was retained at each spatial location, for each inlet flow condition for additional processing. There were 486 sets of data selected for further processing and combining into the 320 final sets of measurements. Table XI shows the initial combined runs and indicates how many runs were combined for each of the initial 486 selected runs. These runs are paired with their sister run at that spatial location.

To this point, no attempt had been made to improve the quality of the data acquired by removing "outlier" noise that had been included in the online data processing. To accomplish this, each of the 486 runs was processed to provide three plots. These were similar to the two plots shown in Figs. 15 and 16 and as well as a frequency histogram output similar to that shown in Fig. 17.

Some explanation of Fig. 17 is required. For each run or set of data, all data in each rotor passage spatial location (radial/axial) was combined into an ensemble averaged rotor passage equal to one pitch (10°). This passage was further divided into 50 windows, each of which was 12 minutes wide. Spaced along the abscissa of Fig. 15 are histograms of each of these 50 windows. The ordinate of Fig. 17 represents velocity ranges. The second row from the top is the first velocity range that contains the lowest velocity measurement. The second row from the bottom contains the range with the highest velocity measurement. Each range is approximately 5 m/sec wide.

The number displayed in each of these bins or velocity ranges represents the number of successfully processed velocity measurements in that applicable velocity range. This number, however, has been non-dimensionalized with respect to the velocity range in that window containing the greatest number of measurements. For example, range 50 in window 10 (circled) displays a 4. Range 45 (circled) in the same window contains a 9 and represents the velocity range with the highest number of successfully acquired velocity measurements. The displayed 4 indicates this velocity range contains 30% to 40% of the number of measurements in range 45. Thus the figure gives a feel for the distribution of velocities in each of the passage averaged windows at an axial/radial location.

To reduce the "outlier noise," the number of measurements in each velocity bin of a data acquisition run was reduced by a constant number of measurements. This number was calculated to be 20% of the number of measurements of the bin that contained the maximum number of velocity measurements. This procedure can be considered roughly equivalent to adjusting the triggering level in the signal processor to minimize noise. The 20% figure was arrived at by individually reviewing each of the 486 runs to insure that the 20% reduction would not adversely affect the accuracy of the result.

Following this processing, the two orthogonal velocity vectors for each inlet flow condition, spatial (axial/radial/circumferential) position, and goniometer orientation were combined to create a combined axial/tangential velocity vector. Figure 18 shows a typical output. Goniometer orientation was taken into account here to insure that the correct axial/tangential velocity components were calculated.

Finally, the 243 remaining runs were combined to generate 158 separate datasets; one for each axial/radial location at each of two inlet flow conditions. There were two locations where no data were acquired due to time constraints during a run period. Each spatial location was divided into 50 circumferential bins (arcs in space) that were 12 min wide.

D. ERROR ANALYSIS

Schenk [86] divides the possible errors inherent in a given experiment into three general classes: (1) accuracy errors, (2) precision errors, and (3) uncertainty errors. Accuracy errors are repeatable deviations from correct values that are the result of inherent errors in the system of the experiment. For example, a positioning device that consistently reads "x" cm lower than its actual position. Errors such as these have been accounted for in the data acquisition and reduction phases of this investigation whenever they have been found or suspected. For example, frequency shifting stability and laser table positioning and leveling with respect to the compressor facility were established prior to the data acquisition phase.

Precision errors and uncertainty errors are random errors occurring during the acquisition of data.

Precision errors are errors, for example, in the stability and repeatability of the equipment used in the study. The stability of the compressor rotor speed during the acquisition of the laser data is an example of this type of error. Information is provided concerning the compressor test facility and the system positioning table and represents the "best" measured information concerning the maximum possible inaccuracies in these variables.

Uncertainty errors are errors in the data acquisition for which a maximum imprecision cannot be measured. The uncertainties of these

errors must be estimated statistically. Errors such as the velocity uncertainties are provided below.

Various errors discussed below can be divided into errors resulting

(1) in the operation of the test compressor, (2) in the position of the

probe volume, and (3) in the act of measuring the fluid velocities and

processing those measurements.

1. Compressor Test Facility

The compressor test facility has been in operation for a number of years and the inaccuracies associated with its operation are well documented [76]. Table VI presents a synopsis of the inaccuracies associated with the measurement of test facility operational parameters and conventional instrumentation used at the facility.

2. Laser Anemometer system positioning error

The accuracy of the optical system positioning table is 0.05 mm over a range of 25 cm in either direction. This figure is applicable to all three axes of movement. The direction of the input beams bisecting line can be determined to within ±0.01° using the goniometer over the goniometer operational range of ±3° from the compressor rig horizontal.

The measured precision error from the vertical and horizontal of the input laser beams is less than 1.4°. This results in a maximum measured velocity error of less than 1.5%.

The translating table was measured to be square with respect to the compressor facility to within less than 0.1°, resulting in a maximum positioning error of less than 0.38 mm over the distance of movement.

3. Discussion of Laser Anemometer system measurement error

There have been a number of excellent publications dealing with the assessment of errors in the data acquired using Laser Anemometry.

Information from three of these, Strazisar [87], Seasholtz [88], and Strazisar and Powell [89] was utilized extensively in assessing the errors present in the data acquired.

As has been previously mentioned, data was acquired over 22 of the 36 blade passages in the rotor at any of the desired radial/axial locations. At each of these locations an arc is described in space that traverses the 22 passages. Along this arc, each of these blade passages is divided into 50 windows or bins starting at the suction surface of the passage and extending to the next suction surface. This occurs 22 times. There are, then, 1100 windows (50 per passage). The passage to passage variations in the velocity fields have been minimal in previous investigations involving similar rotors. Therefore, to reduce the magnitude of the data processing problem, the measurements in the

windows in each of the 22 measured passages have been combined into one of 50 equivalently positioned windows (with respect to the rotor blade suction surface) in an "average" passage. Then the velocity for any given window may be calculated as:

$$V_{pav}(k) = \frac{1}{nme(k)} \sum_{m=1}^{nne} V(m, k)$$
 (36)

where $k=1,...,N_{wp}$, and nme is the number of measurements. The standard deviation is given by:

$$\sigma_{pev}(k) = \left[\frac{1}{nme-1} \sum_{m=1}^{nme-1} \left[V_{pev}(k) - V(m,k) \right]^2 \right]^{\frac{1}{2}}$$
 (37)

So, the average velocity in any of the 50 windows at an axial/radial location in the average passage is the average of all the velocity measurements in any of the windows at the same axial/radial location and the same circumferential location with respect to the suction surface of the rotor passage in which a particular velocity measurement was acquired.

Strazisar [87] states that the velocity probability density distribution is broadened by several factors which include:

- (1) the random turbulent fluctuations in the flow,
- (2) the flow unsteadiness that occurs at frequencies which are not integral multiples of the rotor rotational frequency,
- (3) the average velocity gradients across the width of the measurement window (typically 1% or 2%),

- (4) the flow variations caused by rotor speed drift during data acquisition (set to 0.4%), and
- (5) the individual Laser Anemometer measurement errors (such as PMT noise).

There are a number of uncertainties and bias errors inherent in the laser anemometer method of velocity data acquisition. Reference [87] provides a detailed discussion of the various errors associated with the laser anemometer. Table XII is a listing of estimates of the size of the errors involved.

Two sources of measurement error are: statistical biasing error and angle biasing error.

Statistical biasing results from the fact that, in a uniformly seeded flow, more particles of higher velocity cross the field of view per unit time than for lower velocity flows. Therefore, the calculated mean velocity measurement over a given period of time yields a velocity higher than the true mean. This bias can be removed using the relation:

$$V_{av} = \frac{(V_{bi})_{av}}{\left[1 + (\frac{V}{V_{av}})_{bi}^{2}\right]}$$
(38)

where "bi" denotes the biased measurements during the course of testing. Typical values of $(V/V_{av})_{bi}$ in the regions of usable data are of the order of 0.2%. Peak values in the leading edge region were as high as 10%. Since the vast majority of data included far smaller errors than the 10% near the leading edge region, this error was not taken into account in the data provided.

Angle biasing [87] is a result of variations in the flow direction with time. More measurements per unit time occur when the flow direction is parallel to the normal direction of the fringe pattern than when the flow direction fluctuates away from the fringe normal direction. This is because a particle entering the probe volume is more likely to cross the required minimum number of fringes to generate a signal that will be processed as valid. The error is proportional to the angle between the "fringe normals" and the mean flow direction. In the current experiment, the measured flow velocity measurements were frequency shifted. When frequency shifting is used, the frequency of one of the two beams that form the probe volume is slightly changed. In this experiment, the frequency shift was 40 mHz. The result is a change in the frequency of the signal emitted from a particle in the probe volume. Using the frequency model of the probe volume, it would appear that the fringes in the probe volume move. In such a probe volume, a stationary particle would emit a signal indicating a velocity as the fringes sweep across it. This signal, of the order of magnitude of the through-flow velocity, is optically added to all the measured velocity signals. This artificial velocity added to the optical signal insures that virtually any particle that enters the probe volume will cross the necessary fringes to generate a valid signal, thus eliminating the problem of angle biasing. Of course, this artificially added velocity is removed from the calculated velocity component during the post processing of the signal. More detailed discussion of these various statistical errors is presented in Refs. [87-89].

Another potential cause of error is particle lag. Particle lag is an inertial effect resulting from differences in density between the fluid being measured and the particles used to "seed" the flow that are actually being measured. There have been a number of analyses performed in the past to consider this problem [90-91]. It has been shown that particles less than 1 μ m in diameter are required for accurate flow measurement. Measurements of the particles used in seeding the flow indicated that the mean particle size was near 0.5 μ m. Less than 5% of the measured seed was greater than 1 μ m in diameter. In this well behaved, steady state, subsonic environment, error due to particle lag should be very small.

Not all of the errors in the measurement of the average velocity, or more precisely, the error in a single Laser Anemometer measurement can be directly assessed. This error associated with a single L.A. measurement is a function of, among other things, flow turbulence intensity, optical noise from various components (photomultiplier tube, Bragg cell, blade flash, etc.), and electronic noise from various components (signal amplifiers, cabling, connections, etc.).

It is, therefore, standard procedure at the Lewis Research Center to determine the precision error by using:

$$C_{v} = \frac{Z \cdot \sigma}{\sqrt{N}} \tag{39}$$

where C_{γ} is the confidence interval as a fraction of the calculated average velocity, σ is the calculated standard deviation, N is the

number of measurements, Z is a measure of the confidence interval (nominally set to 1.97 for a Gaussian confidence interval of 95%). This calculation is used to determine the error margins presented in Tables A5 and A6 of Appendix A below.

V. RESULTS AND DISCUSSION

A. INTRODUCTION

A considerable amount of experimental data were acquired over the course of this investigation. It included "conventional" steady-state aerodynamic data that encompassed, among other measurements, surveys of total pressures, static pressures, and temperature at the inlet and exit of the rotor. The axial through-flow and the absolute tangential velocities were obtained via laser anemometer measurements. In order to contrast these data, computational predictions of the flow field were made as well.

All of this information was obtained for two different inlet velocity profiles: a baseline inlet flow condition and an enhanced (thickened) endwall boundary layer condition. The latter flow was created, as was explained earlier, by placing axisymmetric distortion screens on the hub and shroud of the rotor upstream of the inlet.

Furthermore, all of the experimental data was acquired in a stationary reference frame. The computational predictions were made in a relative reference frame that was defined to be stationary with respect to the spinning rotor.

It must be understood that in a turbomachine, the flows into and out of a rotor are best analyzed in a stationary reference frame. This is because the component generating the rotor inlet flow conditions is stationary as is the downstream component that is affected by the rotor

exit flow. On the other hand, the rotor is spinning and the effects of the flow-rotor interaction are best considered in a reference frame that is stationary with respect to the rotor: a relative reference frame.

This reference frame is created by the addition of the rotor wheel speed to the fluid velocity.

Consideration of the various frames of reference were used to organize the presentation of the data. Accordingly, this chapter is divided into five main sections: (A) Introduction, (B) A Review of Data Acquired in the Stationary Reference Frame, (C) A Comparison of Results in the Stationary Reference Frame, (D) A Comparison of Results in the Rotating Reference Frame, and (E) Assessment of the Grid Velocity Deviations.

Section B is a station-by-station assessment of the acquired data. The section will provide a familiarization of the rotor flow, the type of data and its accuracy, and will briefly discuss some of the difficulties in interpreting the information gathered. Data from both inlet test configurations will be considered here because many of the difficulties in acquiring and interpreting the data are common to both cases. The computational results will also be reviewed when necessary to help clarify questionable data and to point out differences between the computational and experimental results.

The following section, section C, examines the inlet and exit flow features to assess what effects the different inlet profiles exert on the exit flow conditions.

Section D contains considerations of the differences between the two inlet profiles in the rotating reference frame, how these differences propagate downstream, and any unique flow features that occur in the rotor. The baseline flow condition will be considered first, and will be followed by the enhanced inlet boundary layer flow condition. Differences between the two will be noted. This section will also include comparisons between the computed predictions and the experimental data. Differences and similarities will be noted.

Section E provides a discussion of the Grid Velocity Deviations. The principal thrust of this experiment was to determine the effects of thickened endwall boundary layers on the development of the classic passage vortex. Since the radial velocity component was not measured, the actual passage vortex could not be measured. Nevertheless, some information on the development of the passage vortex can be obtained by comparing the measured velocity components with respect to each other and with respect to a calculated primary flow component. This section addresses this point.

Throughout the analysis of these flows, the computational results will be used as necessary to clarify the experimental results.

Before discussing the results obtained, some information which will help clarify the figures to be presented will be given herein.

In particular, Fig. 19 shows the coordinate system that is used in this analysis. When referring to a percent span, distance is measured from the hub, rather than as a percent immersion.

One way that the laser anemometry information is presented here is using cross channel plots. With one exception, these plots are of data at a constant chord location. The exceptions are the station 1 plots which are at a constant axial location upstream of the rotor. The view of these plots is downstream along the streamsurfaces seen in meridional view in Fig. 9.

All cross channel plots presenting both experimental results and computational predictions are viewed looking downstream into the rotor. Compressor rotation as viewed is clockwise. Therefore, the suction surface is at the bottom of each figure and the pressure surface on the top. The hub is at the left of the figure and the shroud is at the right.

Data presented in the figures containing line plots has been smoothed to improve the appearance and to clarify the flow features. To accomplish this, the separate measured velocities, axial and tangential, were individually reviewed and smoothed. As explained earlier, data was acquired at a number of axial/radial locations. At each location, velocities were measured along an arc spanning 22 passages. This data was combined into an arc spanning an average passage from one suction surface to the following blade suction surface. The data was smoothed by considering values along each of these arcs. A running average of the velocities in the nearest four windows along the arc to the window averaged was calculated. The velocity to be smoothed was not allowed to deviate more than 10% from that calculated average velocity.

A particularly difficult problem occurs in dealing with windows where no velocity measurements were acquired. These null velocity windows are not well handled by the spline curve fitting routine used and had to be eliminated from consideration. This was done by replacing all null velocity windows with values calculated by using linear interpolation between the velocities in the two windows bracketing the area of no measurements along the measurement arc considered.

With the exception of the -5% chord location, all tangential velocities were constrained to be within a range from 0.0 to 150.0 m/sec and axial velocities were constrained to be within 50.0 and 150.0 m/sec for this smoothing procedure. These limits were determined by reviewing the range of values in the calculated predictions for the regions where data was acquired and by reviewing the data itself to insure no regions existed that contained accurate velocity measurements outside of these ranges. The -5% chord location was not considered because it contained velocities outside of those ranges. Appendix A contains tables of all the measured test results obtained during the course of this investigation.

Two sets of cross channel plots that have not been smoothed are presented in Figs. 28 and 29. These are colorbar plots that are simple four corner averages.

Also presented in this Chapter are a number of mid-channel plots, specifically, Figs. 21, 23, 25, 27, and 32. They represent information from a window near mid-pitch, normally window 25, at each of the axial/radial locations. This mid-pitch location is not an "average"

location. It does, however, represent a location approximately equidistant from either blade surface. Thus, it is likely to contain information that is freer from blade light reflection noise. While blade-to-blade effects are more obvious at the more downstream locations, these plots can be instructive, with reservations, when searching for spanwise variations in the flow. Since the inlet conditions are axisymmetric and vary only in the spanwise direction, it is worthwhile to consider the information in these figures.

Appendix A contains computer printouts listing the measured data and calculated results. The 'cp' is the window number. Window 1 starts on the suction surface and window 50 is the window that ends at the suction surface of the blade which is one counterclockwise passage away. In the first column of the tables are the 5% span from hub values, in the next column are the 10% span from hub values and the values in each column thereafter are 10% farther from the hub. The final column lists values from the 90% span measurements. No data was successfully acquired at 95% span.

B. REVIEW OF DATA ACQUIRED IN THE STATIONARY REFERENCE FRAME

The velocities measured by the laser anemometer were the axial and the absolute tangential velocity. The radial velocity vector could not be obtained using the current laser anemometer configuration. The other basic data obtained included the number of laser measurements at each axial/radial/circumferential location in each of the 22 blade passages

average passage. This passage contains data obtained from all surveyed passages. Earlier experiments [67-69] have indicated that the differences between passages are extremely small and thus no important aerodynamic effects will be overlooked when combining all acquired data into one typical passage. The measured average passage axial and absolute tangential velocities, number of measurements, and calculated uncertainty ranges are presented in Appendix A.

The first part of this section contains a detailed review of the velocities measured at each location. This review should provide some strong indicators of which data obtained is accurate.

In areas near blade surfaces and passage endwalls, the quality of the acquired data is poorer due to increased optical noise. Data acquisition is further hampered by the difficulties in adequately seeding the flow in these regions. Seeding difficulties and metal blockage from the blades and rear window frame edge also reduced the quality of the measured data at locations further downstream.

We will look at the data one station at a time, starting with the inlet survey location and proceeding downstream. We will examine the baseline (uniform inlet) case and the AEBL (enhanced endwall boundary layer) case together but will consider the baseline case first at each station.

Please refer to Appendix A and Figs. 20 to 23, 41 and 42 for the following discussions in section A.

1. Station 1, Inlet Survey

In general, it can be said that the farther upstream and the farther from the endwalls that the data was acquired, the higher is the quality of the data obtained. The measurements made at the inlet survey location were the highest quality of any data at any of the test stations. The baseline case showed only one location where the information acquired was of questionable accuracy. This was at the 90% span location and is most likely the result of the accumulation of seed particles on the observation window. It should be noticed that in Table Al in Appendix A the velocities varied from 50 m/sec to 100 m/sec within the space of only three windows. The only negative flow angles measured occur at this span location. The differences between the 80% span and the 90% span coupled with the rapid variations in velocity at this location, far upstream and steady state, indicate the data at this location is unreliable.

Data at 90% span for the enhanced endwall boundary layer case does not have the large velocity variations or the negative flow angles.

Further, the number of measurements, error margins, and comparisons with the 80% span data suggest that this data is accurate.

The axial velocity profiles indicate a velocity bulge at the 10% span (Figs. 20 and 21) for the baseline configuration. This bulge does not seem to exist for the thickened endwall boundary layer test configuration. The aerodynamic measurements indicate a much smaller profile of this nature (Fig. 12) for both the baseline configuration and

the enhanced endwall boundary layer configuration. However, the aerodynamic measurements show much weaker "bumps" in the velocity profiles. This may reflect actual flow conditions or it may be the result of a change in operating conditions between the two span locations. Figure 21 is a midpitch plot of the axial velocities. The midpitch plot is useful in that it is the most distant from each blade surface and, therefore, the signal is least likely to be corrupted by stray light. The data for one circumferential position is much easier to interpret.

The effects of the distortion screens are apparent in the enhanced endwall boundary layer test configuration with lower velocities extending farther from the endwalls and the velocity profile having a more distinctive parabolic shape.

There is a tangential velocity component that has been measured for both the baseline and the enhanced endwall boundary layer test configurations at the inlet. This velocity component is not apparent in the aerodynamic measurements and, given the inlet configuration, it is not indicative of the actual flow conditions and has been discussed earlier.

2. Station 2, -5% Chord

This location is immediately upstream of the rotor and the effects of the rotor bow wave are very apparent in the velocity and flow angles measured.

The velocities measured at 90% span appear to be invalid for the baseline flow case. The results shown in Appendix A are lower and more unsteady. Higher uncertainties have been calculated for the 90% span location.

A study of the number of measurements and the uncertainties

(Appendix A) for the 5% and the 90% span readings highlights some of the difficulties in assessing the validity of these measurements. The 5% span data shows a large number of measurements and, as a result, very low uncertainties. Far more measurements have been processed for this radial location than for measurements at larger radial locations. The total number of measurements taken at any location was approximately 5000, but, for this location, approximately 10000 measurements were taken. However, the very large number of processed measurements at this location would indicate a spurious measurement, such as reflection from the hub or the blades, was processed as a valid velocity, since it reflects a much higher data rate.

In contrast, measurements at the 90% span location are more sparse, which indicates that a large number of measurements were found to be unusable and thus were not processed. The questions here are twofold. First, the validity of the unusually large number of measurements at 5% span and, second, the higher uncertainties at 90% span resulting from the fewer measurements at that location. Both of these argue against the usefulness of the data taken at these radial locations.

3. Station 3, 5% Chord

This chord location is immediately inside the rotor. It was very difficult to obtain data at this location, because of the blade blockage due to blade geometry.

No velocities were measured at the 5% span location for the baseline configuration and the information obtained at the 80% and 90% span locations was also not usable. The velocities measured at 10% span also reflect some deterioration with almost one out of five of the windows containing corrupted data.

The information from the enhanced endwall boundary layer configuration appears valid from 10% span to 80% span. Velocities measured at 5% and 90% are not indicative of actual flow conditions.

Uncertainty measurements, large variations between neighboring windows, and the disparity between the magnitudes at 5% and 90% spans and of the 10% and 80% spans suggest that these measurements are not correct.

Since this chord location is inside the blade row, the effects of blade blockage are apparent. In the baseline configuration, the zero measurements in windows 1 through 8 and again in windows 47 through 50 reflect the presence of a blade in one or both of the incoming laser beams. The numbers in windows 1 through 4 probably reflect that the system is trying to process data from reflections from the blades or endwalls.

The enhanced endwall boundary layer test configuration also contains data affected by blade blockage with 15% to 20% of the windows being blocked by the presence of a rotor blade.

Figure 41 presents the computed axial velocities for stations 3 to 7.

Station 3 results for the baseline experimental results and the baseline computed prediction show good qualitative agreement. The axial velocity contour plots show axial velocities varying from 90 m/sec near the pressure surface to 120 m/sec near the suction surface.

The computed predictions for the enhanced endwall boundary layer configuration show that the axial velocities also appear very similar to the experimental results. The computed results reflect a region of high velocities that peak at about 140 m/sec near midspan on the suction surface side of the passage. There exists a much larger region of 130 m/sec velocities that exist around this area. The experimental results for the enhanced endwall boundary layer show no velocities of 140 m/sec or greater but these results do show a large region of 130 m/sec. Both the computed predictions and the measured experimental results for the enhanced endwall case also indicate a region of lower velocities (50 m/sec) in both of the pressure surface endwall regions.

The enhanced endwall boundary layer case axial velocity prediction shows a large region of low velocities near the shroud for the enhanced endwall boundary layer case. This might be the result of the lower total velocities near the endwall for this test configuration, due to the inlet profile.

<-2

Figure 42 presents the predicted absolute tangential velocities at stations 3 to 7 for both the baseline and the enhanced endwall boundary layer test cases. for both inlet configurations, regions of low tangential velocities exist near the suction surface. Computed predictions for both cases show blade pressure surface effects as the absolute tangential velocity field shows increasing velocities in the region near the pressure surface. Few differences exist between the predictions for the absolute tangential velocities for the two test cases. The exception is a slightly larger region of zero absolute tangential velocities for the enhanced endwall boundary layer test configuration.

The measured tangential velocities for both the baseline and the enhanced endwall boundary layer test cases reflect the same trends as the predicted computation results. A region of low or zero absolute tangential velocities exists from hub to shroud near the suction surface. Then, as the pitch increases towards the pressure surface, the tangential velocities increase. This region of increasing tangential velocities near the pressure surface is not prominent in the experimental results. This is the result of the poorer quality of the data that occurs in the regions near solid surfaces.

No regions of zero velocity are apparent for either set of experimental results. This would indicate that the spurious tangential velocity bias that was observed at the inlet station was also present in the downstream measurements.

4. Station 4, 25% Chord

Station 4 measurements highlight another of the difficulties in assessing the data acquired near the endwalls of the passage.

The baseline case velocities at 5% and 90% seem very reasonable but are lower in magnitude than the velocities nearer the center of the passage. In both locations, higher tangential velocities are obtained in conjunction with much lower axial velocities, indicating greater turning in these regions.

These trends are reflected, to some extent, in the computational predictions. However, when the quality of the data deteriorates, it is often accompanied by a gradual lowering of the magnitude of the measured velocity component. The first component to be lost using this laser anemometer system is normally the axial component. Therefore, the information presented at 5% and 90% span may indeed reflect accurate measurements of the velocities or it may reflect the gradual degradation of the axial velocity component. Probably both possibilities occur to some extent here; although the proportions of each is not known.

Blade blockage at this location is substantial. It is almost 20% at 10% span and decreases only to 10% at 80% span.

The data acquired at the 52, 10%, and 90% span locations in the enhanced endwall boundary layer configuration is not usable.

5. Station 5, 50% Chord

The velocities measured at this chord location are much better than the measurements at the previous intrablade locations. This is the result of a more favorable blade geometry at this location and a narrowing of the passage due to the convergence of the endwalls.

Enhanced endwall boundary layer data for the 5% span can only be considered reasonable from windows 17 to 35. At this location, the number of measurements in these windows is very small, leading to large uncertainty errors. Notice the large number of measurements at other window locations, i.e., 37-39. The velocities here are very unreasonable, but uncertainties are low due to the large number of measurements. In general, the most reasonable velocity measurements are the ones with the lowest number of measurements. Data at 10% span can be considered reliable from windows 17 to 34. It generally indicates the same structure that is found at 5% span.

The data acquired at 90% span shows reasonable numbers of measurements and uncertainties. Still, the magnitudes of the measured absolute velocities are reduced by almost 50% from data at 80% span. Notice also the large variation in velocities between windows for both the axial and tangential velocities. Measurements at 90% span cannot be considered reliable.

The predicted computed axial velocity field has changed significantly between station 3 and station 5. Both sets of computational results show regions of lower axial velocities near the

shroud suction surface corners and regions of higher velocities near the hub suction surface corners. For both test inlet configurations, the axial velocity gradients in the shroud side of the passage are generally in the radial direction. Both sets of experimental data also show flow gradients that are essentially radial in the outer or shroud side of the passage. Both sets of experimental results also reflect somewhat higher velocities in the hub suction surface corner.

The measured enhanced endwall boundary layer velocities show a small region of higher axial velocities (140 m/sec) near the shroud suction surface corner that is not reflected in the computational prediction.

Although the experimental data of axial velocities does show a great deal of scatter, it generally shows the same flat velocity profile as the predicted computed results with the tendency towards higher velocities in the hub suction surface corner.

The computational results show a larger region of decreasing axial velocities along the suction surface in both cases. This is indicative of the development of a more prominent boundary layer along this surface. This velocity gradient is only faintly apparent in the experimental data. The most likely explanation for this discrepancy is the poorer quality of the data near solid surfaces. It is possible that the blade has been slightly mis-positioned on this plot.

Overall differences in the axial velocity contour plots between the two computed predictions are small at this location.

Station 5, 50% chord, predicted results indicate a continued growth of the endwall effects in the shroud region. For both sets of predictions, the predicted results show a region of increasing absolute tangential velocities in the shroud suction surface corner. Both show a region of lower absolute tangential velocities roughly one third of the span from the shroud and one third of the pitch from the suction surface. Both predictions also show a region of higher absolute tangential velocities in the hub pressure surface region.

However, differences in the two predicted sets of absolute tangential velocities remain minimal at this station which is at 50% chord. Higher absolute tangential velocities are predicted by the baseline results. However, this may be partially the result of the higher wheel speed of the baseline operating condition, which resulted in approximately a 3 m/sec difference at the tip.

A comparison with the baseline experimental data indicates a region of lower velocities in the midpassage region with higher tangential velocities near the blade surface and near the endwalls. On the shroud side of the passage, there are indications of higher tangential velocities near the shroud suction surface corner, where velocities increase from 70 m/sec to 100 m/sec. There are also indications at the 90% span location of higher tangential velocities (approximately 10 m/sec). Also, there does appear to be a region of lower velocities in the tip pressure surface corner of the passage. It is not clear if this is the result of a lower quality of data or reflects the predicted results. the experimental results do indicate a larger region of

relatively constant tangential velocities at approximately 70 m/sec than are indicated by the computed predictions.

The region of low velocities in the hub suction surface corner for both test inlet configurations is obviously the result of spurious measurements. Both sets of experimental results show some indications of higher absolute tangential velocities in the hub suction surface corner.

The measured absolute tangential velocity data at the 90% span for the enhanced endwall boundary layer configuration has been considered questionable. This data, which indicates much lower tangential velocities near the shroud appears to no be usable, since the predicted results clearly indicate much higher velocities in the shroud region.

6. Station 6, 75% Chord

Baseline data at this location appears valid across the passage with the exception of data at the 90% span location. At least 10% of the passage appears blocked by the presence of blades. Of course, this percentage increases nearer the hub.

The axial velocities at 90% span for the baseline case are somewhat lower but the tangential velocities are higher, resulting in much greater turning angles. Consider the large variations in velocity for both axial and tangential components as presented in Tables A1 and A2 of Appendix A and Figs. 20 and 22. Notice the reduction in tangential velocities at 80% span and then an increase at 90% span.

The velocities measured at 80% span appear more consistent with the velocities at other locations.

The baseline configuration axial velocities at 80% span show some interesting features. At 80% span the magnitude of these axial velocities is approximately that of the velocities at 70% span. A close inspection of these velocities reveals some large window-to-window variations. Calculated uncertainties are also much larger than those at 70%, due to the fewer measurements taken at this axial/radial location.

7. Station 7, 95% Chord

Baseline configuration velocities at this location provide data from 10% to 90% span. The measurements at 5% span are not self consistent or consistent with data gathered at more distant radial locations. The large calculated uncertainty values are the result of poorer axial measurements at this location.

The number of valid measurements for the axial 5% span are very low and the velocities are generally much lower. Window 25 shows, perhaps, the only valid velocity at 5% span and this velocity results from only 2 measurements.

Notice the reduced blade blockage at this location, varying from 20% at 10% span to less than 10% at 90% span.

The data at 5%, 10%, and 90% are not usable for the enhanced endwall boundary layer test configuration.

The experimental results show a gradually increasing axial throughflow velocity component from the pressure surface tip corner towards the suction surface corner region. The experimental data also shows lower axial velocities in the suction surface tip region.

Overall, and consistent with the more upstream locations, the velocity gradients for the axial velocities in this location are flatter in both the pitchwise and spanwise directions.

The experimental tangential velocity results are flatter than the predicted results. Both the baseline and the enhanced endwall boundary layer case contain tangential velocities near the suction surface side.

The computed predictions indicate, for both cases, a region of higher axial velocities in the hub suction surface region, as do the experimental results. Measured baseline velocities peak at 140 m/sec, as do predicted maximum velocities. The peak enhanced endwall boundary layer velocities are greater than 130 m/sec but do not exceed 140 m/sec.

The baseline configuration, the 130 m/sec axial velocity contour moves generally from the suction surface tip region towards the hub pressure surface region for both the measured (contour line I) and the computed prediction (contour line N).

The measure enhanced endwall velocities continue to reflect somewhat lower axial velocity gradients over most of the passage than were predicted by the solver.

Both sets of experimental data show lower axial velocities near the suction surface that indicate the continued growth of the boundary layer. However, these gradients are somewhat less prominent in the case

of the baseline data. Both sets of experimental data only hint at the lower axial velocities in the suction surface tip region. However, this is not a prominent feature in the experimental data.

The predicted station 7 absolute tangential velocity contour plots remain very similar. Overall, the most prominent difference between the two test cases is the somewhat higher absolute tangential velocity level of the baseline flow case.

Both sets of computed predicted results show an increasing absolute tangential velocity component from the midpassage toward the suction surface tip region. Both show a rather constant increase in the absolute tangential velocity toward the hub from the midspan location. Both show a large region of increasing absolute tangential velocity inside a significant suction surface boundary layer. Both predicted test cases show much thinner pressure surface boundary layers. Both sets of computed predictions are very similar in appearance.

Experimental measurements of the absolute tangential velocities at the station 7 location show good qualitative agreement with the calculated predictions. Experimental data shows a region of increasing absolute tangential velocities near the suction surface. Also apparent are regions of increasing absolute tangential velocities from mid span towards the hub surface.

8. Station 8, 105% Chord.

Data acquisition was extremely difficult at this location for a number of reasons. First, the presence of the rear edge of the window frame reduced optical signals reaching the PMT. Second, seeding was difficult since aerodynamic spreading of the seed was greatest at this location. Third, blade blockage, even at 105% chord, reduced the quality of data in those windows immediately downstream of the rotor blades. The existence of blade wakes greatly enhanced the spreading of seed particles.

Nevertheless, some data was acquired at this station. Baseline inlet configuration data was valid from 40% span to 80% span. The measurements at 50% span show a slight reduction in the axial velocity. This lower axial velocity is not apparent in the aerodynamic data measurements downstream of the rotor. The measured axial velocities at 5%, 10%, 20%, and 30% are much lower than the velocities measured at greater radial locations. These velocities are of the order of 50 m/sec or less. This trend is not shown in the aerodynamic data downstream of the rotor. The tangential velocity measurements show a fairly consistent variation in velocities from 10% span to 90% span.

Uncertainty measurements for these velocities are also very low.

The baseline tangential velocities appear valid over most of the passage from 20% span to 90% span.

The extremely low axial velocities from 5% span to 30% span generate absolute flow angles of over 60°. These flow angles do not agree with the acquired aerodynamic data. Therefore, it appears likely that these axial velocities are in error.

The enhanced endwall boundary layer data at station 8 was of poorer quality than the data acquired at station 8 during the baseline configuration test. Extremely low axial velocities at 10%, 30%, 40%, and 90% span, coupled with large window to window variations and calculated uncertainties make axial velocity measurements at this locations highly suspect. The extremely high axial velocities at 20% span are not physically possible.

The absolute tangential measurements at 5%, 10%, 20%, 30%, 40%, and 90% span show large window-to-window variations, large negative velocities, and numerous windows containing no measurements. The signal was stronger for the tangential component and it was acquired to 20% span with the baseline configuration. The enhanced endwall boundary layer test case data was only valid for the 50%, 60%, and 70% span. It is most probable that for the enhanced endwall boundary layer test case that the only trustworthy data exists from 50% to 80% both for the axial and the absolute tangential velocities.

9. Summary

The quality of the measurements acquired during this experiment was strongly dependent upon the location at which they were acquired.

Overall data quality deteriorates the farther downstream into the flow passage that the measurements were taken. Data acquired at station 8 is usable only in the outer half of the passage at best.

Data acquired near passage boundaries is also of lower quality than the data acquired at mid-passage. The data acquired at 5% and 90% span is generally marginal.

Blade metal blockage greatly increased the areas of the passage where velocities could be measured near blade surfaces. Since the beam crossing angle is only approximately 4°, increased areas of poor data near the blade surfaces must be the result of the blockage of signal reflected from the particles towards the collecting optics lens.

An overall summary, including those axial/radial locations at which data is not usable was presented earlier in Table X.

Qualitative agreement between the computed predicted results and the measured experimental data is good. However, quantitative agreement between the two sets of results can be improved.

Unfortunately, velocity measurements could not be obtained near the blade surfaces or in the endwall regions. As a result, the predicted velocity gradients off the suctions surface and in the endwall regions could not be observed.

C. COMPARISON OF RESULTS IN THE STATIONARY REFERENCE FRAME

In this section, the inlet and exit flow fields will be discussed in some detail. As was mentioned above, these flows are best considered in a stationary reference frame. This is because the components both upstream and downstream of the rotor are stationary and their interactions with the flow are in an absolute reference frame.

Figure 24 includes line plots of the absolute total velocities measured at the inlet (station 1), 95% chord (station 7), and exit (station 8) locations for both test flow conditions. The station 7 data is presented here since the station 8 information is of such poor quality. Station 7 is at 95% chord and is, therefore, very near the trailing edge of the blade. Therefore, it will be used when considering the exit station flow.

A comparison between the absolute flow velocities measured at the inlet shows that the baseline case is very flat between 20 and 80% span. Here the velocity variations of the axial velocities are of the order of 5 m/sec. The enhanced endwall boundary layer varies over 15 m/sec in this range of span. Thus, the baseline variation represents about 5.5% of the average inlet velocity vector while the thickened endwall boundary layer variation is of the order of 15.5%.

At Station 7, 95% chord, the baseline total velocities vary from 167 m/sec to 144 m/sec from 20 to 80% span while the enhanced configuration shows a change from 165 m/sec at 20% span to 147 m/sec at 80% span. These numbers are approximate and are taken from the window

25 readings. These profiles are more graphically displayed in Fig. 25. This figure includes plots of the mid-pitch window (window 25) absolute total velocity at stations 1, 5, 7, and 8. Here the baseline configuration changes in the spanwise direction 25 m/sec while the enhanced endwall boundary layer profile is somewhat shallower at 20 m/sec.

Differences in the shape of the inlet velocity profiles have become very similar by station 7, 95% chord.

Both the station 7 and station 8 line plots are difficult to interpret. The baseline case shows an obvious gradient increasing from the blade tip pressure surface corner to the hub suction surface corner. The change in total velocities is less than 25 m/sec. The enhanced boundary layer condition shows a similar gradient but it is much less apparent due to the quality of the data. It also shows a change in total velocities of about 20 m/sec. Both flow cases have very low blade to blade velocity gradients. This is indicative of the low blade loadings that result in the relatively low pressure rise across this rotor at 60% speed.

Overall, for both flow conditions, blade-to-blade variations seem to disappear in the blade tip region. Somewhat nearer the midspan, very shallow gradients from suction surface (high values) to pressure surface (low values) exist.

Only a small region between 60 and 80% span present any usable data at station 8. In this region, only small differences in velocity magnitudes and overall flow appearances between the two flow test cases

can be seen. Station 8 data in this small region of the outer passage appears relatively flat from blade to blade. The only information that reliably extends beyond the 60% span location towards the hub are the baseline absolute tangential velocities. This data shows higher absolute tangential velocities in the hub suction surface corner.

Of course, total velocity is only part of the picture. The absolute flow angle must also be considered in comparing the differences between the two inlet flow conditions. Figure 26 shows the absolute flow angles at stations 1, 7 and 8. Figure 27 contains the midchannel window plots of the absolute flow angles at these stations.

The angle plots present essentially similar profiles at the inlet and exit with a slightly steeper angle gradient at station 7 for the baseline (approximately 0.6°).

Station 7 absolute flow angles, for both the baseline and the enhanced endwall boundary layer case contain large midpassage regions where the absolute flow angles are between 30° and 35°. Both have regions of slightly lower turning (less than 30°) at the 70% span location.

Notice that the region of slightly lower turning is somewhat larger for the enhanced configuration. Overall, the flow in the outer half of the enhanced passage is turned slightly less than that in the baseline configuration while the flow in the inner or hub half has very nearly the same turning as the baseline configuration.

Because the data acquired at station 8 is largely unusable, direct observation of much of the exit flow is not possible. The station 7 location is near the exit and the following statements can be made concerning the flow at this location. It is appears that the crosschannel variations both blade-to-blade and hub-to-shroud are small with absolute total velocities varying only 20 m/sec and flow angles varying only 5°. In general, the regions of highest measured turning and greatest velocity are in the hub suction surface corner. The regions of lowest measured total velocity and lowest measured turning are near the tip and in the tip pressure surface corner.

The overall design intent of this rotor is to provide a constant exit flow angle of about 50° and a gradually decreasing total velocity from hub to shroud. It should be noted that this is at design speed and mass flow. At the part speed operation of the present test, the rotor did maintain the decreasing velocity profile for both configurations. It did not, however, provide a constant exit flow angle for either test configuration.

It appears that the differences between the two inlet velocity profiles are reduced at the exit of this rotor and are primarily confined to the outer half of the passage. Overall, the enhanced endwall boundary layer configuration contains velocities that are slightly lower than the baseline configuration. The blade-to-blade velocity gradients appear smaller in the outer half of the passage. Lastly, the enhanced

endwall boundary layer case turning is slightly smaller in the outer half of the passage by 2°.

D. COMPARISON OF RESULTS IN THE ROTATING REFERENCE FRAME

1. Introduction

While the absolute reference frame is the most useful for flows upstream of the rotor, where the inlet conditions are determined, and downstream of the rotor, where the stators and combustors are found, the most useful reference frame for designing and studying the rotor itself is the relative reference frame. In this reference frame, the rotational speed of the rotor is vectorially added to the fluid velocity.

This section will consider the flow in the relative reference frame. A comparison of the two test cases in this reference frame should help show why the differences between the two test cases becomes less prominent as the flow moves through the rotor.

The flows will also be reviewed to ascertain any other prominent flow features that have developed in the passage.

2. Inlet flow features.

Figure 28 shows the relative total velocities at each station and Fig. 29 shows the relative flow angles. These colorfill charts, which have not been smoothed, can give a better feel for the qualitive changes in flow features. However, color reproduction is somewhat limited and some of the more subtle flow features can be lost. Figures 30 and 31 provide the same information for the inlet station, station 1, presented as line plots.

When viewing these plots the most striking feature about the flow in the relative reference frame is the similarity of the flows at station 1. Moving from the absolute to the relative reference frame reduces the severity of the differences between the two test cases because the addition of the large tangential velocity component due to wheel speed reduces the effects of the differences in axial component on the total relative velocity. Still, these differences have not completely disappeared.

Relative total velocities for both inlet cases vary from about 210 m/sec to about 270 m/sec and a plot of the midpitch relative total velocities shows little effective difference between the two test cases (Fig. 32).

Relative flow angle plots show slightly more prominent differences at station 1. Here the baseline configuration shows a greater relative flow angle at the inlet along the outer or shroud half of the passage than the enhanced inlet. The midpitch (Fig. 33) plots of flow angle

differences at station 1 reveal differences of less than 3°. The enhanced endwall boundary layer configuration showing larger relative flow angles near the endwalls and the baseline case contains greater relative flow angles at midspan. It would be more appropriate to examine the incidence angle shown in Fig. 34. This figure shows the relative flow angle differences from the input grid. This grid, an "H-Grid", was generated by dividing the blade passage into 11 constant span, axisymmetric surfaces. Each of these surfaces was further divided into a grid with equally spaced lines in the meridional direction and equally spaced lines between the passage boundaries. This input grid is shown in Fig. 35. These angle differences will be explored more completely in a later section that will deal with the secondary flows generated in these cases. However, upstream of the rotor, these differences are equivalent to the blade leading edge incidence angle. As would be expected, the baseline configuration shows a much smaller variation in incidence than the enhanced configuration. Variations in the baseline are from 15° at 5% span to 11° at 60% span. The enhanced endwall condition shows variations from 19° at 5% span to 9.5° at 60% span.

The measured design speed inlet incidence angles, Fig. 36a [75], are much lower at 6°. The design is with a uniform inlet flow and is at design speed and at design mass flow. Figure 36b, from Reid and Moore [77], shows the peak efficiency incidence angles measured at 60% design speed.

Overall, then, a much greater incidence angle occurs at all spans for both inlet flow test cases. Obviously, this requires a greater turning of the flow as it initially enters the passage.

The baseline case requires a more constant turning along the span than the enhanced case where the incidence is substantially larger at the endwalls than at midspan. In fact, the required turning at the blade inlet for both cases is of the same magnitude as that which occurs through the remainder of the passage.

Thus, at the inlet of the rotor, the velocity profiles are very similar due to the addition of the rotational component. If the flow successfully negotiates the initial turning, the differences between the two test cases have been greatly reduced. Further, the initial flow angles into the rotor passage would tend to oppose the development of the classical passage vortex with the endwall flows moving towards the blade pressure surfaces and the midspan flow more in the direction of the suction surface.

Station 3 is situated at the 5% chord location. This is downstream of the blade leading edge and is the farthest upstream location at which data was acquired inside the passage. The quality of the data acquired at this location is poorer than many of the results acquired farther downstream in the passage, particularly near the endwalls and the blade surfaces. Nevertheless, it does provide some information concerning the condition of the flow immediately downstream of the blade leading edge.

A review of the relative flow angles (Figs. 29 and 31) and the absolute flow angles (Fig. 26) at the 5% chord location shows that the

flow near the midpitch of the channel has not yet been affected by the presence of the blades. This can be seen by the region of very low absolute turning angles near the midpitch location.

The negative flow angles near the suction surface indicate that the flow is being accelerated and turned by the presence of the blade leading edge. Conversely, the large positive flow angles reflect the deceleration of the flow along the pressure surface.

Direct comparison reveals that there are definite differences between the two test flow cases. The enhanced test case shows generally more positive flow angles on the hub side of the passage than the baseline case, while the reverse of this is the case in the mid passage region. The shroud side of the passage, specifically the 80 and 90% span locations, shows no clear difference between the two flows.

The relative flow angles reveal that, overall, in the midpitch region away from the blade surfaces, the enhanced boundary layer flow is somewhat less turned (>5°) than the baseline flow. This is a direct result of the lower incidence angle of the enhanced boundary layer case in the midspan region. The higher relative flow angles near the hub are the result of the lower axial velocity and, thus, the higher incidence angle of the enhanced boundary layer test case in this region. While the same effect may occur in the tip region, the differences are not clear cut. This may be the result of the quality of the data acquired here or the effects of the tip clearance.

In summary, the rotational effects of the rotor reduce the differences between the total velocity profiles at the rotor. Further, the rotational effects result in a difference in the incidence angles presented to the rotor by the two test cases. The differences are higher incidence angle region near the endwalls and lower incidence angles near midspan for the enhanced endwall boundary layer case. This incidence variation of the enhanced endwall case results a flow condition where the midspan flow of the enhanced case contains a less prominent movement towards the pressure surface of the passage than the baseline condition and, conversely, a more prominent movement of the flow towards the pressure surface by the flow near the hub and possibly near the shroud in the enhanced endwall flow case.

It should be noted here that in the classical development of the passage vortex, the flow situation is reversed. In that instance, the midspan flow has a more prominent movement towards the pressure surface than the endwall flow.

3. Exit Flow Features

Due to the poor quality of the data acquired at the exit station, station 8, most of the observations of the exit flow features must be made at the 95% chord location, station 7.

First, the effects of the rotor on the relative total velocities and flow angles will be considered.

Relative total velocities and relative flow angles at station 7, in those areas where the data is acceptable, are very similar. The two sets of data reflect no important differences. However, the enhanced endwall configuration shows a slightly more parabolic profile between the endwalls with the midspan velocities about 5 m/sec higher. Figure 32 shows the midspan relative velocity gradient from hub to shroud and Figure 30 shows the cross channel line plots. The flow reflects little variation in the relative total velocities blade-to-blade with the highest velocities in the midpitch region and slightly lower velocities near the blade surfaces.

Figure 33 presents the midpitch window relative flow angle differences for the two flow configurations. These represent the differences between the relative flow angle and the input computational grid shown in Fig. 35. This grid reflects what is considered the direction of the primary through flow in this investigation, i.e., the relative flow is assumed to be along the grid lines that run from inlet to exit. The grid is generated in an axisymmetric coordinate system.

As was mentioned in the previous section, the inlet profiles have lower incidence angles at midspan and higher incidence angles near the endwall for the enhanced endwall configuration as compared to the baseline configuration.

The differences at inlet were about 3.25° between 25 and 50% span and over 1.65° from 50 to 80% span. This demonstrates the large differences in inlet velocity profiles.

At 95% chord, in contrast, the midspan "bulge" appears to be largely gone with the change in difference between 20% span and 50% span only about 0.75° at window 25. Generally, the differences between the two sets of flow angles is smaller at 95% chord with the greatest differences near the hub. Overall, the enhanced endwall boundary layer shows somewhat less turning than the baseline case with the greatest discrepancies near the hub. Nonetheless, the overall character of the flow angles is similar.

To summarize these results, it can be said that the inlet velocity differences are minimized in the rotating reference frame by the vectorial addition of the wheel speed. The total velocity and flow angle differences are further reduced as the flow moves through the rotor.

The magnitude of the differences in the total relative velocity profile has been reduced for both inlet profiles because of the addition of the rotational component. The reduction has been most striking for the enhanced endwall case.

The enhanced inlet relative flow angles have lost the parabolic distribution that they entered the passage with. Instead, both test cases show a gradual and somewhat linear decrease in flow angle differences from hub to shroud. In any event, the magnitude of the spanwise differences in flow angles has been greatly reduced in both the baseline case (from 3.2° to 1.3°) and in the enhanced endwall boundary layer (from 6.3° to 3.0°).

The relative flow angle cross channel plots, Fig. 31, display indications of greater turning along the suction surface corner. This is only slightly apparent for the enhanced case but is more prominent in the baseline case. Overall, however, the blade-to-blade relative flow angle gradients are smaller across the entire passage.

4. Additional Flow Features

a. Introduction

A primary goal of this study was to enhance the understanding of the flow physics underlying the generation of the classic passage vortex inside a compressor rotor. Before turning to a more detailed review of the results concerning this goal, a few additional observations of the flow fields in question will be made.

b. Exit Flow Rotational Effects

A flow feature that is apparent in the plots is the addition of a rotational component to the exit flow for both inlet flow configurations. Figure 22 presents lineplots of the absolute tangential velocity. As was mentioned earlier, the small tangential component at inlet is an error caused by the data acquisition system. It was not measured by the aerodynamic probes and there is no physical reason for

the existence of such a tangential velocity. The flow at the inlet is axisymmetric and has only an axial flow component.

As shown in Fig. 22, larger tangential velocities can be seen developing nearer the hub with the greatest velocities in the hub SS corner at station 6, 75% chord. This is also apparent at the 95% chord location. Tangential velocities were acquired for the baseline configuration at station 8 across most of the passage. This is the only velocity component successfully acquired over the inner half of the passage at this station and it also shows higher velocities near the hub.

The absolute flow angles at station 7 (midpitch values are shown in Fig. 27) and the aeroprobe measured flow angles display roughly the same trends in flow angle variation from hub to shroud. Both sets of data demonstrate a decreasing flow angle from the hub towards the midspan. The aerosurvey probes reveal a very gradual increase in absolute flow angle, while the laser anemometer survey data indicates a decrease until near the tip. The LA survey data is along a midpitch window while the aerosurvey data represents an average exit flow condition. The aerosurvey measurements contain values of the flow near the blade surfaces where turning is greater.

In summary, the exit velocity profiles include a clockwise rotation in both test conditions. This is shown by the higher flow angles near the hub that decrease towards the shroud. As was previously mentioned, both tangential and axial velocities decrease from hub to shroud.

Interestingly enough, in the rotational reference frame the added rotational component is counterclockwise. Figure 31 includes the line plots of the relative flow angles and Fig. 29 has the colorbar plots. Station 7, 95% chord, reveals increasing flow angles from the hub to the shroud. The baseline configuration indicates slightly lower relative flow angles in the suction surface corner. This feature is not quite so obvious in the enhanced endwall boundary layer configuration.

c. Radial Flow Migrations

Figure 37 shows the total relative velocities for the first 15 windows away from the suction surface towards the pressure surface.

Data for five chord positions, both inlet configurations, and for five spanwise locations (10%, 30%, 50%, 70%, and 90%) is presented. These total relative velocities have been nondimensionalized by dividing them by the velocity in window 15.

The data shown at 5% chord at all of the spanwise locations only extends part of the distance to window 1 because the blade blockage is included in windows 1 through 7.

Baseline configuration data at 90% span indicates a great deal of scatter and therefore it is difficult to observe any trends at this span location except at the 95% chord location where a velocity gradient is apparent for both flow configurations.

Most noticeable here for the baseline configuration is that the effects of an adverse pressure gradient over the suction surface are not apparent until 95% chord near the hub.

As the span increases towards the rotor tip, the effects become noticeable earlier in the flow. At 30% span, the effects seem to begin by 75% chord. At 50% and 70% span the effects are apparent at 50% chord and possibly by 25% chord. At 70% span, the effects are very prominent by 75% chord.

The enhanced configuration at 10% span actually shows usable data only at 75% chord. At this location, it appears that there are no effects from an adverse pressure gradient.

However, as opposed to the baseline configuration, all remaining span locations of the enhanced flow configuration show the effects of the pressure gradient. It is apparent by 25% chord at all span locations and is very prominent by 75% and 95% chord. A likely cause of this is the migration of the blade boundary layer due to the enhanced endwall boundary layers.

E. GRID VELOCITY DEVIATIONS

The major objective of this investigation was the observation and analysis of the development of the passage vortex that is thought to be developed in a turning passage from the spanwise shear. The classic passage vortex is created in a turning passage such as a compressor rotor by the spanwise total pressure gradient (velocity gradient).

While this vortex is generated by the velocity gradients that exist in the endwalls of both test configurations, it should be more pronounced in the enhanced endwall boundary layer configuration because of the more prominent velocity gradients generated by the insertion of the distortion screens upstream of the rotor.

Since the total velocity vector could not be resolved because radial velocity information was not obtained, this section must be confined to an examination of the "velocity deviations" that can be calculated from the two velocity components that were measured.

The velocity deviations from the grid lines provide an indication of the development of this passage vortex.

There are a number of different specific definitions for the term "secondary flow." It is generally defined as the difference between the measured three dimensional flow and some "primary flow." For the purposes of the investigation, the primary flow field is considered the flow along the generated primary computational grid (Fig. 35). This has been used primarily because of the ease of generation of this field and the uncertainties in the definition of the term "primary flow" field. In fact, since the rotor is operating subsonically and it is far from the measured stall point, this velocity field should be near any actual potential flow field.

Therefore, the tangential velocity deviation was calculated for the two inlet flow configurations and compared instead to assess the development of the passage vortex.

Consider Fig. 38a. The relative velocity can be written as [92]:

$$W_0 = W_{p0} + W_{s0} \tag{40}$$

where $()_{p} = primary flow,$

() = secondary flow, and

$$W = \sqrt{W_0^2 + W_z^2} \tag{41}$$

The grid slope is:

$$Tan\psi = r\frac{d\theta}{dz} \tag{42}$$

and the relative flow angle is:

$$\beta = \tan^{-1}(W_0/W_x) \tag{43}$$

The calculation develops as:

$$W_p = W\cos(\beta - \psi) \tag{44}$$

$$W_{p\theta} = W_p \sin \psi = W \sin \psi \cos (\beta - \psi) \tag{45}$$

$$W_{s0} = W_0 - W_{p0} (46)$$

and, finally,

$$W_{s\theta} = W_{\theta} - W \sin \psi \cos (\beta - \psi) \tag{47}$$

A second calculation can be made for the radial or meridional secondary velocity. Considering Fig. 38b it is seen that

$$W_r = W_{pr} + W_{gr} \tag{48}$$

$$W = \sqrt{W_x^2 + W_x^2} \tag{49}$$

$$Tan\phi = \frac{dr}{dz}$$
 (50)

$$\alpha = \tan^{-1}(W_r/W_g) \tag{51}$$

$$W_{p} = W\cos(\alpha - \phi) \tag{52}$$

$$W_{pr} = W_p \sin \phi = W \sin \phi \cos (\alpha - \phi)$$
 (53)

and

$$W_{ar} = W_r - W \sin \phi \cos (\alpha - \phi)$$
 (54)

The grid used in these calculations is shown in Fig. 35.

Figure 39 presents line plots of the calculated flow deviation velocities, $W_{s\theta}$, at each of the axial locations where data was acquired.

Line plots of relative flow angle differences, which are the differences between the relative grid and the relative flow angles, are shown in Fig. 34. As may be seen at Station 1, the inlet station,

relatively flat velocity deviations are obtained for both configurations. The positive velocities represent a fairly large positive incidence for both test configurations. This is consistent with the flow condition selected for this test. The positive deviation velocities indicate that the flow has not turned far enough to be parallel to the grid lines.

The inlet baseline deviation velocities are relatively constant across the passage. The spanwise variations tend to reflect variations in the grid extrapolations upstream of the rotor. These extrapolations reflect the three-dimensionality of the blades themselves.

A listing of the difference between the two inlet velocity deviations, $V_{\tt sec.b}-V_{\tt sec.a}$ is given in Appendix A, Table A9.

Figure 40 is a three-dimensional plot of the velocity deviation differences. When looking at the plots in Fig. 40, the rotor hub is on the left side and the compressor shroud will be on the right. The blade suction surface is towards the back of the plot and the pressure surface is towards the front.

For these plots, no differences in flow deviations that were greater than 25 m/sec were permitted. This was done to improve the scaling and readability of the plots. It was determined that differences larger than 25 m/sec were unrealistic.

The first plot shows the inlet survey location. The differences in the velocity deviations is very noticeable with the baseline configuration having the greatest deviations at midspan and the enhanced configuration having larger deviations at the endwalls. This is a

manifestation of the thickened endwall boundary layers of the enhanced test configuration.

The differences in the deviation velocities at the inlet are very noticeable, but are small. The enhanced configuration is 6 m/sec larger near the hub and about 5 m/sec larger near the midspan. At the 90% span line, the differences are much smaller with the enhanced, about 1 m/sec larger at the 80% span location.

At station 7, the 95% span location, the character of the differences has changed with the enhanced velocity differences larger at all span locations. Rather than having the largest discrepancies between the enhanced and the baseline at the endwalls, now there is a gradual decrease in the differences from the hub to the shroud. The enhanced configuration shows a deviation velocity about 6 m/sec larger at 20% span and this decreases to near 0 at 90% span.

It was mentioned earlier that the character of the relative flow angle differences also changes from inlet to exit. At the inlet, the enhanced endwall configuration shows greater incidence than the baseline configuration and the enhanced mid-span region shows a lower incidence than the baseline. At station 7, the hub endwall shows this same difference with the enhanced configuration showing larger relative flow angles than the baseline. The enhanced endwall boundary layer is approximately 2° greater. This is nearly the same magnitude as the difference at the inlet of the rotor. However, by mid-span at the inlet, the baseline incidence is greater than the enhanced endwall by

approximately 2°. At station 7, the enhanced configuration shows the greater deviation from the grid by about 2°.

The difference in the relative flow angles is much smaller near the tip region at the inlet of the flow with the enhanced configuration showing about 0.5° higher incidence at the 80% span than the baseline configuration. It is difficult to estimate such a small difference but it appears that this 0.5° difference has disappeared by 95% chord, station 7.

These results can be summarized as follows. In the rotating reference frame, the enhanced endwall boundary layer configuration flow contains a prominent clockwise rotational component in the inner half of the passage that is not apparent in the baseline configuration inlet flow. There is also a weaker counterclockwise flow in the outer third of the passage of the enhanced endwall flow that does not exist in the baseline configuration. It is, however, weaker due to the larger wheel speed in the outer half of the passage.

In both cases, due to the wheel speed, the relative total velocity increases from hub to shroud at inlet. This gradient is slightly larger for the enhanced case than for the baseline case along the inner half of the passage and slightly less for the outer half of the passage.

By the 95% chord location, the character of the flows has changed in the following way. In the inner half of the passage, the enhanced boundary layer configuration turns about 2° less than the baseline configuration with that number slightly higher at the hub. In the outer

half of the passage, the difference in turning essentially goes to zero by the 80% span location.

The differences between the flow deviation velocities at the 95% chord location in the inner or hub side of the passage are essentially constant at 4 m/sec. In the outer half of the passage the differences decrease to essentially zero by the 80% span location.

Thus the differences at exit between the flows are constant along the inner half of the passage and tend towards zero in the outer half of the passage.

During the development of the classic passage vortex, larger velocity deviations and deviation angles exist in the mid-span region of the passage as the higher energy fluid in this region tends to move towards the pressure surface. Correspondingly, the endwall flows would tend to move towards the suction surface in the passage, resulting in lower velocity deviation angles.

In the test, the addition of the wheel speed has greatly reduced the magnitude of the differences in the inlet velocity profiles and, at least initially, have added rotational effects to the inlet flows that are opposite to the classic passage vortex. The result is that near the exit, the secondary velocities and relative flow angle differences are constant along the inner half of the passage. In the outer half or shroud side of the passage, the enhanced endwall boundary layer configuration develops slightly larger velocity deviations than the baseline configuration. This suggests that a weak passage vortex is developing in the outer half of the passage for the enhanced endwall

boundary layer configuration. Note, however, that the differences in relative flow angles at this location are only 1° or so, indicating a very weak passage vortex at this location if any exists at all.

Figure 43 presents cross channel plots of the differences of the relative flow angles, β , for the computed predictions. For these plots, the enhanced endwall boundary layer relative flow angles were subtracted from the corresponding relative flow angles for the baseline configuration. These plots show the differences in the flow directions for the two sets of predictions.

Station 1 in Fig. 43 is immediately downstream of the rotor leading edge at approximately 2% chord. The remaining stations are at the equivalent locations that the experimental results were acquired at; these locations are 5, 50, 95, and 105% chord.

A classic passage vortex flow would appear as a flow containing a region of lower turning or greater relative flow angles in the center region of the passage and regions of greater turning near the endwall regions where low momentum fluid would be found. The station 1 figure indicates the opposite trends where the larger positive differences indicate the enhanced endwall boundary layer flows are moving more towards the suction surface than the baseline configuration. Further, the negative numbers in the endwall regions indicate the enhanced endwall boundary layer flow is moving more towards the pressure surface than the baseline configuration. These movements are opposite the movements that would be expected in the classic passage vortex.

Notice that the variations are most prominent on the hub side of the passage with the differences going from +1.5° near midspan to -1° near the hub. The variations are much less prominent on the shroud side of the passage with the differences only varying down to 0.5°. this may be due to the increasing radius which results in higher relative tangential velocities and correspondingly greater relative flow angles.

By station 5, 50% chord, the differences between the two flow predictions have been greatly reduced with the variations between 0° and -1° over most of the passage outside of the boundary layers.

At station 8, predicted relative flow angle differences are less than ±1° over the entire passage, with the exception of small regions near the endwalls and in the blade wakes. There is little indication of the development of a passage vortex anywhere in the predicted flow fields.

VI. SUMMARY AND CONCLUSIONS

The purpose of this investigation was to assess the development of the passage vortex inside a high speed compressor rotor. This was done by obtaining a detailed map of the flow field inside a high speed rotor operating subsonically for two different inlet flow field configurations. One flow field configuration consisted of an "undistorted" inlet velocity profile and the other configuration consisted of a parabolic inlet profile generated by placing an axisymmetric distortion screen on the hub and on the shroud well upstream of the rotor.

The two flow fields were evaluated and compared. The similarities and differences were noted and the following statements can be made:

- (1) The differences between the two inlet velocity profiles tend to wash out as the flow moves through the rotor.
- (2) There is a clockwise rotation added to the absolute flow as it moves through the rotor. This rotation is counterclockwise in the relative reference frame.
- (3) The total velocity, axial velocity, and tangential velocity all are greatest near the hub and decrease as the flow is observed further towards the shroud.
- (4) There are some indications of the development of a passage vortex at the 95% chord location when the differences between the calculated flow deviations are calculated. Computational predictions indicate little or no passage vortex development for the test conditions examined in this project.

This investigation marked the first time that a detailed evaluation of the effects of inlet flow field distortion on the internal flow field of an axial compressor rotor was made. The results provide significant information to aid the compressor designer by indicating that the generation of a passage vortex due to thickened endwall boundary layers of the type normally found at the middle stages of an axial compressor has only minor effects on compressor rotor performance.

VII. RECOMMENDATIONS

There are a number of continuations to the present research that could be pursued in order to obtain a better understanding of secondary flows and the passage vortex. The following is a list of suggested efforts:

- (1) The laser anemometer system should be improved to allow the acquisition of velocity measurements nearer the endwalls and blade surfaces. It is in these regions that many secondary flow effects become most apparent.
- (2) The laser anemometer system should be improved to allow the acquisition of velocity measurements at higher operating speeds. It was not possible to acquire measurements at design speed with this laser anemometer system. Yet, the effects of supersonic flows and the associated shocks can be significant causes of secondary flows.
- (3) The measurement of the third, or radial velocity, component has been accomplished for this rotor in a follow-on effort. Unfortunately, scheduling conflicts allowed the measurement of only one inlet configuration, the baseline configuration. This information needs to be reduced and measurements for the AEBL configuration need to be acquired.
- (4) Measurements, including radial velocities, need to be made at other flow conditions on the operating line. Certainly near the stall point the increased blade loading will encourage additional secondary flows.

- (5) Stronger inlet velocity gradients need to be observed.

 These stronger gradients should enhance the development of the passage vortex and will allow easier observation and analysis of the resulting flows.
- (6) Experimental data acquired during this project should be closely reviewed to assess the possibility of improving the experimental results with additional post processing of the data.
- (7) Laser Anemometer measurements need to be made much closer to the blade leading edges to observe the flow physics occurring as the flow turns to enter the passage.
- (8) Additional computational results must be done to more accurately map the predicted operating line and thus more closely match the computational operating point and the measured operating point.
- (10) Inclusion of a tip clearance model is important. This model will be included in the solver in the first quarter of 1991 and should be used to assess the effects of tip clearance on the flow.

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TABLE 1. - PERFORMANCE PARAMETERS OF SEVERAL COMPRESSOR RESEARCH FACILITIES

Organization	Type of compressor	iniet tip diameter, m	Hub/ lip ratio	Approx- imate inlet velocity, m/s	Apprex- imate maximum, rpm		umber blader		Maximum tip speed, m/s	Maximum pressure rise	Types of rotor measurements
The Pennsylvania State University	Low speed stage	0.9398	0.5	30	1700	45	21	25	61	≃●	Various conventional - hot wire, temperature total and static pressures inside and downstream or roter
L.H. Smith (1955); John Hopkins University	Low speed stage	0.9398	0.5	30	1250	45	21	25	61	~•	Pitchwise sverage downstream of rotor, total pressure
LeBoerf Ecolg CG- WTRAL De Lyon; France	Transonic stage	0.55	6.78		12000	•	•	•	345	1.38	Pitchwise average downstream of rotor using conventional pressure and temperature
United Technologies Research Center	Low speed rotor	1.52	0.8	35	510		28		40	≃•	Rotating pressure probe provides circumferential surveys of wake. Static pressure readings on blude surfaces
General Electric, Cincinnati	Four low speed stages	1.524	●.7	<u></u>	750	•	38 to 96	37 10 94	61	~•	Pitchwise overage surveys behind rotor wake
Lewis Research Center	Transonic stage	0.505	0.7	75	17200	•	36	٠	450	1865	Conventional pressure and temperature upstream and downstream, conventional and secondary flow LDV
Hunter Cumpaty	Low speed rotor	1.524	0.4		525	•	22	•	42	~•	Hot wire tip clearance measurements in the wake region

	COMPONENT LIST F	TABLE II FOR LASER ANEMOMETER SYSTEM
NUMBER	COMPONENT	FUNCTION OR SPECIFICATION
(1)	Lexel Laser	Green line:514.5 NM Argon Ion
(2)	90° Turning Mirror	Turns beams from horizontal to vertical
(3)	90° Turning Mirror	Turns beams from vertical to horizontal
(4)	Collimator	Collimetes Bearn
(5)	Polarization Rotator	Two Quarter Wave Plates. Optimum beam polarization,
(6)	Offset Beam Splitter	Splits Beam into two. One at centerline, one at 135°. Optimum spnot achieved.
(7)	Polarization Rotator	Half wave plate orients center beam polarization for beam splitter
(8)	Beam Splitter	Center beam split into beams at 225° and 45°, 50mm apart. Optimum power split not achieved
(9)	Bragg cell w/ compensation wedge	135° beam shifted by 40 Mhz. Path length compensation require for remaining beams.
(10)	Polarization Rotator	Half wave plate to properly orientate the shifted beam.
(11)	Beam Stop	Blocks Extraneous beams created by Bragg shifting
(13)	Beam Pair Selector	Remotely actuated beam stop selects either horizontal or vertical beam pair.
(14)	22 mm beam spacer	Spaces beams to a separation of 22 mm.
(15)	Beam Expander	Expands beams by 2.27X
(16)	13 mm Beam Spacer	Spaces beams to a separation of 13 mm.
(17)	90° Turning Mirror	2.54 cm mirror turns transmitted beams 90° into goniometer mirro
(18)	90° Turning Mirror	10.16 cm mirror turns transmitted beams and collected light 90°
	Goniometer Cradle	into probe volume. Directs transmitted beams off radial to minimize blade shadowing
(19)	Focusing Lens	122 mm focal length, 7.62 cm diameter (Transmitting and Collecting).
- (20)	90° Turning Mirror	7.82 cm diameter (Collecting Optics).
(21)	Orange Pass Filter	Blocks all light except in the fluorescent range.
(22)	Focusing Lens	160 mm focal length, 7.62 cm diameter (Collecting Optics).
(23)	Pinhole and Photomultiplier Tube	Mounted on a remotely adjustable base.

TABLE III LASER ANEMOMETER PROBE VOLUME SPECIFICATIONS					
	DESIGN CRITERIA*	MEASURED THROUGH FLOW**	MEASURED TANGENTIAL FLOW**		
LENS FOCAL LENGTH (f)	122 mm	122 mm	122 mm		
BEAM SPACING	9. mm	8.832 mm	8.295 mm		
PROBE VOLUME HALF CROSSING ANGLE (#)	2.156*	2.073*	1,947*		
FRINGE SPACING (d _p)	6.838× 10 ⁻³ mm	7.112× 10 ⁻³ mm	7.572× 10 ⁻³ mm		
BEAM WAIST DIAMETER (d _{e-2})	2.713× 10 ⁻² mm	2.713× 10 ^{°2} mm	2.713× 10 ⁻² mm		
PROBE VOLUME DIAMETER (d _m)	2.715× 10 ⁻² mm	2.7148× 10 ⁻² mm	2.7146× 10 ⁻² mm		
PROBE VOLUME LENGTH	0.721 mm	0.75 mm	0.799 mm		
FRINGE VOLUME (V _{FR})	2.782× 10 ⁻⁴ mm ³	2.892× 10 ⁻⁴ mm ³	3.079× 10 ⁻⁴ mm ³		
NUMBER OF FRINGES (N _{FR}) UNSHIFTED	9	11	11		

All values are calculated using the measured crossing angles and a separation between the two parallel incoming beams of 9mm.

^{**}All values are calculated using the measured crossing angles and assume a focal length of 122mm with parallel incoming beams

TABLE IV LOCATION OF CONVENTIONAL SURVEY PROBE INSTRUMENTATION					
SURVEY STATION NUMBER	AXAL LOCATION* (cm)	NUMBER OF PROBES	NUMBER OF INNER WALL STATICS	NUMBER OF OUTE WALL STATICS	
-1	-30.48	2	2	2	
0	-15.24	2	2	2	
1	-2.568	2	2	2	
2	4.859	2	2	2	
3	10.64	1 circum- ferential 1 radial	2	2	
4	15.24	4	4	4	

TABLE V SHROUD STATIC TAP LOCATIONS					
POSITION NUMBER	% TIP CHORD	AXIAL POSITION (cm)*	POSITION NUMBER	% TIP CHORD	AXIAL POSITION (cm)
1	-200	-5.295	12	50	1.325
2	-173	-4.583	13	60	1.59
3	-120	-3.179	14	70	1.854
4	-60	-1.590	15	80	2.119
5	-20	530	16	90	2.384
6	-10	265	17	100	2.649
7	0	0.	18	140	3.709
8	10	0.265	19	160	4.239
9	20	0.530	20	200	5.298
10	30	0.795	21	223	5.908
11	40	1.080			

Flow (Kg/sec)	±0.3
Rotative Speed (RPM)	±30.
Survey Flow Angles (Deg)	±1.0
Temperatures (Deg K)	±0.6
Total Pressures (N/cm ²)	±0.17
Static Pressures (N/cm ²)	±0.10

TABLE VII OVERALL DESIGN PERFORMANCE PARAMETERS FOR ROTOR 35				
ROTOR TOTAL PRESSURE RATIO	1.865			
ROTOR TOTAL TEMPERATURE RATIO	1.225			
ROTOR ADIABATIC EFFICIENCY	0.885			
ROTOR POLYTROPIC EFFICIENCY	0.877			
ROTOR HEAD RISE COEFFICIENT	0.273			
FLOW COEFFICIENT	0.451			
WEIGHT FLOW PER UNIT FRONTAL AREA (Kg/sec)	100.808			
WEIGHT FLOW PER UNIT ANNULUS AREA (Kg/sec)	199.989			
WEIGHT FLOW (Kg/sec)	20.188			
RPM	17188.7			
TIP SPEED (m/sec)	454.456			
HUB/TIP RADIUS RATIO	0.70			
ROTOR ASPECT RATIO	1.19			
NUMBER OF ROTOR BLADES	36			

MERIDIONAL	AND RADIAL COORDINATE	BLE VIII ES OF THE ENDWALLS AND R L COORDINATES	OTOR BLADES
HUB AXIAL POSITION (CM)	HUB RADIAL POSITION (CM)	SHROUD AXIAL POSITION (CM)	SHROUD RADIAL POSITION (CM)
-22.86	17.528	-22.880	25.654
-15.40	17.526	-15.400	25.654
-7.62	17.526	-7.62	25.645
-2.588	17.539	-2.568	25.643
0.0	17.780	0.0	25.4
1.854	18.255	1.854	24.925
4.137	18.714	3.282	24.511
4.859	18.821	4.859	24.232
6.586	19.035	6.538	24.145
8.89	19.279	8.89	23.993
10.640	19.380	10.64	23.851
12.7	19.431	12.7	23.749
15.4	19.431	15.4	23.749
	b) Blade Leading and	Trailing Edge Coordinates	
LEADING	EDGE	TRAIL	JING EDGE
AXIAL POSITION (CM)	RADIAL POSITION (CM)	AXIAL POSITION (CM)	RADIAL POSITION (CI
0.	17.989	4.096	18.88
.039	18.849	4.017	19.4 6 6
.088	19.662	3.935	20.044
.140	20.428	3.852	20.616
.212	21.157	3.752	21,168
.286	21.856	3.675	21.719
.346	22.533	3.579	22.262
.410	23.197	3.533	22.796
.476	23.841	3.47	23.324
.544	24.480	3.404	23.849
.611	25.11	3.312	24.375

	LASER AN	TABLE IX IEMOMETER SURVEY LOC	CATIONS	
SPAN (PERCENT)	CHORD (PERCENT)	AXIAL POSITION (CM)	RADIAL POSITION (CM)	GONIOMETER OFFSETS (DEGREES)
5	UPSTREAM SURVEY	-2.54	17.858	О.
10	LOCATION	-2.54	18.273	0.
20		-2.54	19.098	0.
30		-2.54	20.752	0.
40		-2.54	21.58	0.
50	_] [-2.54	22.329	0.
60		-2.54	22.804	0.
70		-2.54	23.233	0.
80		-2.54	24.061	0.
90		-2.54	24.881	0.
5	-5%	175	18.151	0.
10	CHORD	157	18.542	0.
20		102	19.319	0.
30		064	20.063	0.
40		0.018	20.825	0.
50		0.104	21.577	0.
60		0.183	22.329	0.
70		0.262	23.066	0.
80		0.348	23.807	0.
90	7	0.439	24,541	0.

		TABLE IX continued		
5	5%	0.231	18.202	O .
10	CHORD	0.244	18.588	-5.,0.,5.
20		0.290	19.352	-5.,2.,5.
30		0.318	20.081	-5.,2.,5.
40		0.378	20.833	-5.,2.,5.
50		0,450	21.575	-5.,0. <u>,2.,</u> 4.,5.
60		0.513	22.316	-5.,2.,5.
70		0.562	23.066	-5.,2.,5.
80		0.648	25.238	-5.,2.,5.
90		0.724	26.501	0.2
5	25%	1.046	18.354	0.
10	CHORD	1.049	18.677	-5.,-1.,5.
20		1.069	19.434	-5.,-1.,5.
30		1.074	20.137	-5.,-1., 4.,5.
40		1.102	20.851	-5.,-1., 0.,4.,5.
50		1.140	21.562	-5.,0,.5.
60		1.168	22.273	-5.,-2.16, 2.16,5.
70]	1.204	22.974	-5.,4.,5.
80		1.247	23.683	-5.,4.,5.
90		1.290	24.381	2.

		TABLE IX continued		
5	50% CHORD	2.065	18.545	-3.,-2.5, 2.5,3.
10		2.057	18.877	-3.,-2.5, 2.5,3.
20		2.045	19.545	-3.,-2.5, -2.5,3.
30		2.022	20.208	-3.,-2.5, -1.,1.,3.
40		2.007	20.876	-3.,-2.5, 2.5,3.
50		2.002	21.547	-3.,-1., 1.,3.
6 0		1.969	22.217	-3.,3 .
70		1.989	22.885	-3.,3.
80		1.994	23.566	-3.,3 .
90		2.002	24.229	0.
5	75%	3.086	18.733	-3.5,5
10	CHORD	3.063	19.040	-5.,-3.5, 0.5
20		3.018	19.655	-5.5,0.
30		2.967	20.272	-6.,-0.6, 0.
40		2.911	20.899	-6.,0., 1.,6.
50		2.865	21.529	-5.5,0.
60		2.860	22.164	-2.5, -2.469
70	,	2.774	22.799	-3.03
. 80		2.741	23.449	4.
90		2.710	24.077	-2.,-4.

		TABLE IX continued		
5	95%	3.901	18.865	-5.,-3.
10	CHORD	3.868	19.169	-2
20		3.797	19.741	-2
30	_	3.724	20.328	-2
40		3.635	20.917	-2
50		3.553	21.518	-2.
60		3.465	22.121	-2.
70		3.401	22.728	-4.
80		3.340	23.365	-4 .
90		3.277	23.957	-4., 0.
5	105%	4.308	18.933	0.
10	CHORD	4.270	19.213	0.
20		4.188	19.769	0.
30		4.105	20.345	0.
40		3.995	20.925	0.
50		3.899	21.516	0.
60	_	3.795	22.111	0.
70		3.716	22.708	0.
80		3.640	23.327	O .
90		3.561	23.922	0.

TABLE X. - SCHEMATIC OF MARGINAL OR UNACCEPTABLE DATA POINTS.

(1) Axial velocity component.

Radial				8	tation			
position	1	2	3	4	5	6	7	
Chord	1	-5%	5%	25%	50%	15%	95%	105%
1 (5%)			·	•	+	?+	•	•
2 (10%)				+			+	•
3 (20%)								•
4 (30%)								•
5 (40%)								+
6 (50%)								
7 (60%)								
8 (70%)								
9 (80%)			х					+
10 (90%)	X?	•	•	•	•	7.	7.	7*
11 (95%)	•	•	•	•	•	•	•	•

Unacceptable data, both configurations.

X Unacceptable data, base line.

⁺ Unacceptable data, enhanced.
? Indicates questionable data.

(2) Tangential velocity component

Redial	Station										
position	1	2	3	4	5	•	7				
Chord	ı	-5%	5%	25%	50%	15%	95%	105%			
1 (5%)			х		?+	+	? •	•			
2 (10%)							+	+			
3 (20%)								+			
4 (30%)								+			
5 (40%)								+			
6 (50%)											
7 (60%)											
8 (70%)											
9 (80%)								+			
10 (90%)		X	+	7+	7 •	7*	3.0	7*			
11 (95%)	•	•	•	•	•	•	•	•			

^{*} Unacceptable data, both configurations.

X Unacceptable data, base line. + Unacceptable data, enhanced.

[?] Indicates questionable data.

Redial	Station										
position	1	2	3	4	5	6	7	8			
Chord	-	5%	5%	25%	50%	15%	95%	105%			
1 (5%)			•	•	7+	+	٠	•			
2 (10%)				+	?+		+	•			
3 (20%)								•			
4 (30%)		<u> </u>						•			
5 (40%)								+			
6 (50%)		ļ <u>.</u>									
7 (60%)											
* (70%)											
9 (80%)			•					+			
10 (90%)	х	70	•	•	•	7.0	•	•			
11 (95%)	•	•	•	•	•	•	•	•			

^{*} Unacceptable data, both configurations.

X Unacceptable data, base line.

⁺ Unacceptable data, enhanced

[?] Indicates questionable data.

TABLE XI. - DATA RUN COMBINATION SCHEDULE

TABLE XI DATA RUN COMBINATION SCHEDULE												
Final run	Combined	Reduced		,	Original	1						
1	1	1 2	1 3	2								
2	2	3 4	4 5	365 366								
3	3	5 6	6 7									
4	4	7 8	8	451 452								
5	5	9 10	10 11									
6	6	11 12	12 13	403 404								
7	7	13 14	14 15									
8	8	15 16	16 17									
9	9	17 18	18 19	454 453								
10	10	19 20	20 21									
11	11	21 22	37 38	75 76								
	12	23 24	39 78	41	77							
12	13	25 26	368 367									
	14	27 28	369 370									
13	15	29 30	79 80									
	16	31 32	45 44									
14	17	33 34	81 82									
	18	35 36	68 67	456	457							
15	19	37 38	69 71	70								
	20	39 40	72 73									
16	21	41 42	406 405	467 469	468 470							

				,		
17	22	43 44	89 90			
18	23	45 46	91 92			
19	24	47 48	93 95	94 96		
20	25	49 50	97 98			
	26	51 52	102 103	107 104		
	27	53 54	373 372	374		
	28	55 56	101 371			
21	29	57 58	106 105			
	30	59 60	108 111			
22	31	61 62	109 110			
	32	63 64	112 115			
23	33	65 66	113 114			
	34	67 68	116 119			
24	35	69 70	117 118			
	36	71 72	120 123	126 127		
25	37	73 74	407 408	575 578	577 579	
	38	75 76	410 409			
	39	77 78	121 122	125 128		
	40	79 80	131 132			
26	41	81 82	129 130			
	42	83 84	133 134			
27	43	85 86	135 136			
	44	87 88	139 141	142		

28 45 89 90	137 138				
29 46 91 92	143 144	145			
30 47 93 94	146 149	147 150	148 151	<u> </u>	
48 95 96	155 157	156 158	159		
31 49 97 98	152 153	154			
50 99 100	160 161				
32 51 101 102	162 163				
52 103 104	166 167				
33 53 105 106	164 165				
54 107 108	174 175	178 176			
55 109 110	62 63				
34 56 111 112	172 170	173 171			
57 113 114	179 180	182 181			
58 115 116	423 424				
35 59 117 118	183 184	422 421			
60 119 120	187 188				
36 61 121 122	186 185				
62 123 124	190 189	191			
37 63 125 126	192 193				
64 127 128	200 1 99				
38 65 129 130	197 198	196 195	194		
39 66 131 132	201 202	204 203			

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			·			
	67	133 134	212 210	213 211		
40	68	135 136	205 208	206 209	207	
	69	137 138	214 215			
	70	139 140	378 377			
41	71	141 142	375 376			
	72	143 144	216 217			
	73	145 146	218 219			
	74	147 148	220 221			
42	75	149 150	222 223			
	76	151 152	226 227			
43	77	153 154	224 225			
	78	155 156	22 8 230	229		
44	79	157 158	232 231			
	80	159 160	234 235			
	81	161 162	427 428			
45	82	163 164	426 425			
	8 3	165 166	237 236			
46	84	167 168	238 239			
	85	169 170	242 243			
47	86	171 172	241 240			
	87	173 174	245 244	246		
48	88	175 176	247 248			
49	89	177 178	473 472	474		

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				···			·
50	90	179 180	251 252				
	91	181 182	250 249				
	92	183 184	385 384				
51	93	185 186	255 253	254			
	94	187 188	256 257	382 383			
52	95	189	259 258	260			
	96	190	262	200			
53	97	192	263 265				
	98	194 195	269				
54	99	196	268				
55	100	198	267 270				
	101	200	271 273				
		202	271				
56	102	203 204	274 275				
57	103	205 206	277 276				
58	104	207 208	278 279	280	281		
59	105	209 210	282 284	283 287	286		
60	106	211 212	290 288	291 289	312 310	313 311	
61	107	213 214	307 308	309 387	389 388		
	108	215 216	297 299	298 300	301	302	
62	109	217 218	314 315				
	110	219 220	306 305				
63	111	221 222	320 316	317			
64	112	223 224	319 318				
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			,	,			
65	113	225 226	321 322				
66	114	227 228	324 323				
67	115	229 230	325 326				
68	116	231 232	332 329	330			
69	117	233 234	333 334				
70	118	235 236	337 335	338 336			
72	119	237 238	346 345	347			
71	120	239 240	390 392	391 393			
	121	241 242	342 343				
73	122	243 244	348 350	349			
74	123	245 246	352 351				
75	124	247 248	353 354				
76	125	249 250	356 355				
77	126	251 252	357 358				
78	127	253 254	360 359				
79	128	255 256	361 362	363 364			
80	129	257 258	480 475	481 476	477	478	479
81	130	259 260	482 484	483 485			
82	131	261 262	488 486	487			
83	132	263 264	489 491	490 492			
84	133	265 266	494 493				
85	134	267 268	495 496				
86	135	269 270	499 497	500 498			

			,	·			
87	136	271 272	501 503	502 504			
88	137	273 274	507 505	508 506			
89	138	275 276	5 09 510				
90	139	277 278	513 511	566 512	565		
910	140	279 280	515 516	523 517	524 518	526	527
91b	141	281 282	563 564				
92a	142	283 284	533 528	\$34 \$29	530		
92b	143	285 286	562 560				
	144	287 188	539 543	540 544	541		
93	145	289 290	558 559				
94	146	291 292	554 555				
96	147	293 294	547 549	548 550			
97	148	295 296	553 551				
98	149	297 298	567 568				
99	150	299 300	570 569				
100	151	301 302	571 572				
101	152	303 304	586 580	587 581	582		
102	153	305 306	593 596	595 597	598		
103	154	307 308	602 599	603 600	601		
104	155	309 310	604 608	607 609			
105	156	311 312	612 611	613	614		
106	157	313 314	615 616				ı
107	158	315 316	620 625	621	622		

		T	т —		r	Γ	
108	159	317 318	623 624				
109	160	319 320	628 626	629 627	630		
	161	321 322	632 635	636			
110	162	323 324	643 638	644 639	641	642	
	163	325 326	645 646				-
111	164	327 328	669	678 651			
112	165	329	653	654 655	657 656		
	166	330	652	660	661	664	
113	167	332	662	663			
	168	334	671				
114	169	336 337	669	670			
		338	668		 	-	
	170	339 340	673 674				
	171	341 342	675 677	676			
115	172	343 344	683 682				
	173	345 346	684 685				
116	174	347 348	687 686				
117	175	349 350	688				
118	176	351 352	691				
119	177	353 354	692 693				
	178	355 356	695 694				
120	179	357 358	712 711				
	180	359 360	705 708	706 709	707 710		
121	181	361 362	713 719	714 720	715	717	718

:	182	363 364	725 726				
122	183	365 366	731 732	733			
	184	367 368	729 727	730 728			
123	185	369 370	736 734	737 735	738		
	186	371 372	699				
	187	373 374	696 697				
	188	375 376	743 742				
	189	377 378	739 741	740			
124	190	379 380	749 748				
	191	381 382	746 747				
125	192	383 384	750 751				
	193	385 386	700 702	701			
	194	387 388	704 703				
	195	389 390	753 752				
126	196	391 392	757 756	ļ 			
	197	393 394	754 755				
127	198	395 396	758 759				
128	199	397 398	761 760				,
	200	399 400	766 765				
129	201	401 402	762 763				
130	202	403 404	768 770	769 773	771		
	203	405 406	782 784	783 785	786	_	
	204	407 408	780 777	781 779			

	,			·			
	205	409 410	789 787	788			
131	206	411 412	790 792	791 793			
	207	413 414	795 794				
132	208	415	798	800			
	209	416	797	800			
	ļ	418	796	ļ			
133	210	419 420	805 806				
134	211	421 422	803 801	804 802			
	212	423 424	809 810	811			
	213	425 426	808 807				
135	214	427 428	814 815				
					<u> </u>		
136	215	429 430	813 812				
137	216	431 432	817 816	818 819			
138	217	433 434	822 820	823 821			
139	218	435 436	824 825				
140	219	437 438	830 826	831 827	828	829	
141	220	439 440	834 840	835 841	836	837	
142	221	441 442	844 842	845 843	846	847	848
143	222	443 444	851 854	852 855	856		
144	223	445 446	859 857	860 858			
145	224	447 448	861 862				
146	225	449 450	864 863				
147	226	451 452	865 866				
148	227	453 454	868 867				

			γ	,			
	228	455 456	931 932				
149	229	457 458	935 933	934			
150	230	459 460	874 869	875 870	942 871	872	873
	231	461 462	877 881	943 882	951 944	950	
151	232	463 464	876 879	878 880	947 945	952 946	948
152	233	465 466	885 883	953 884	954 887	955	956
153a	234	467 468	890 888	891 889	958 957	960	
153b	235	469 470	895 897	896			
154	236	471 472	918 898	919 899	920 922	921	
	237	473 474	901 902				
155	238	475 476	915 916	917			
156	239	477 478	904 903	914 913			
157	240	479 480	905 906	911 912			
158	241	281 482	910 909				
	242	483 484	928 924	929			
95	243	485 486	927 923	925	926		

TABLE XII ESTIMATED INACCURACIES IN LASER ANEMOMETRY INSTRUMENTATION					
Probe Volume Position (mm)	±0.05				
Beam Director Setting Angle (Deg)	±0.01				
Statistical blasing Error (maximum)	1.0%				
Window Width Velocity Gradient Error	2.0%				
Particle Velocity Lag (Leading edge region)	10%				
Particle Angle Lag (Deg)(blade trailing edge)	6%				

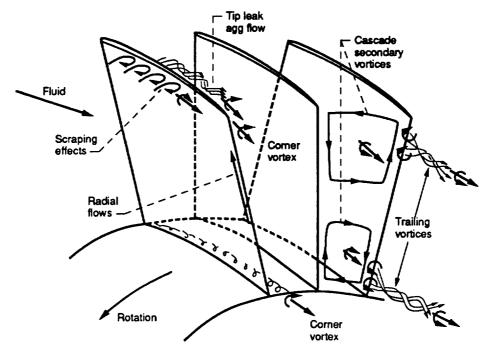


Figure 1.—Secondary flow and vortices in an axial flow compressor rotor.

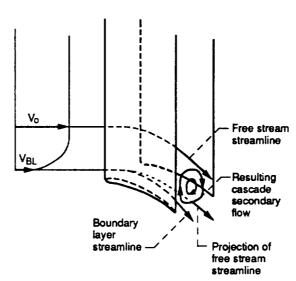


Figure 2.—Generation of secondary flows in a channel bend due to a nonuniform inlet velocity.

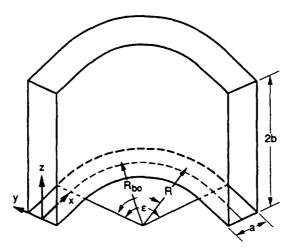


Figure 3.—Channel used for Squire and Winter analysis.

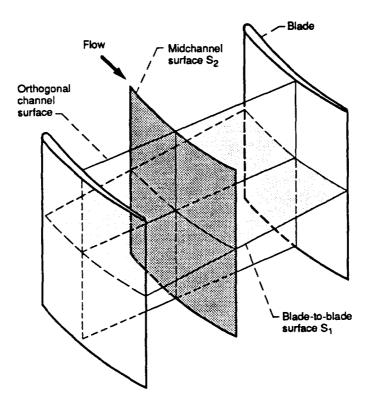


Figure 4.—Two-dimensional analysis surfaces in a turbomachine.

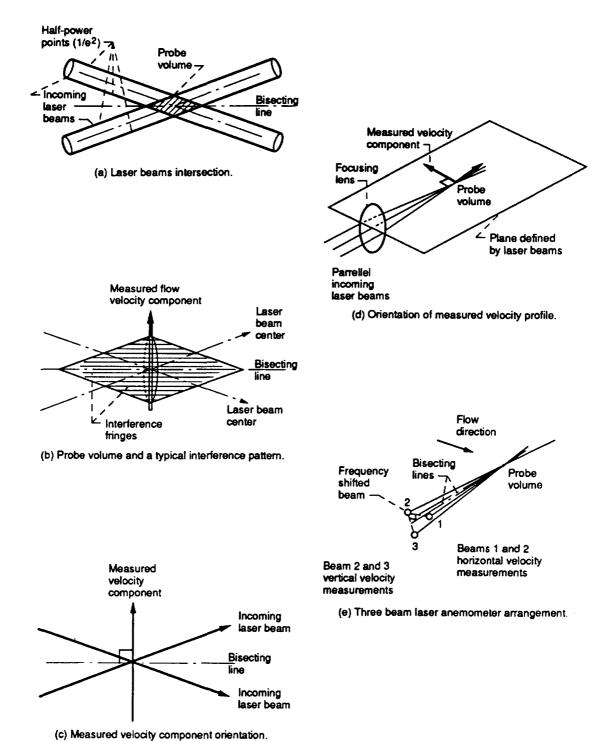


Figure 5.—Typical laser anemometer probe volume.

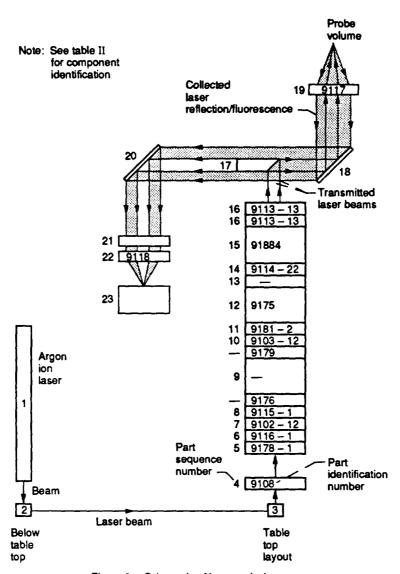
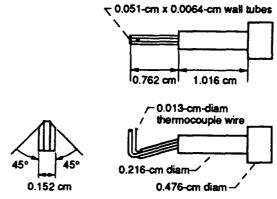
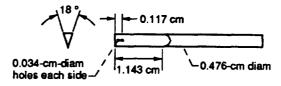


Figure 6.—Schematic of laser optical system.



(a) Cobra probe for total pressure, total temperature, and andle measurements.



(b) Wedge probe for static pressure measurements.

Figure 7.—Aerodynamic survey probes.

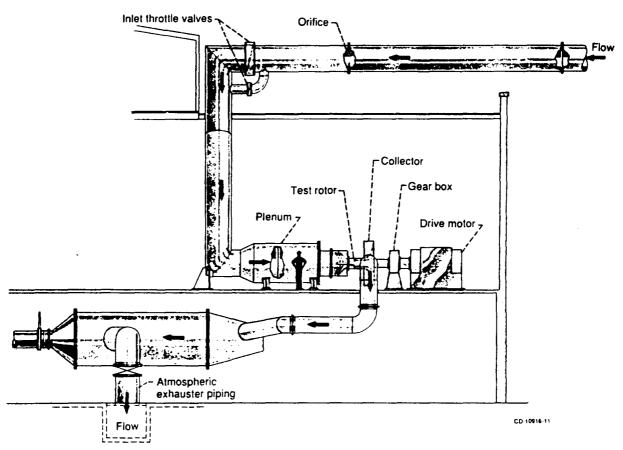


Figure 8.—Test facility.

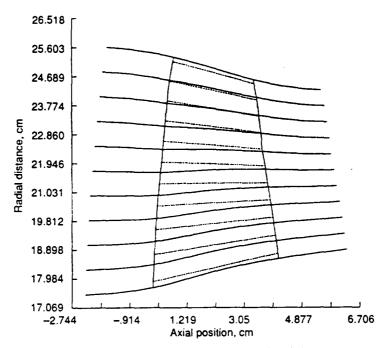


Figure 9.—Compressor rotor meridional view.

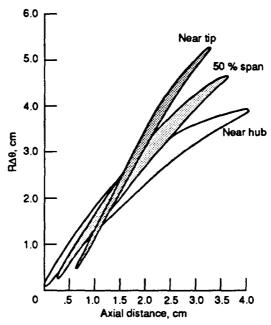


Figure 10.—Compressor rotor blade sections at three radial locations.

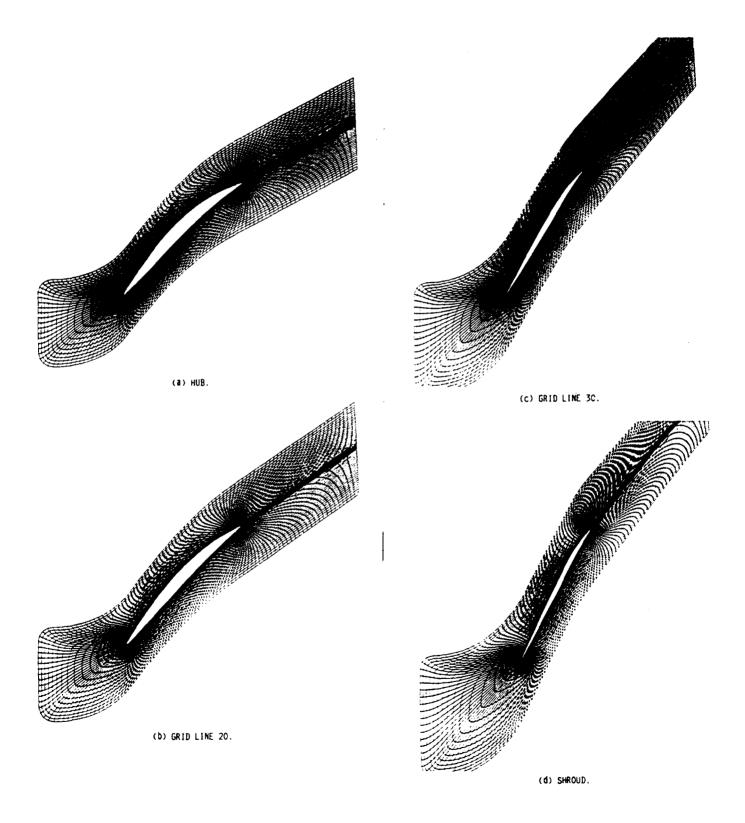
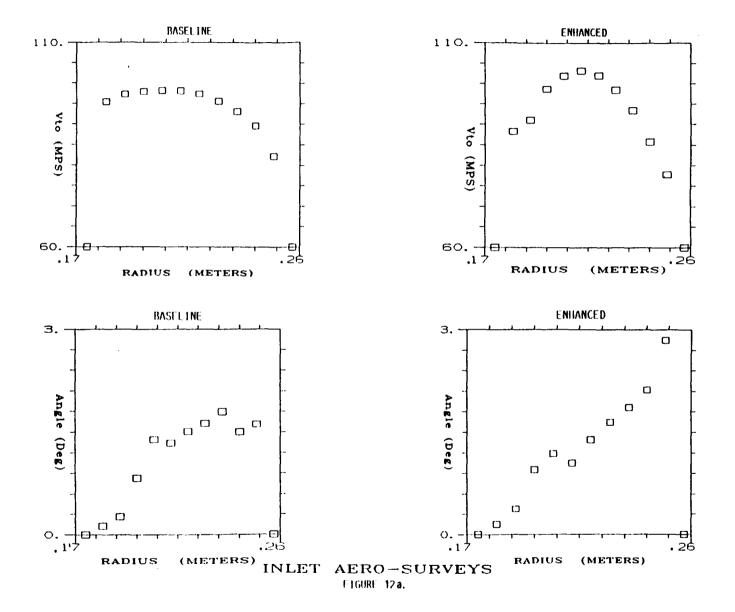
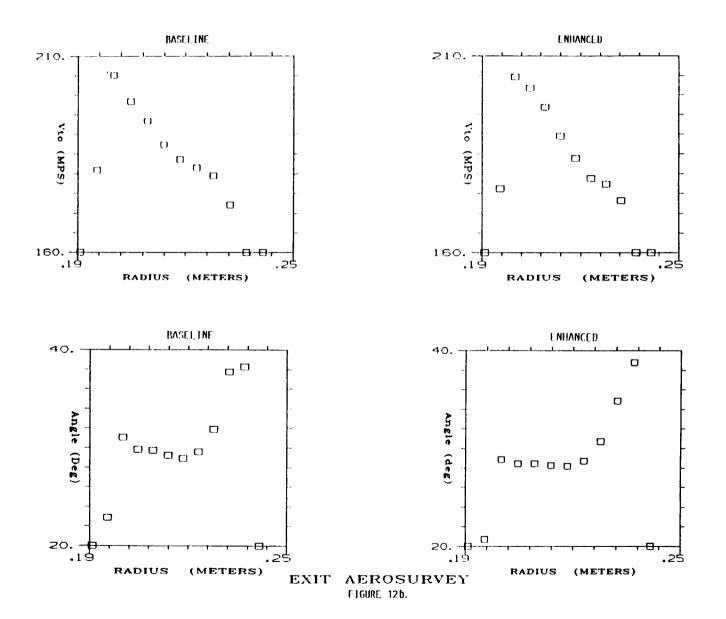


FIGURE 11. - TYPICAL BLADE-TO-BLADE COMPUTATIONAL PLANE.





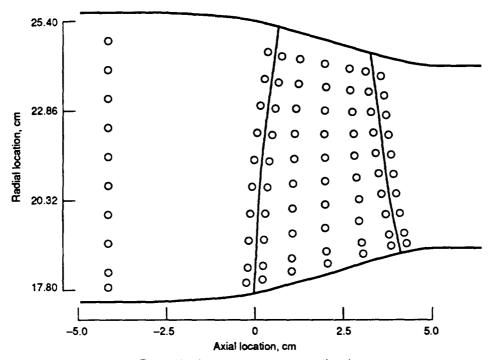


Figure 13.—Laser anemometer survey locations.

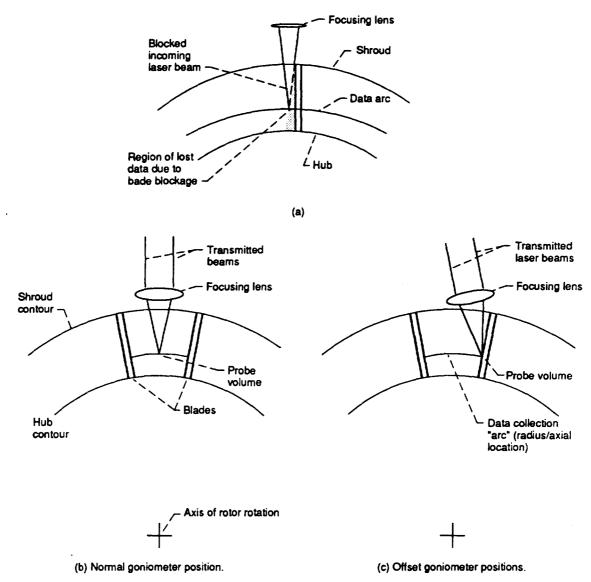
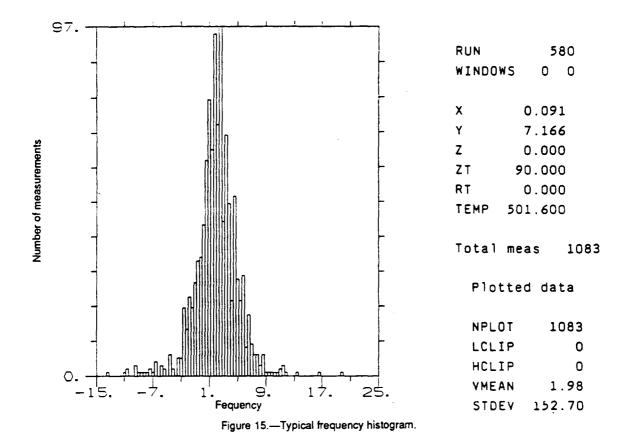


Figure 14.—Goniometer movement to minimize blade blockage.



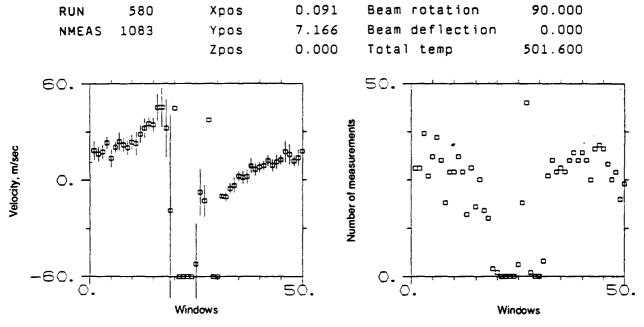


Figure 16.—Typical blade-to-blade velocity plot.

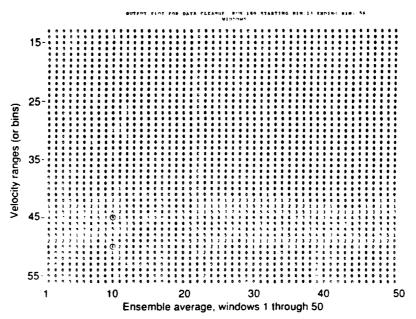


Figure 17.—Nondimensionalized frequency histogram for a typical data run.

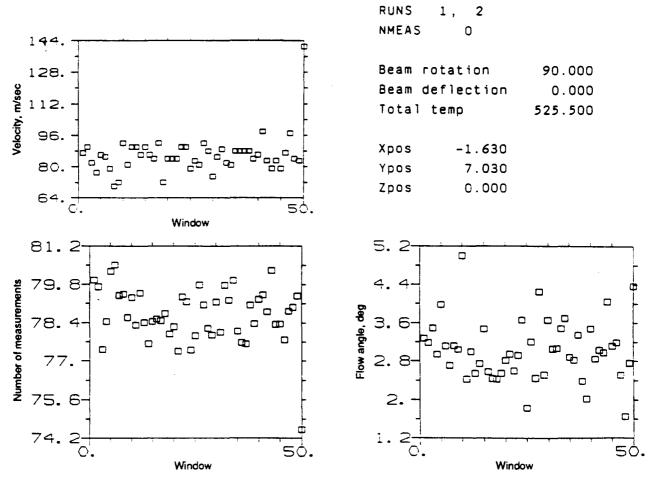


Figure 18.—Typical combined component velocity output, showing axial/tangential velocity, flow angle and measurements per bin.

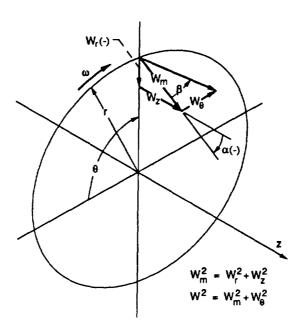
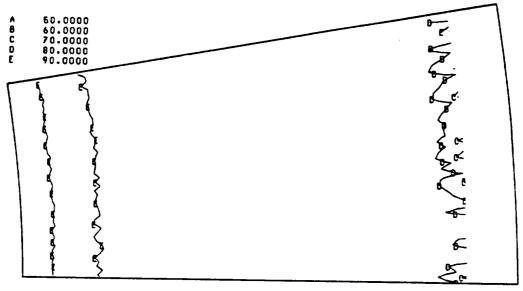
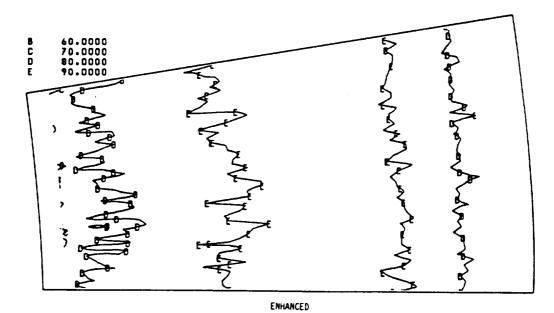


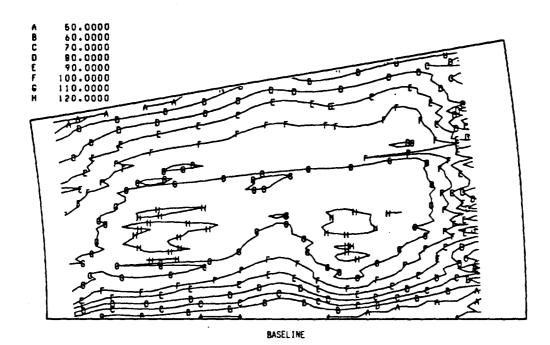
Figure 19.—Cylindrical coordinate system and velocity components.

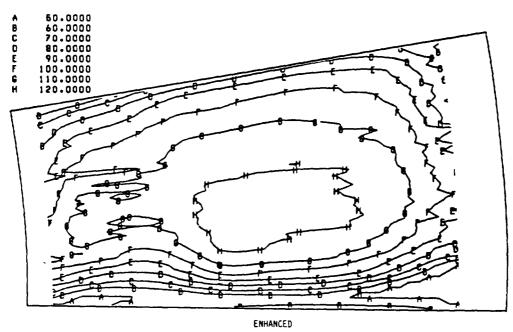




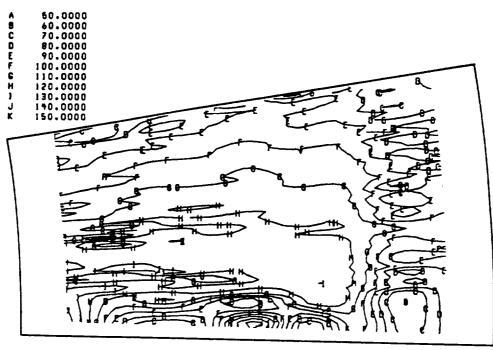


(a) STATION 1.
FIGURE 20. - AXIAL VELOCITY, CROSS CHANNEL.

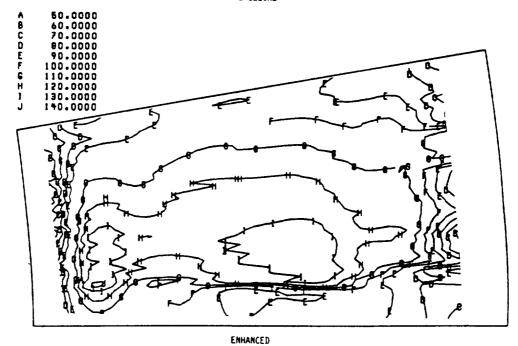




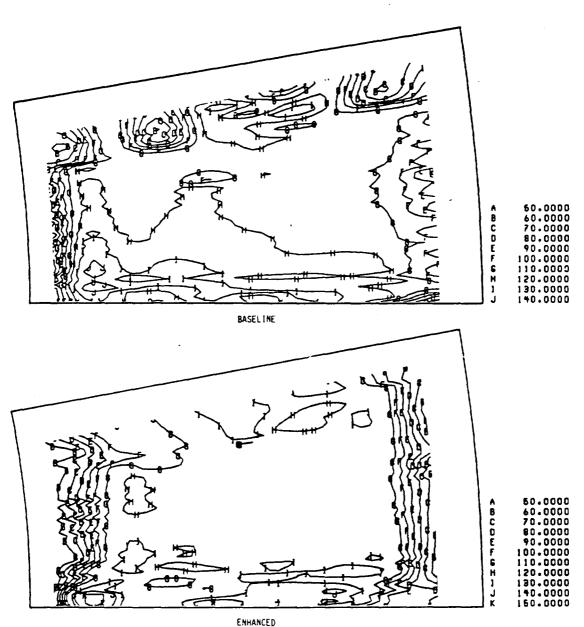
(b) STATION 2.
FIGURE 20. - CONTINUED.



BASELINE



(c) STATION 3. FIGURE 20. - CONTINUED.



(d) STATION 4.
FIGURE 20. - CONTINUED.

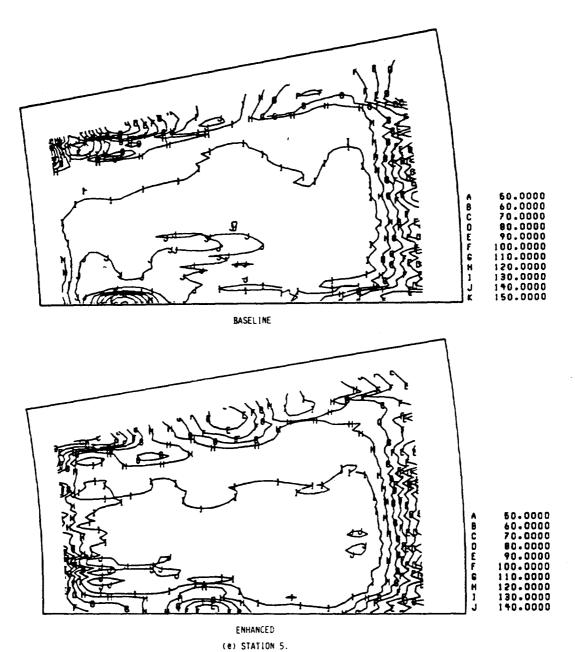


FIGURE 20. - CONTINUED.

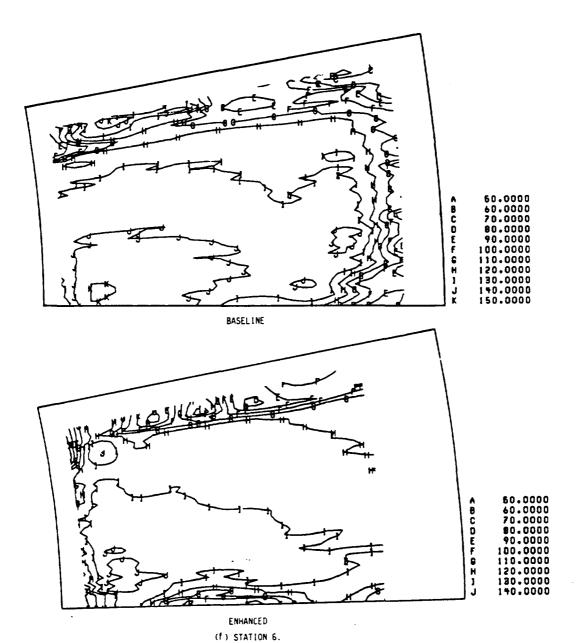
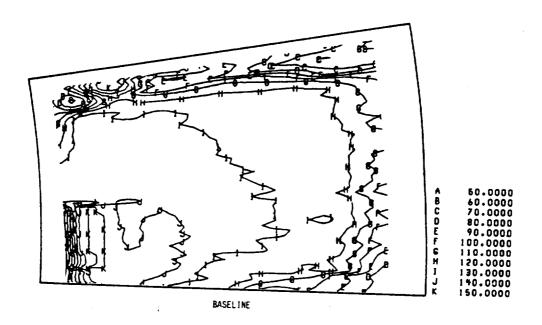
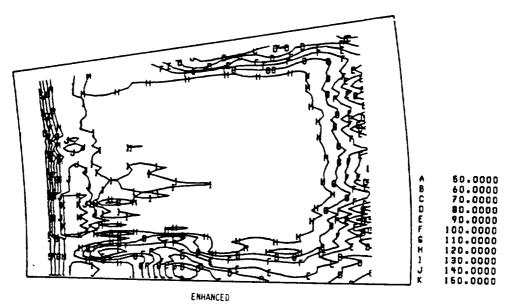
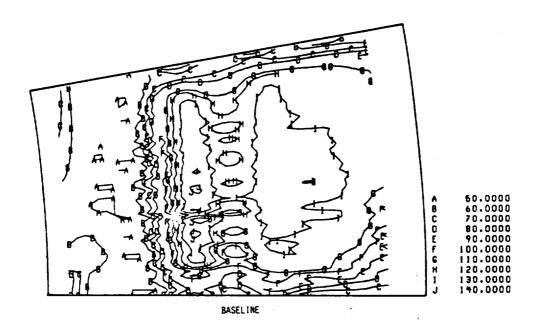


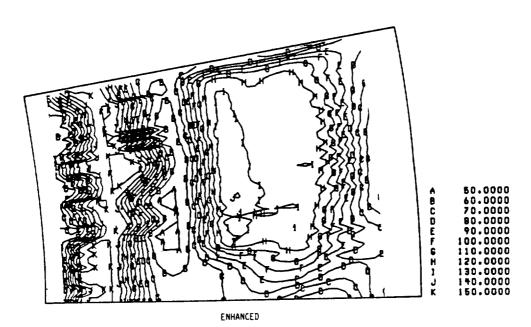
FIGURE 20. - CONTINUED.





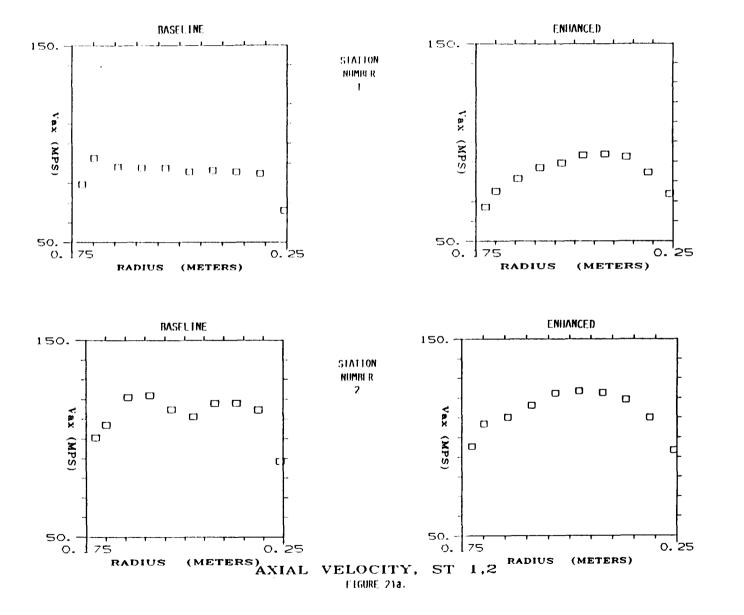
(g) STATION 7. FIGURE 20. - CONTINUED.

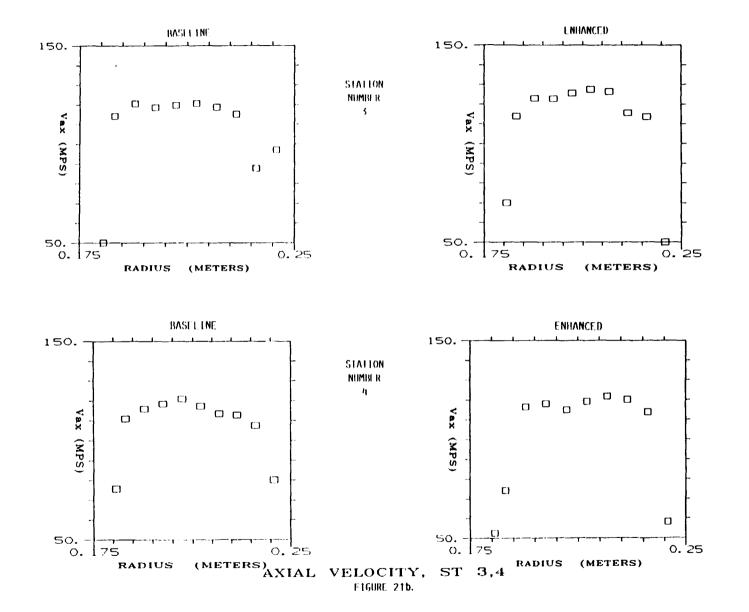




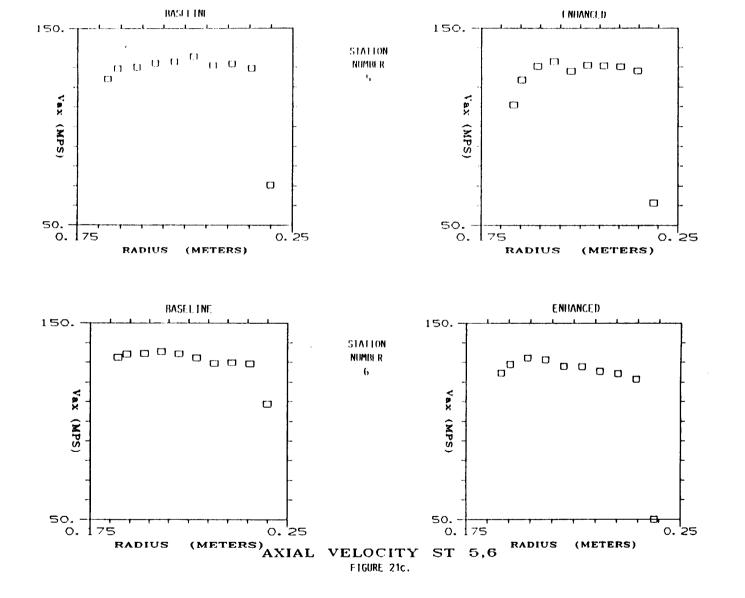
(h) STATION 8. FIGURE 20. - CONCLUDED.

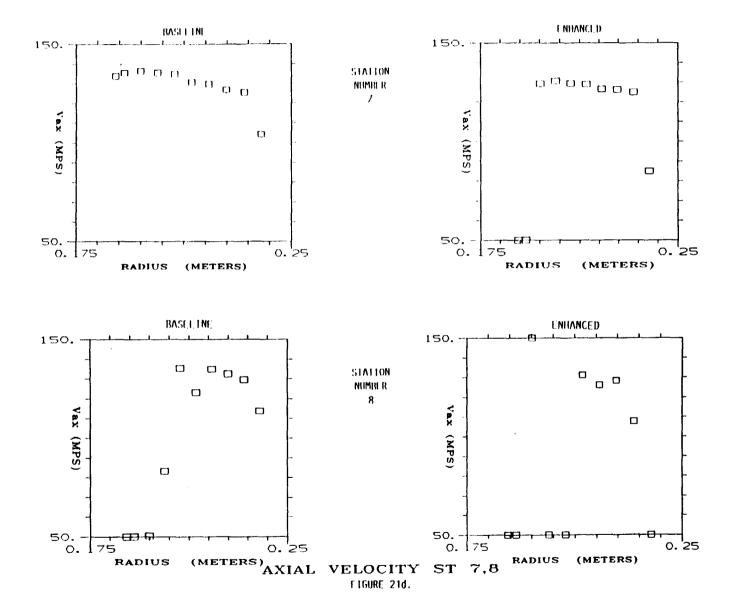


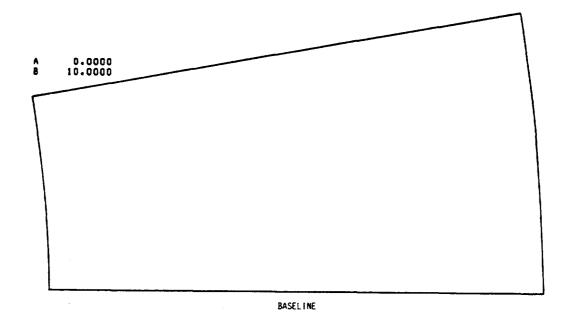


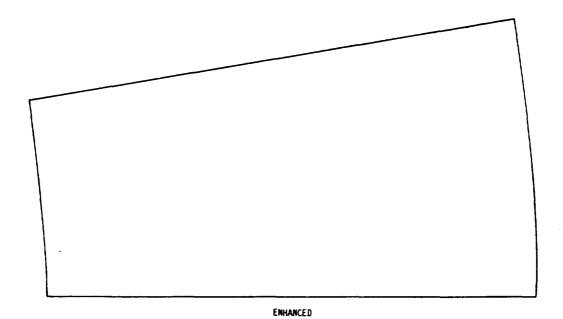




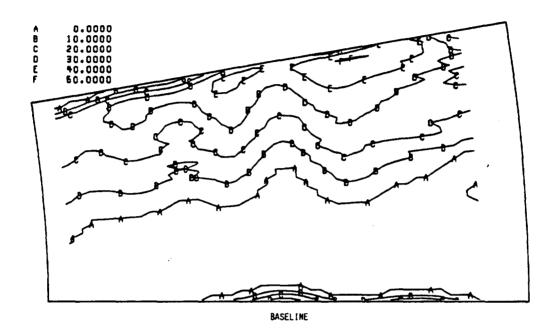


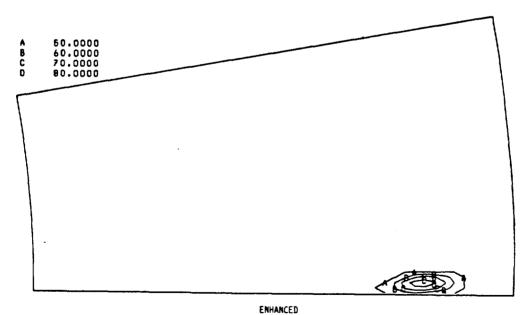




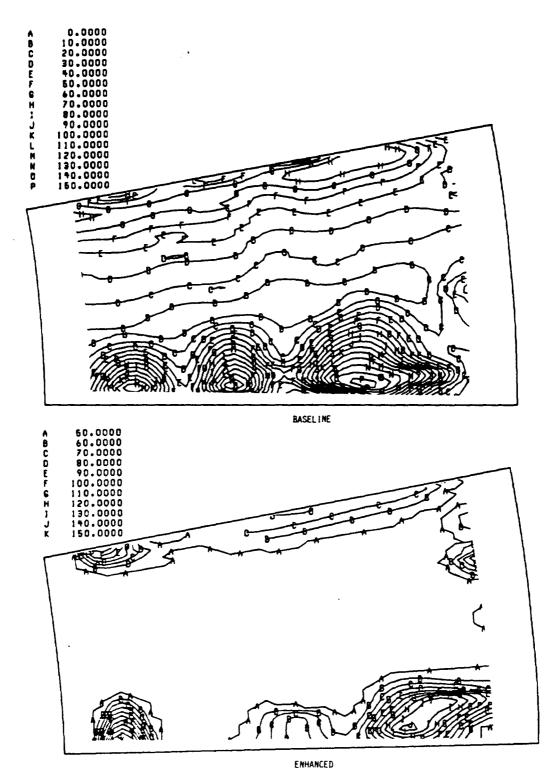


(a) STATION 1.
FIGURE 22. - ABSOLUTE TANGENTIAL VELOCITY.

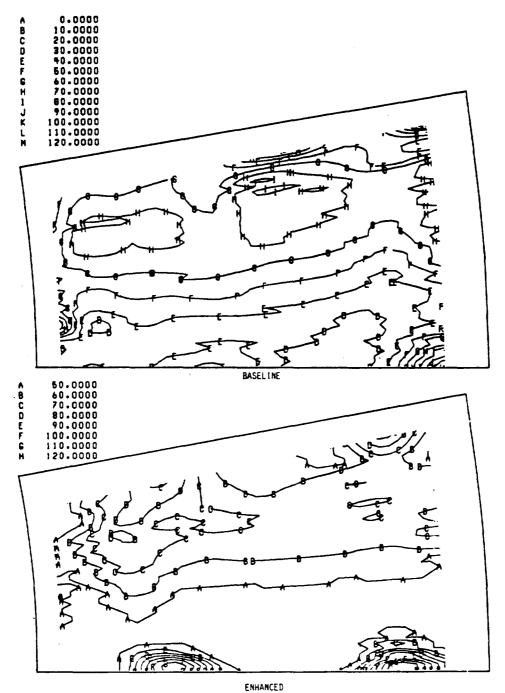




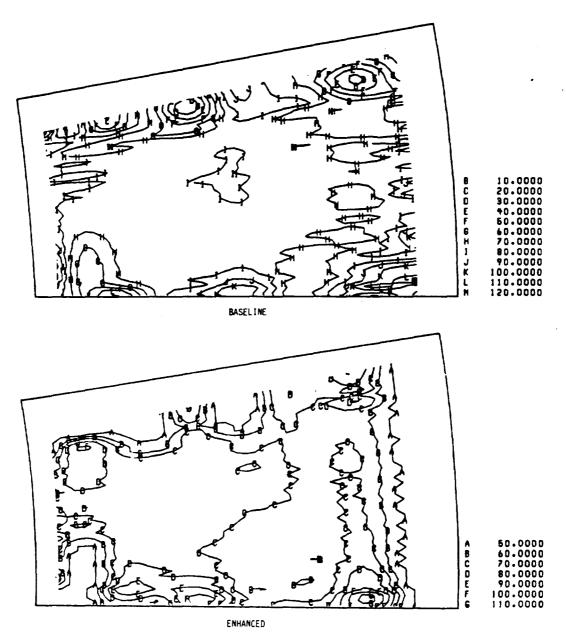
(b) STATION 2.
FIGURE 22. - CONTINUED.



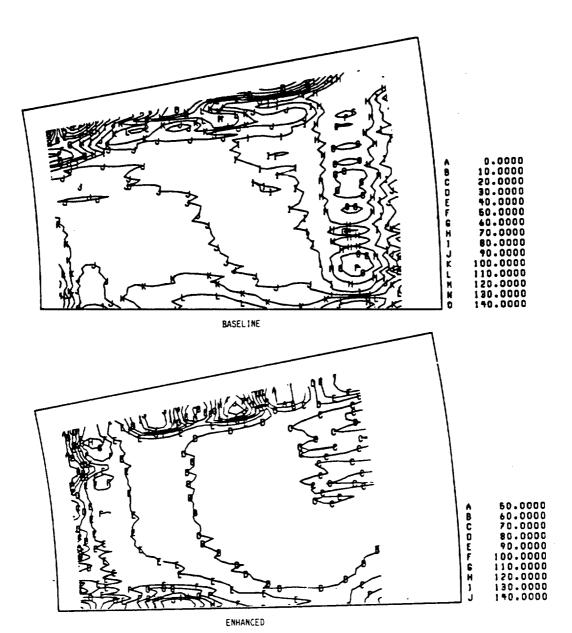
(c) STATION 3.
FIGURE 22. - CONTINUED.



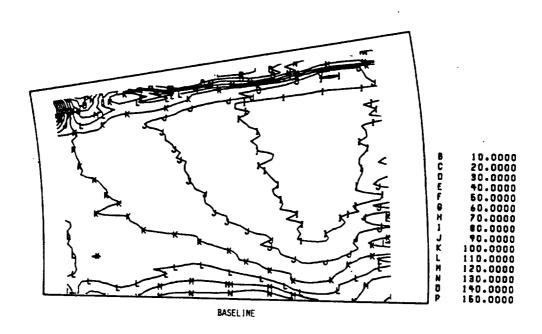
(d) STATION 4.
FIGURE 22. - CONTINUED.

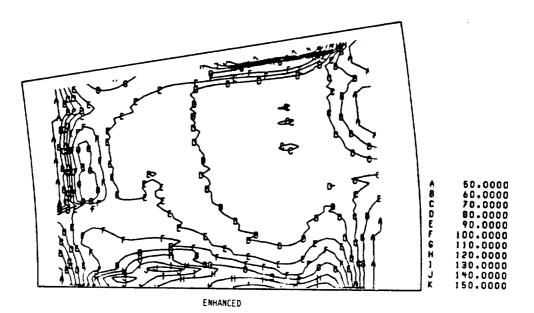


(e) STATION 5.
FIGURE 22. - CONTINUED.

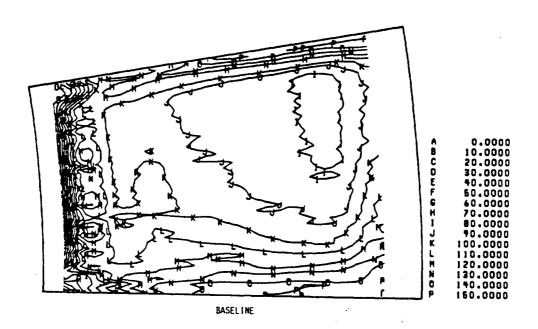


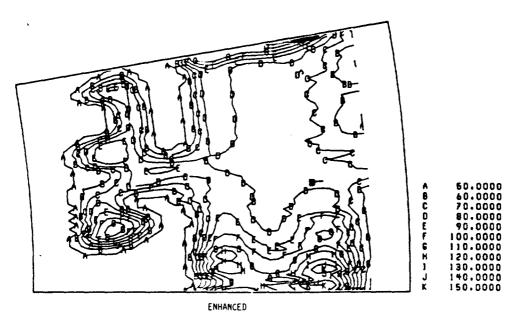
(f) STATION 6.
FIGURE 22. - CONTINUED.



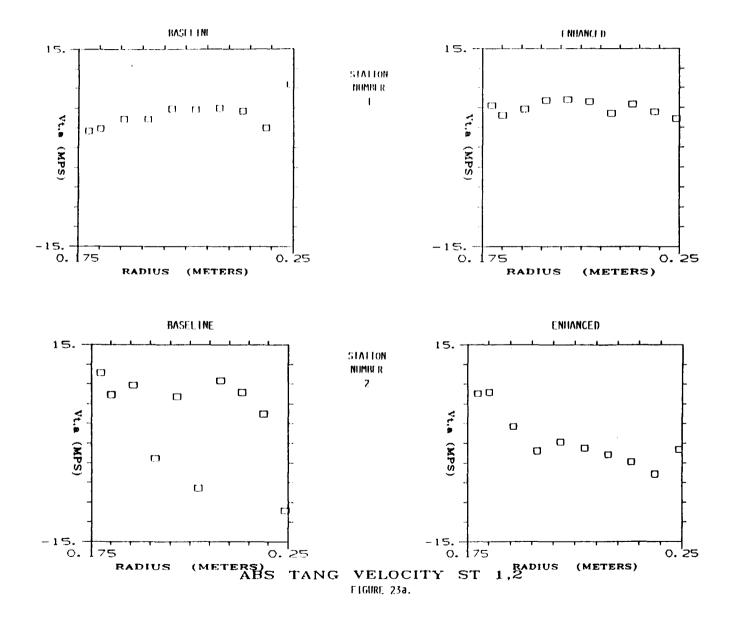


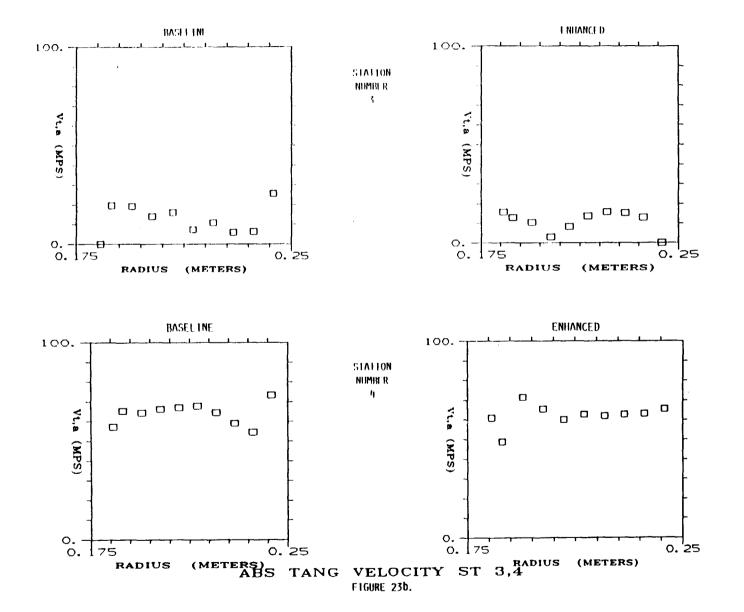
(g) STATION 7.
FIGURE 22. - CONTINUED.

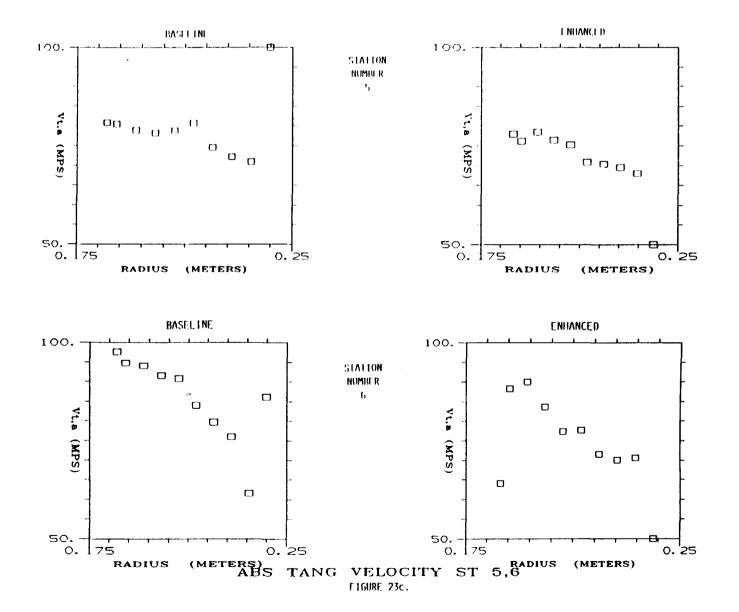


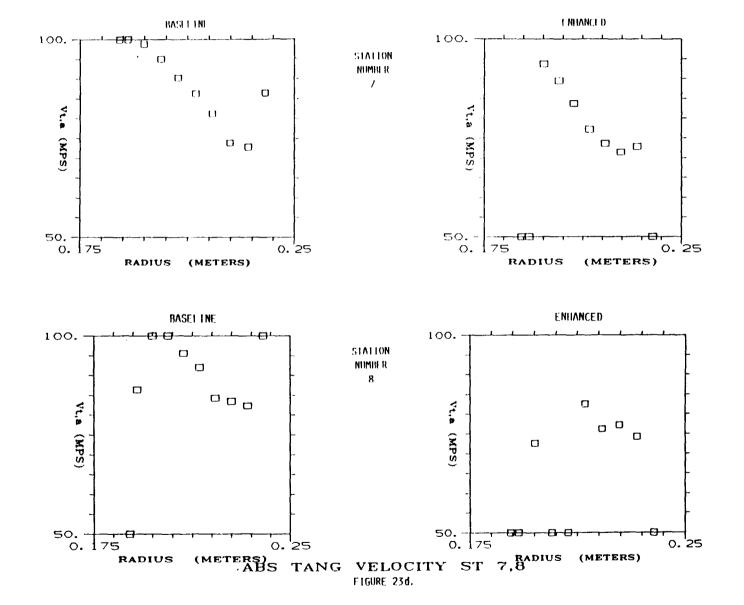


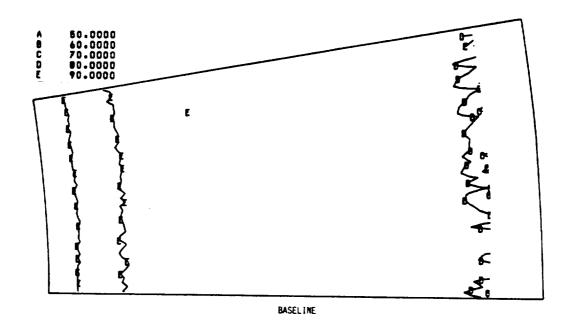
(h) STATION 8. FIGURE 22. - CONCLUDED.











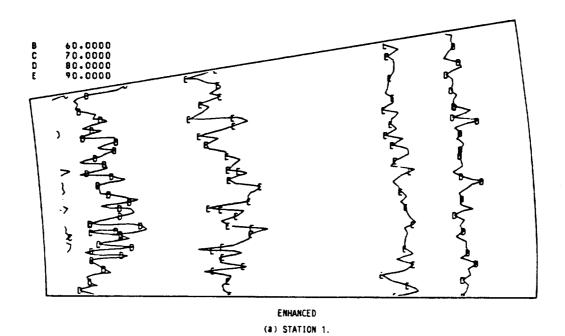
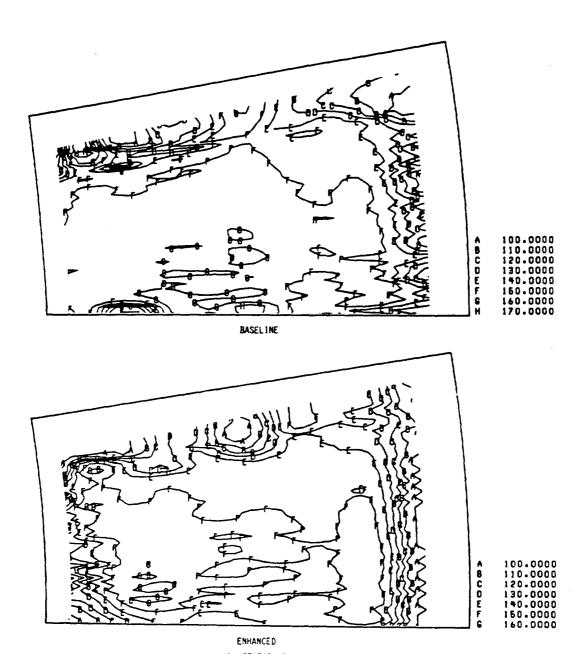
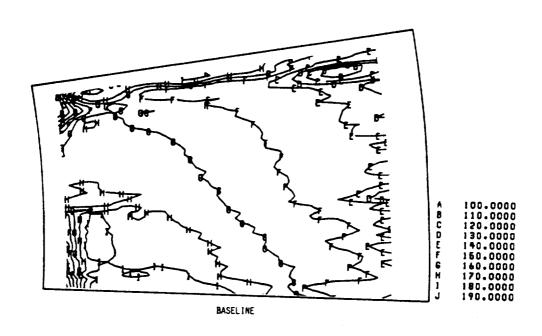


FIGURE 24. - ABSOLUTE TOTAL VELOCITY, CROSS CHANNEL.



(b) STATION 5.
FIGURE 24. - CONTINUED.



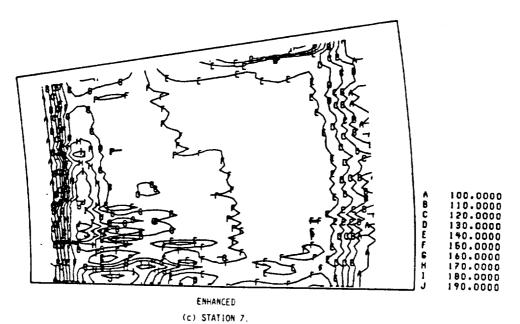
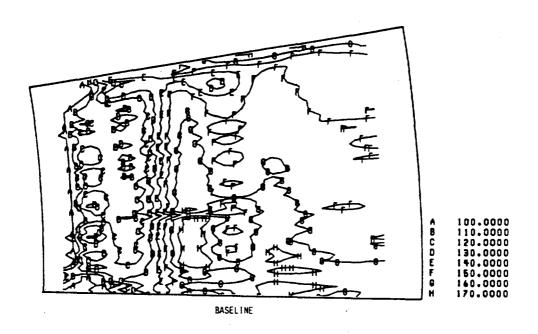
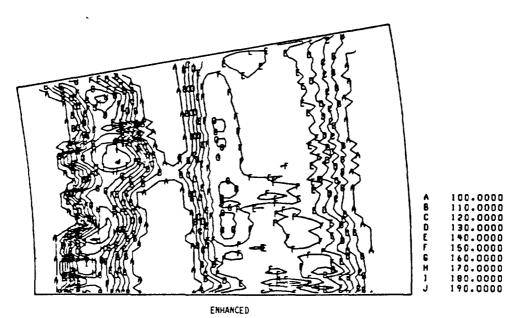
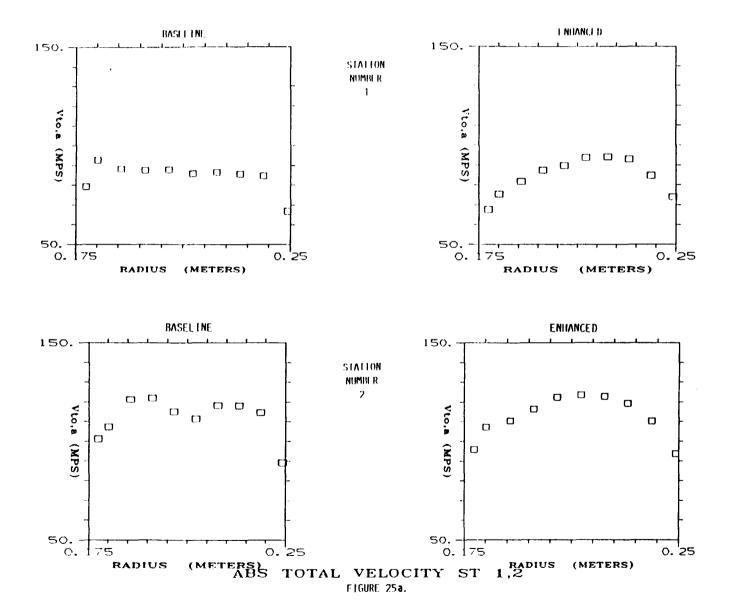


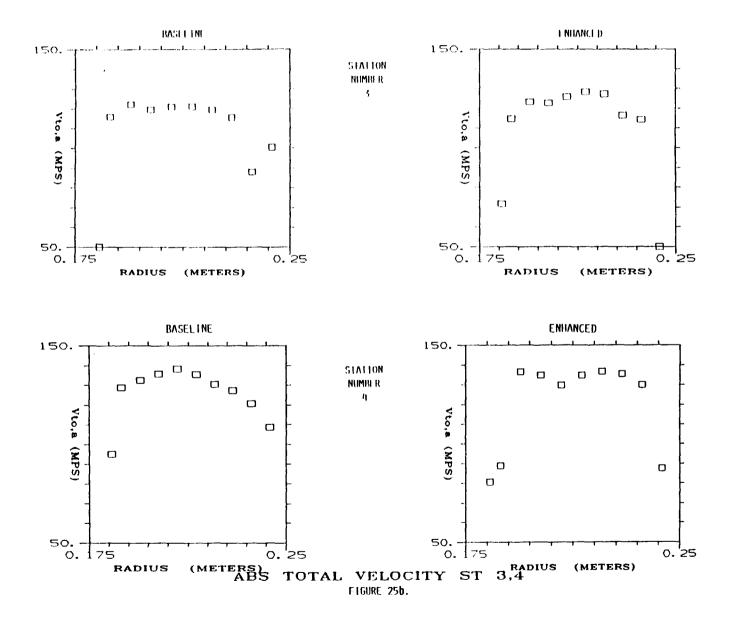
FIGURE 24. - CONTINUED.

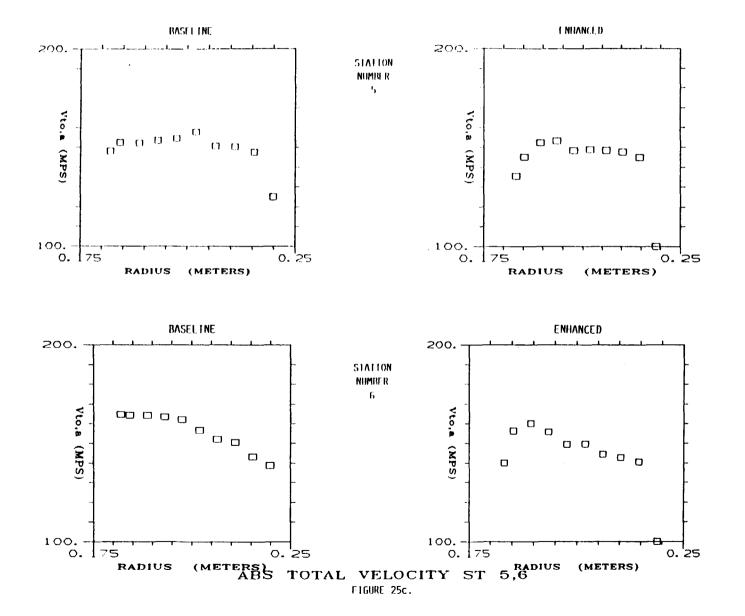


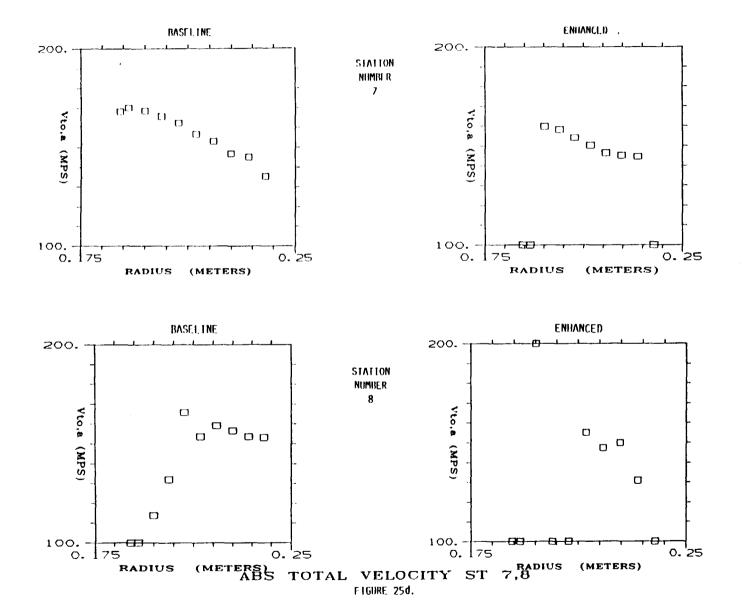


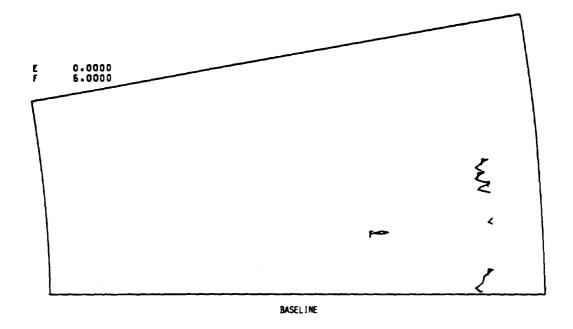
(d) STATION 8.
FIGURE 24. - CONCLUDED.











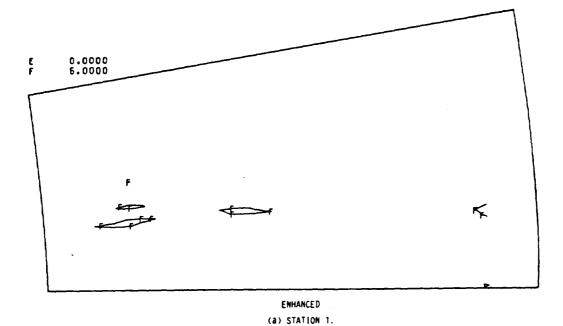
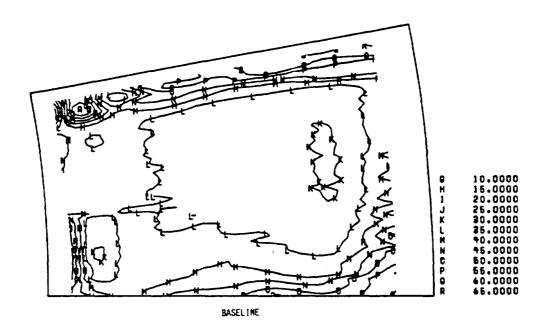
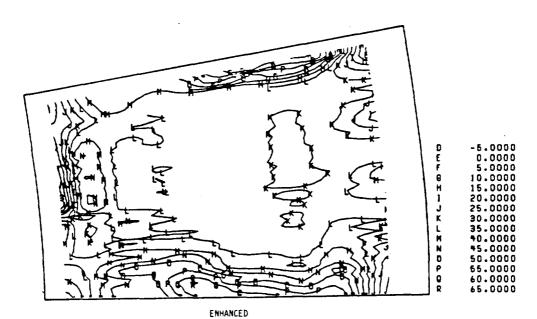
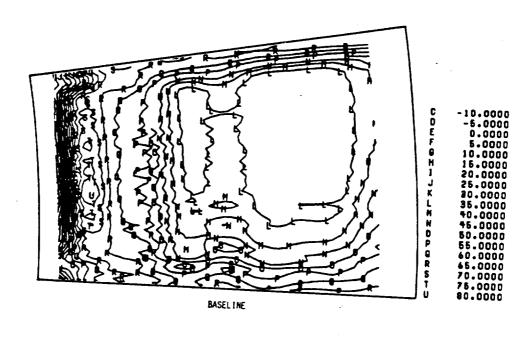


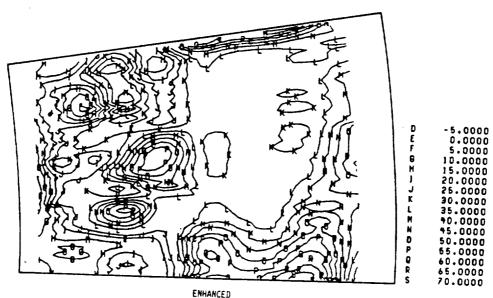
FIGURE 26. - ABSOLUTE FLOW ANGLE, CROSS CHANNEL.



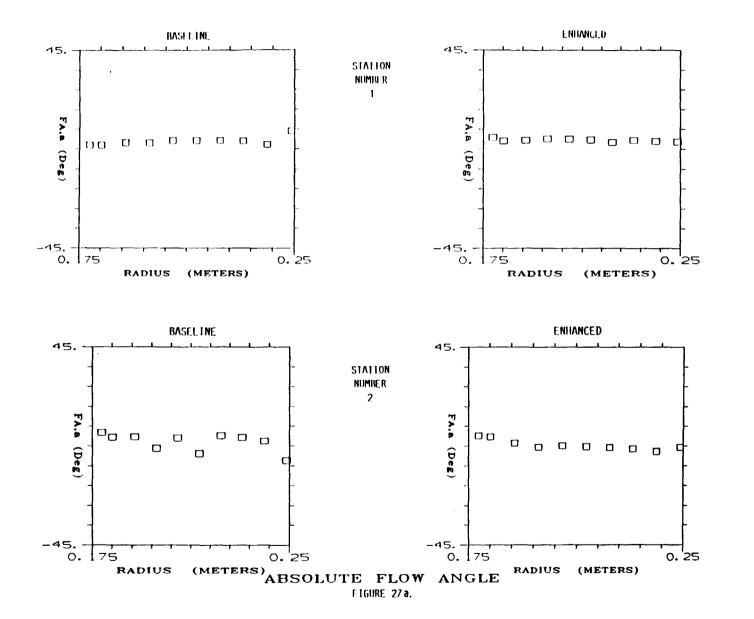


(b) STATION 7. FIGURE 26. - CONTINUED.

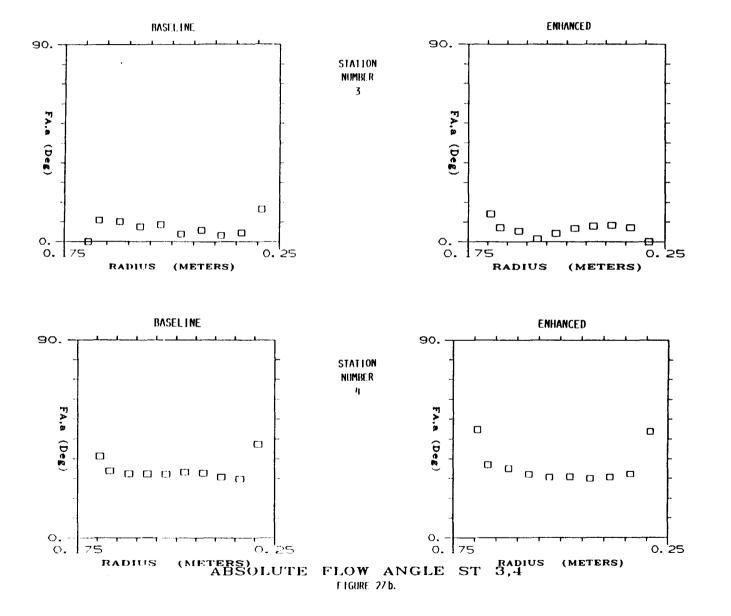


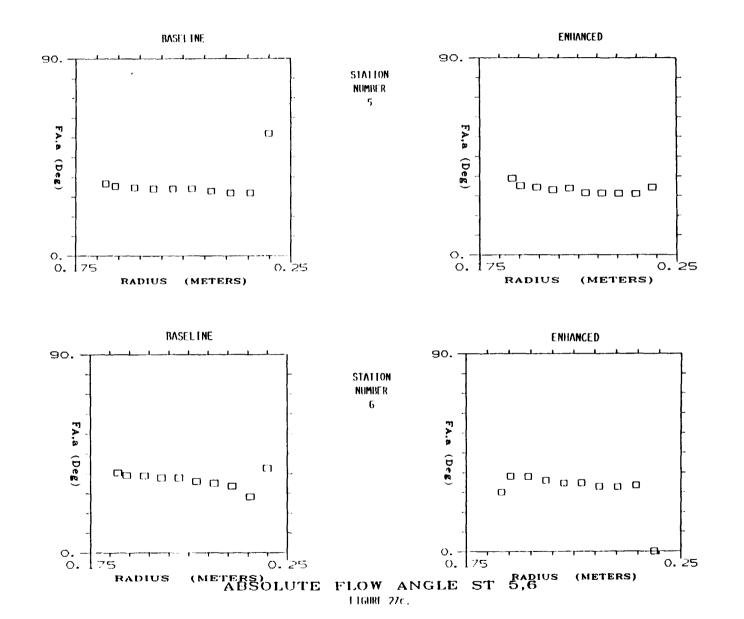


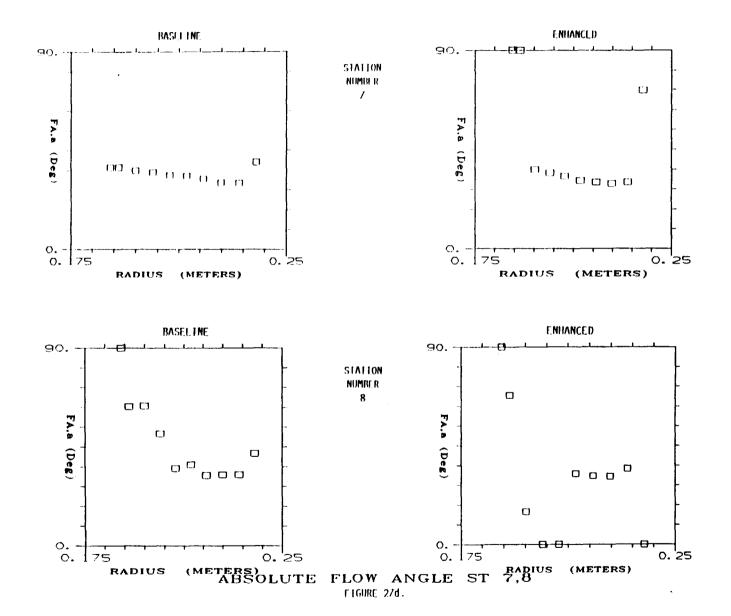
(c) STATION 8.
FIGURE 26. - CONCLUDED.

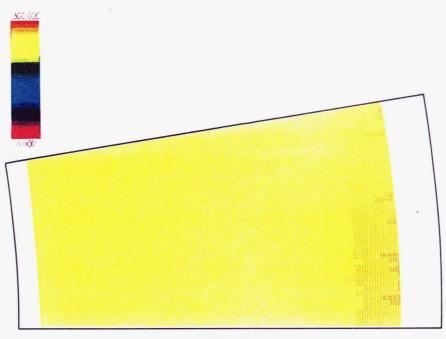




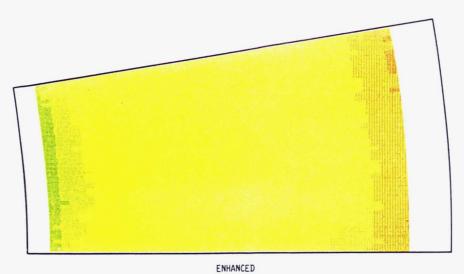






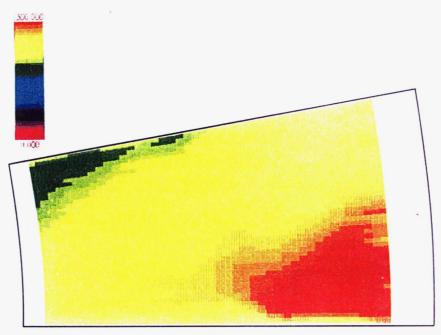


BASELINE

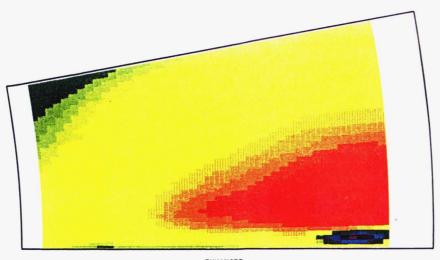


(a) STATION 1.

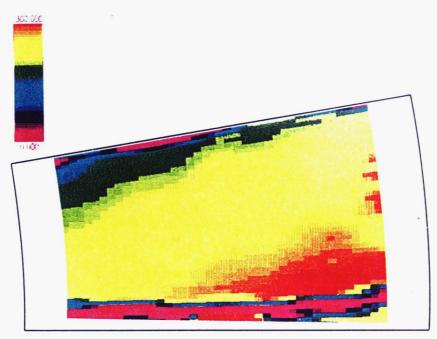
FIGURE 28. - RELATIVE TOTAL VELOCITY, COLORFILL, MPS.



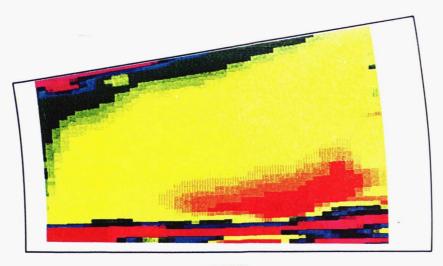
BASELINE



ENHANCED
(b) STATION 2.
FIGURE 28. - CONTINUED.



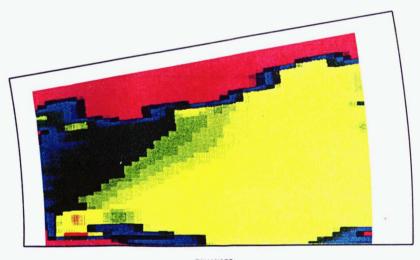
BASELINE



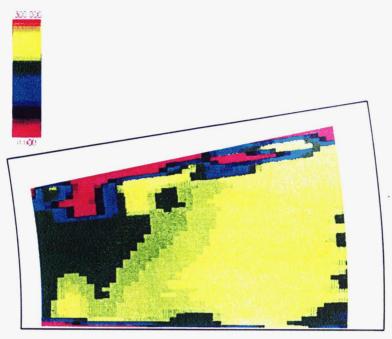
ENHANCED
(c) STATION 3.
FIGURE 28. - CONTINUED.



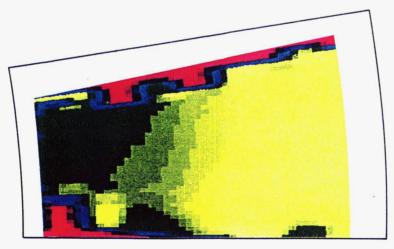
BASELINE



ENHANCED
(d) STATION 4.
FIGURE 28. - CONTINUED.



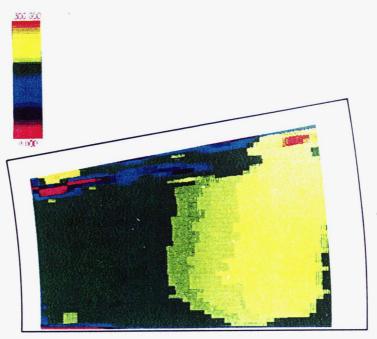
BASELINE



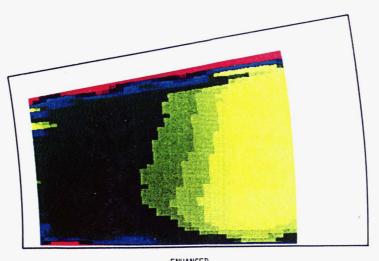
ENHANCED

(e) STATION 5.

FIGURE 28. - CONTINUED.



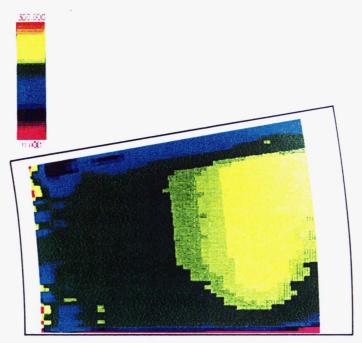
BASELINE



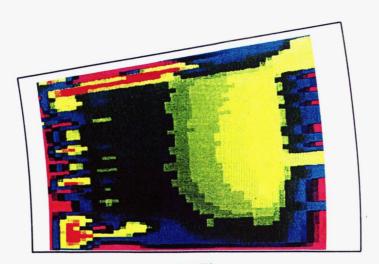
ENHANCED

(f) STATION 6.

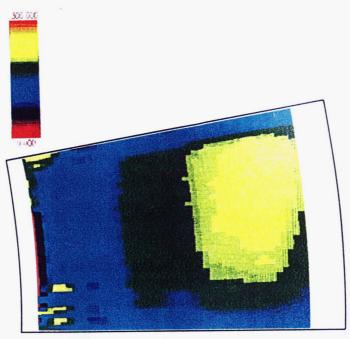
FIGURE 28. - CONTINUED.



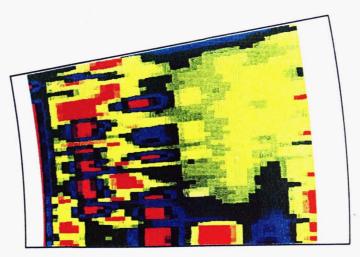
BASELINE



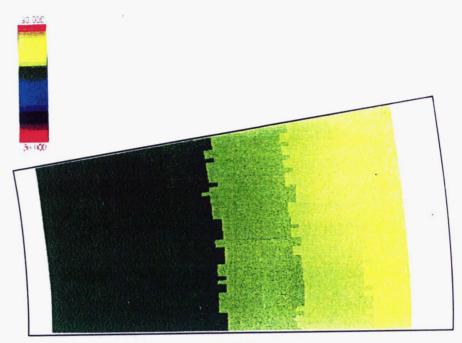
ENHANCED
(g) STATION 7.
FIGURE 28. - CONTINUED.



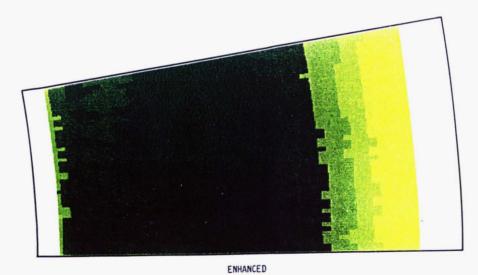
BASELINE



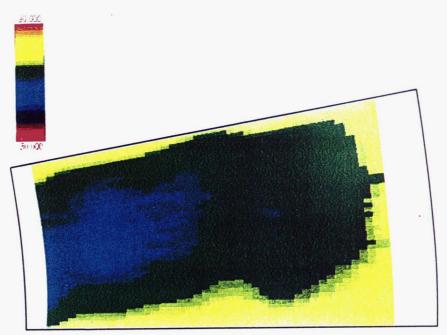
ENHANCED
(h) STATION 8.
FIGURE 28. - CONCLUDED.



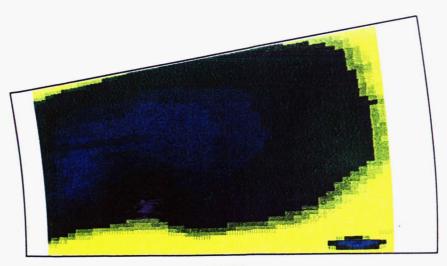
BASELINE



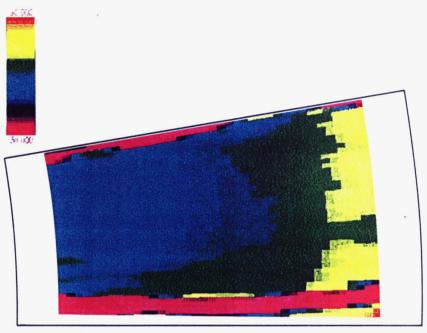
(a) STATION 1.
FIGURE 29. - RELATIVE FLOW ANGLE, COLORFILL, DEG.



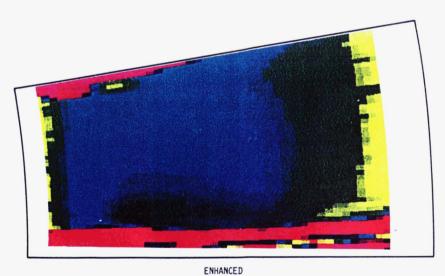
BASELINE



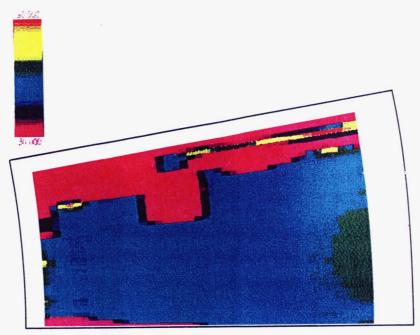
ENHANCED
(b) STATION 2.
FIGURE 29. - CONTINUED.



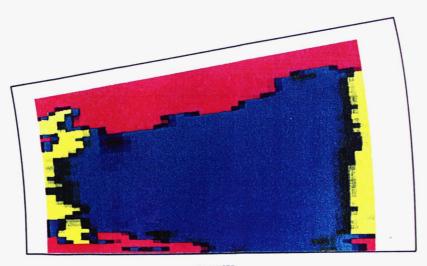
BASELINE



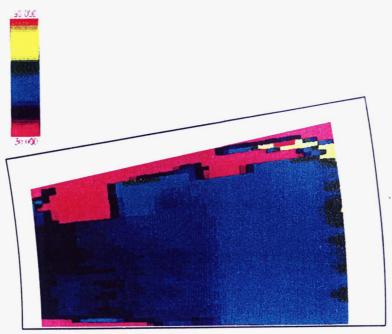
(c) STATION 3.
FIGURE 29. - CONTINUED.



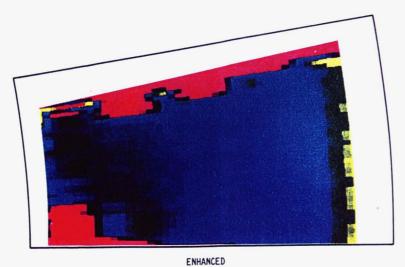
BASELINE



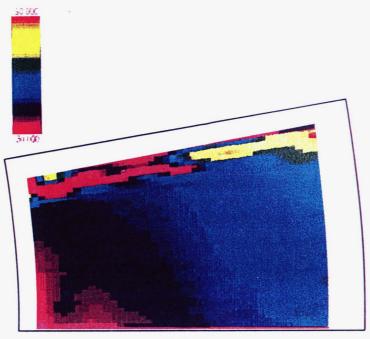
ENHANCED
(d) STATION 4.
FIGURE 29. - CONTINUED.



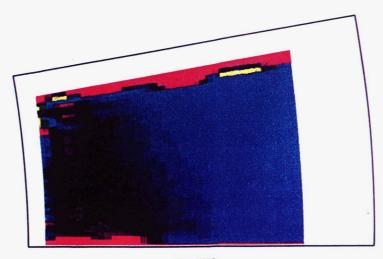
BASELINE



(e) STATION 5.
FIGURE 29. - CONTINUED.



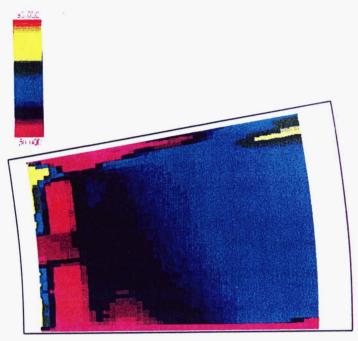
BASELINE



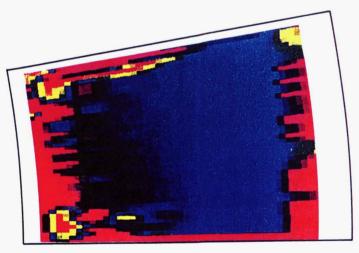
ENHANCED

(f) STATION 6.

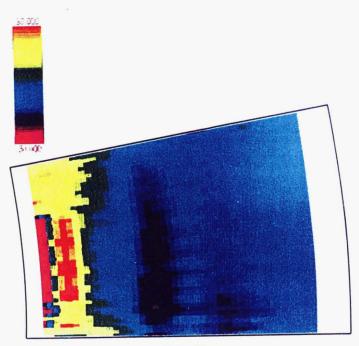
FIGURE 29. - CONTINUED.



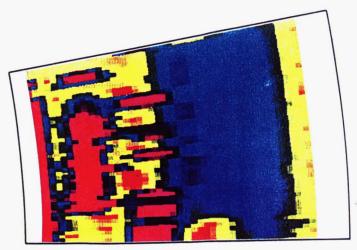
BASELINE



ENHANCED
(g) STATION 7.
FIGURE 29. - CONTINUED.



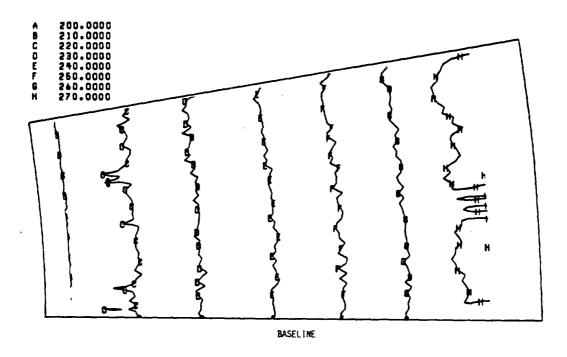
BASELINE



ENHANCED

(h) STATION 8.

FIGURE 29. - CONCLUDED.



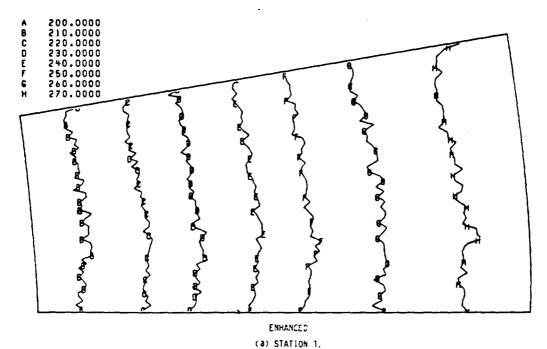
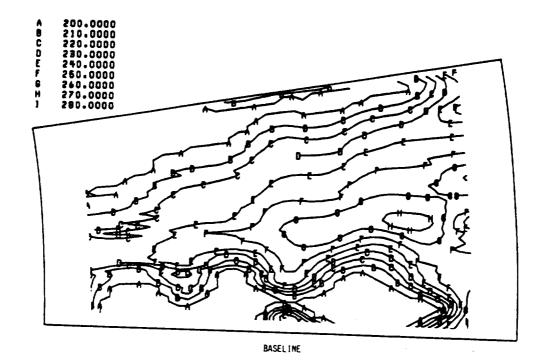
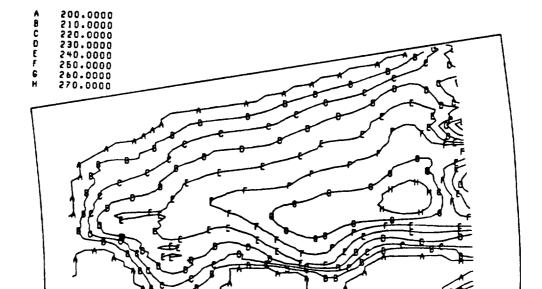
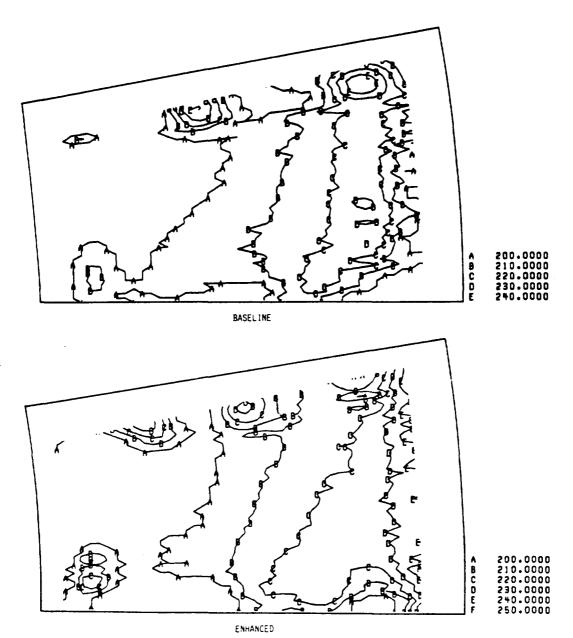


FIGURE 30. - RELATIVE TOTAL VELOCITY, CROSS CHANNEL.

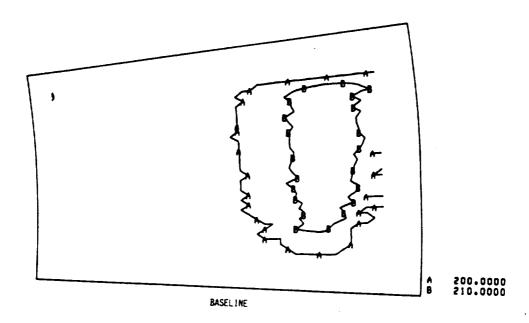


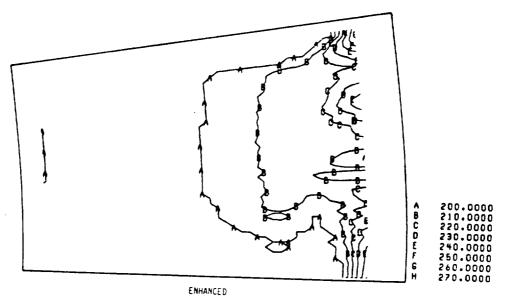


ENHANCED
(D) STATION 3.
FIGURE 30. - CONTINUED.

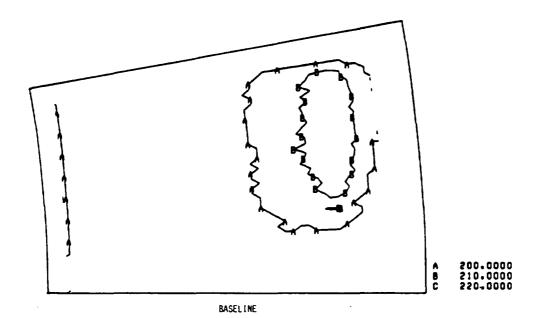


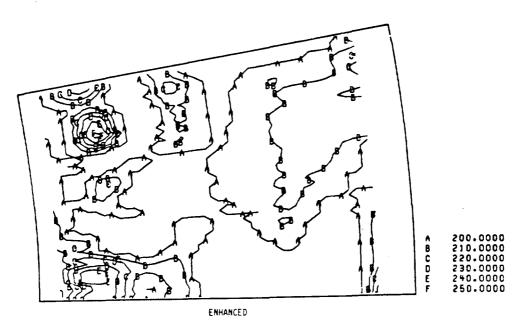
(c) STATION 5.
FIGURE 30. - CONTINUED.



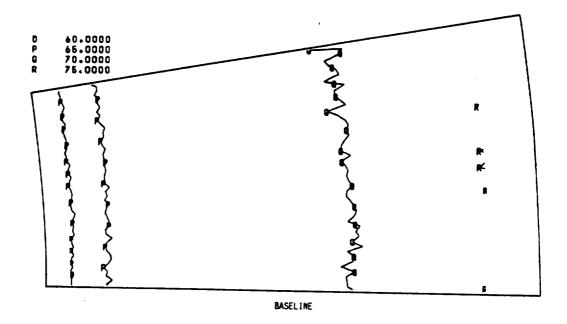


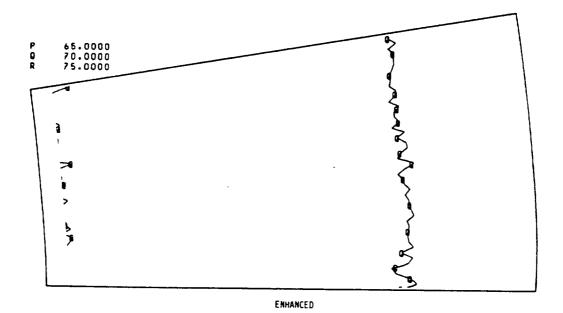
(d) STATION 7.
FIGURE 30. - CONTINUED.



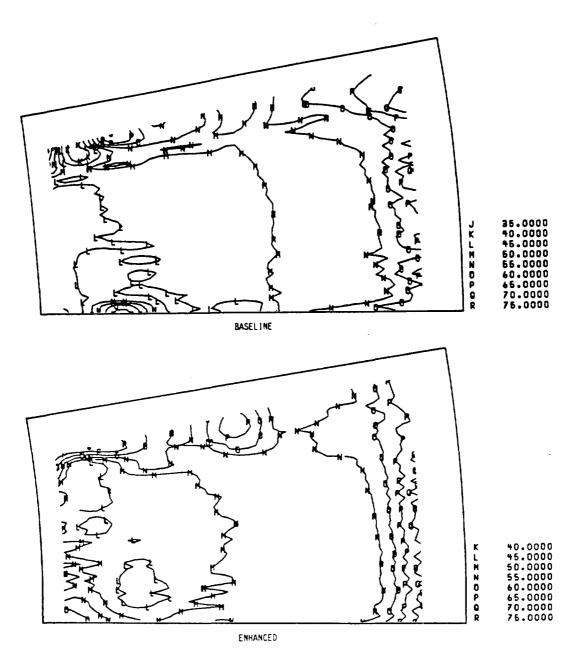


(e) STATION 8.
FIGURE 30. - CONCLUDED.

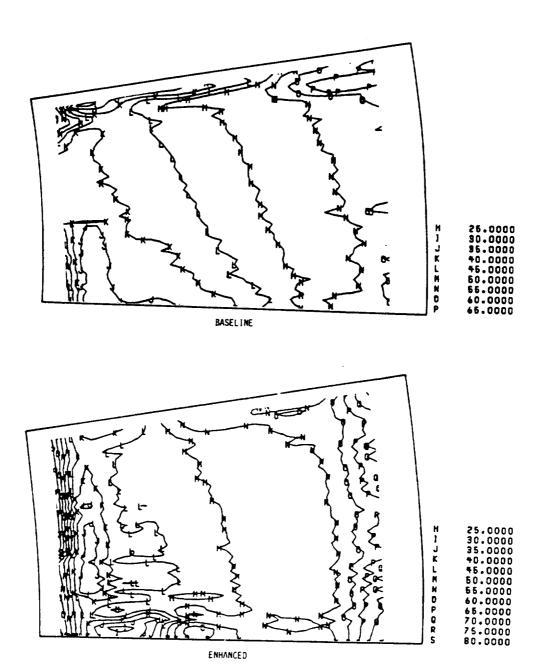




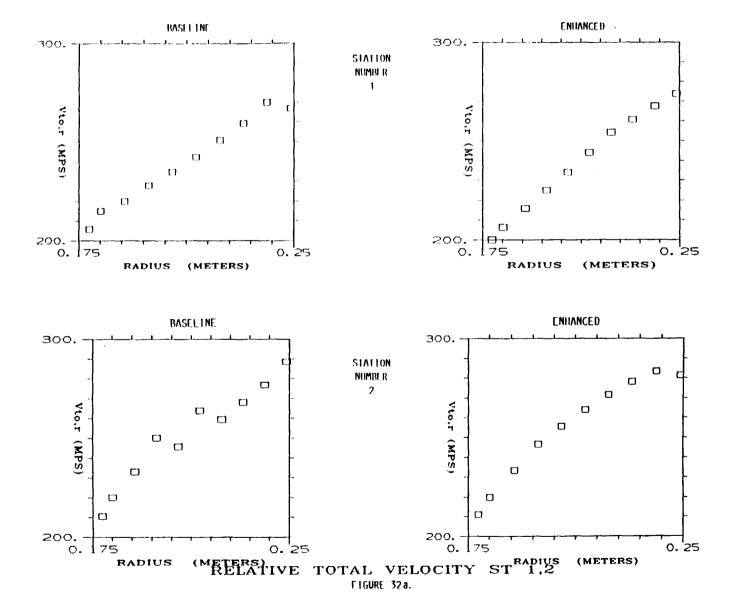
(a) STATION 1.
FIGURE 31. - RELATIVE FLOW ANGLES, CROSS CHANNEL.

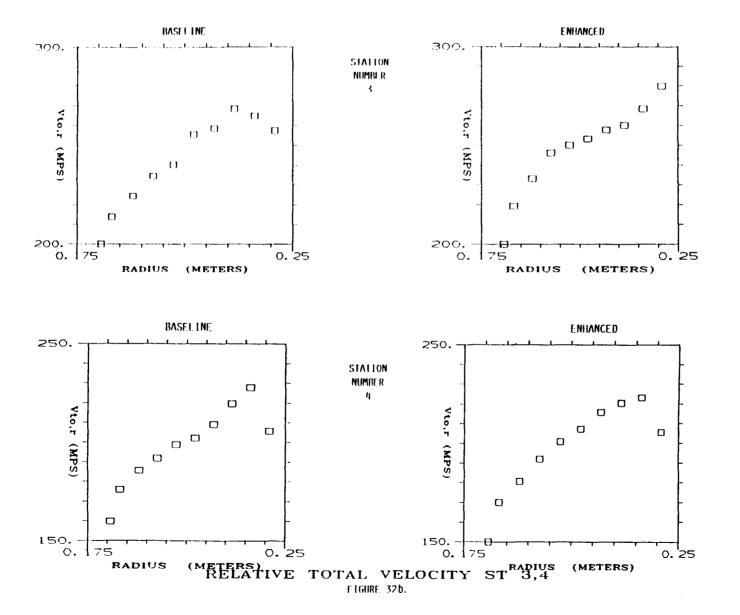


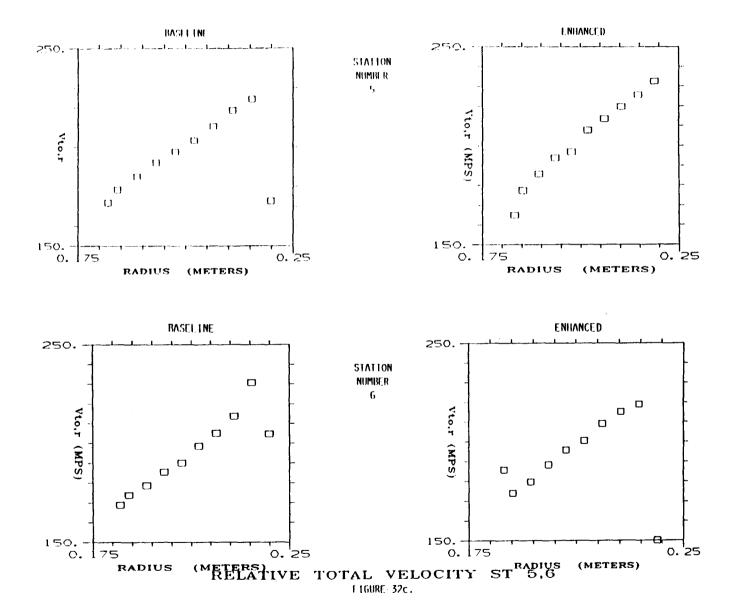
(b) STATION 5.
FIGURE 31. - CONTINUED.

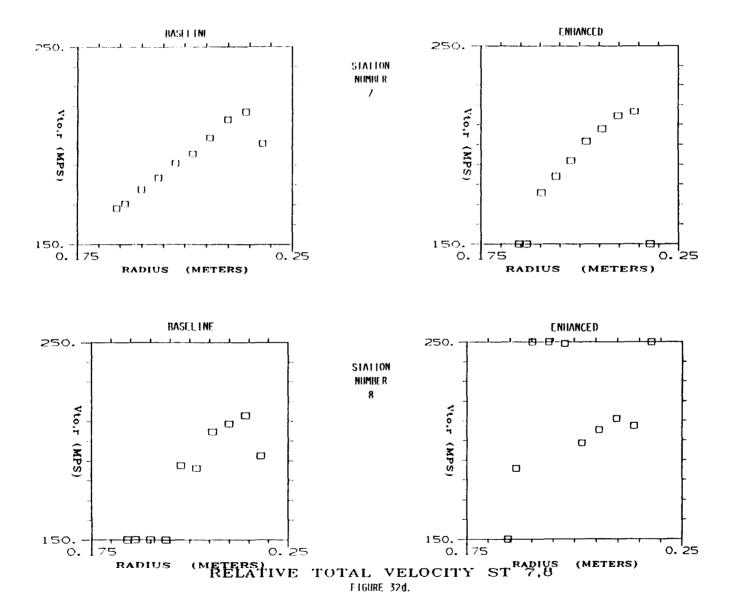


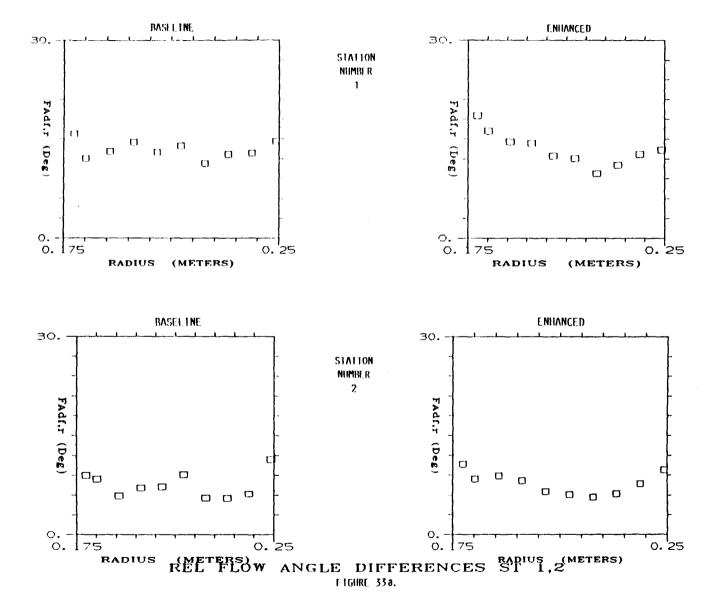
(c) STATION 7.
FIGURE 31. - CONCLUDED.

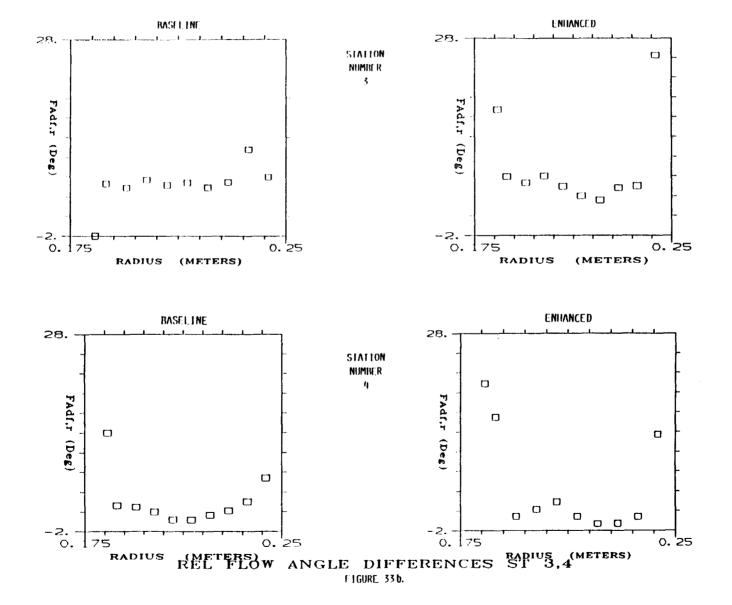


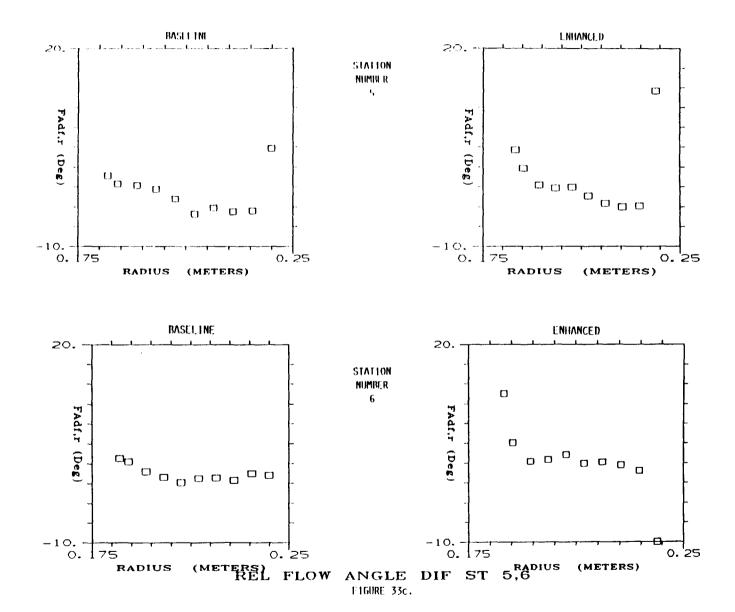


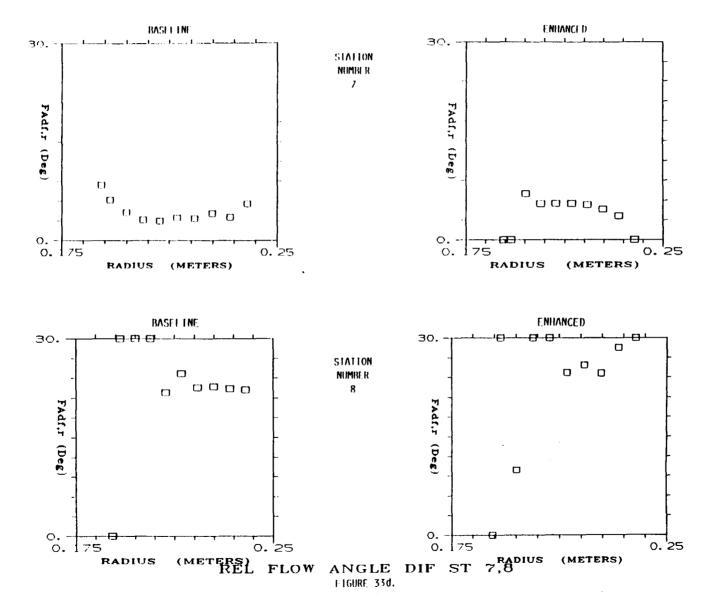


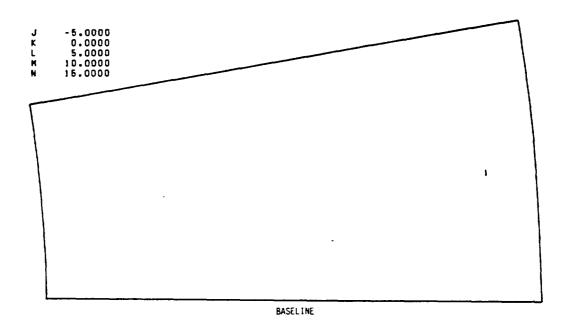












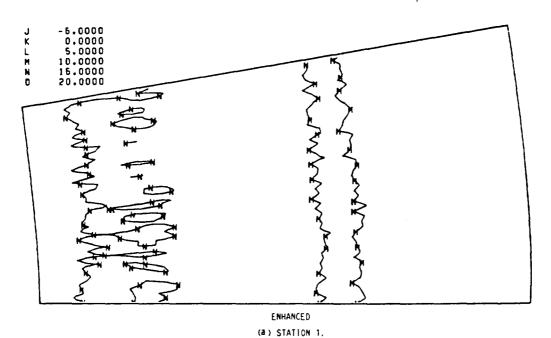
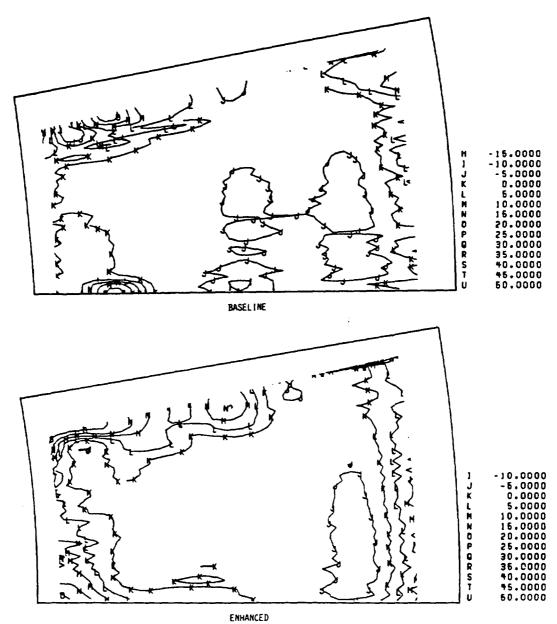
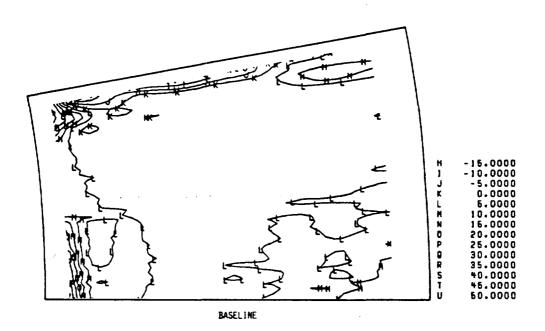


FIGURE 34. - RELATIVE FLOW ANGLE/GRID DIFFERENCE, CROSS CHANNEL.



(b) STATION 5. FIGURE 34. - CONTINUED.



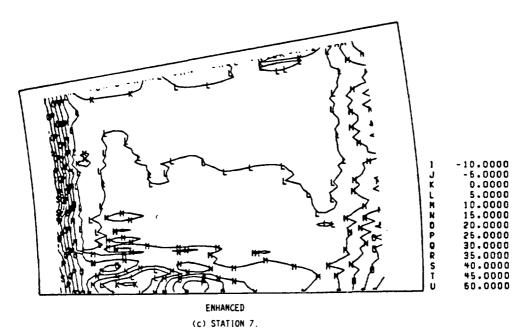


FIGURE 34. - CONCLUDED.

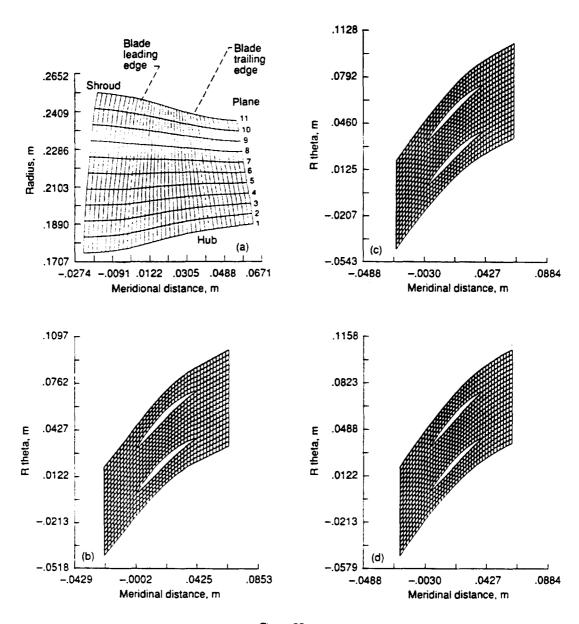


Figure 35.

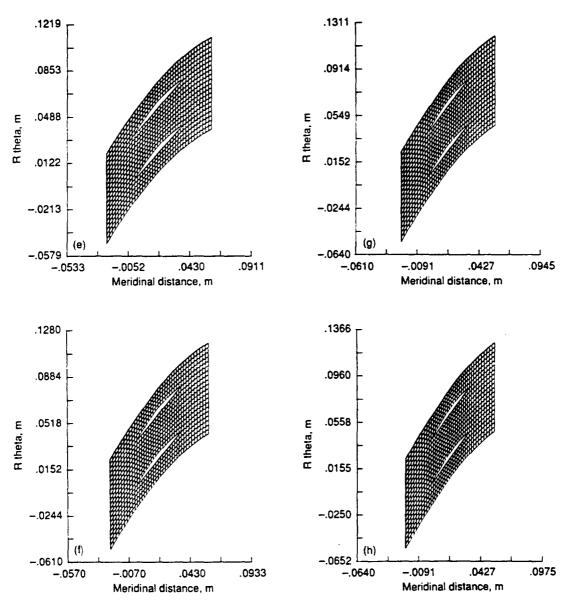


Figure 35.—Continued.

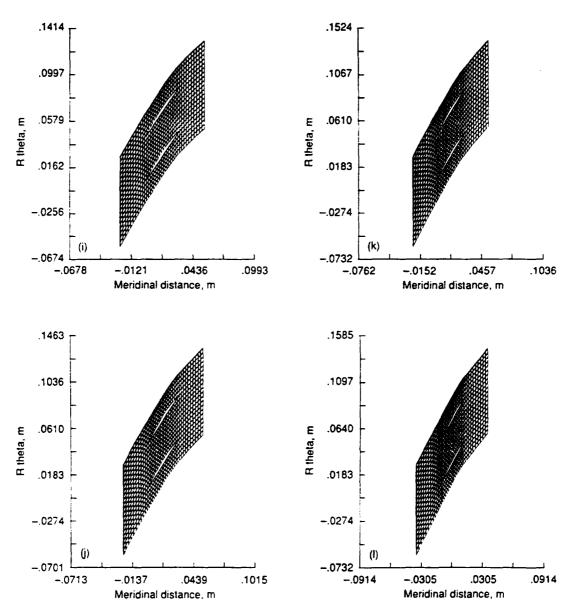
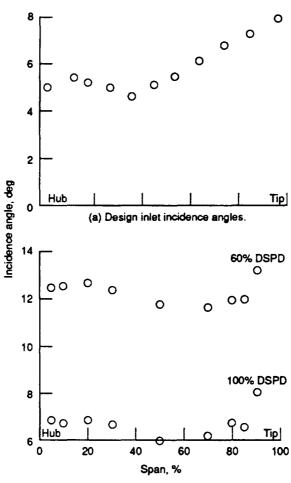


Figure 35.—Concluded.



(b) Peak efficiency incidence angles. Figure 36.—Inlet incidence angle.

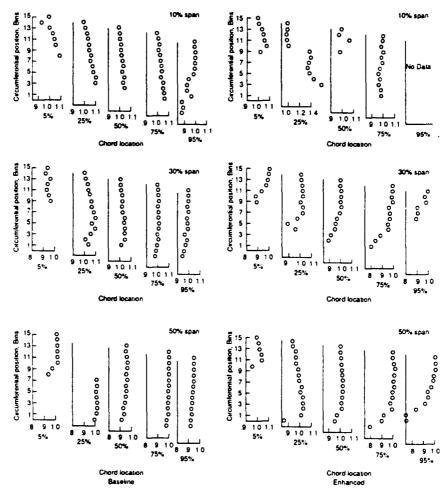
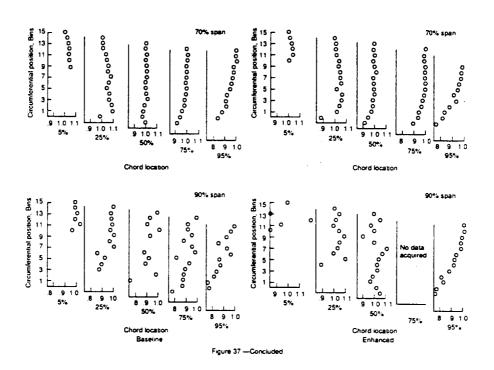


Figure 37 —Non-dimensionalized blade suction surface relative total velocity profiles, VN_{Bint5}



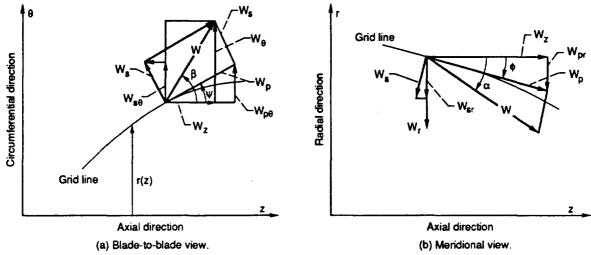


Figure 38.—Flow deviation calculation nomenclature.

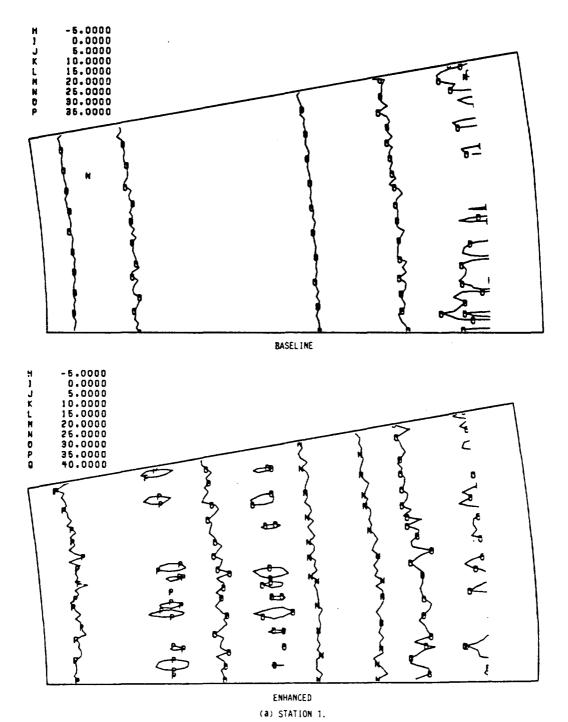
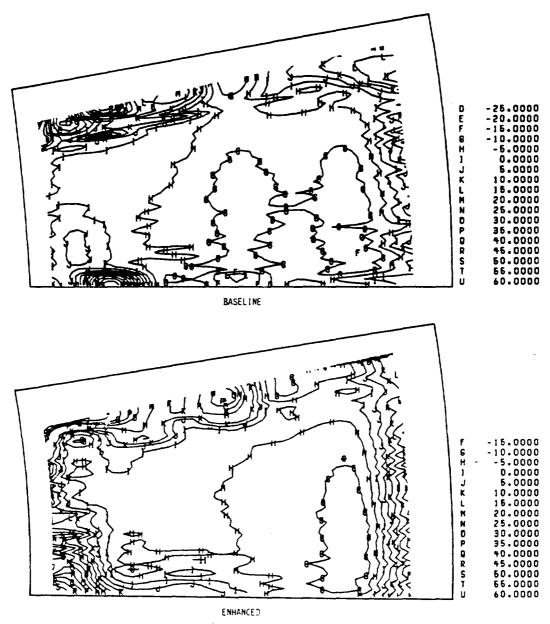
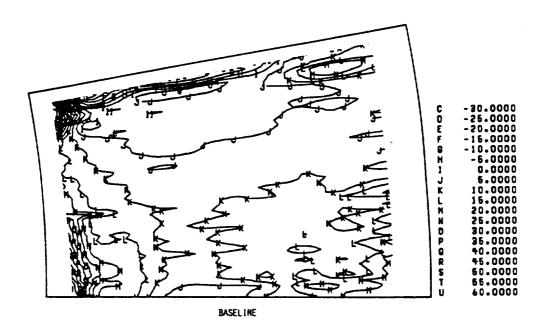
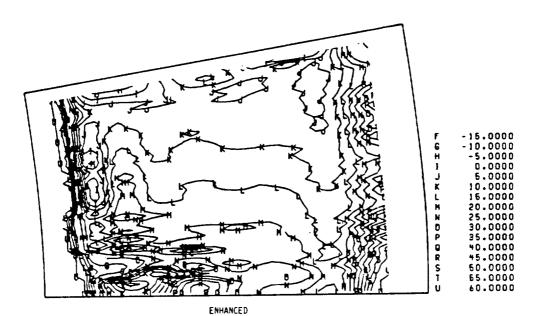


FIGURE 39. - DEVIATION VELOCITIES, CROSS CHANNEL.



(b) STATION 5.
FIGURE 39. - CONTINUED.





(c) STATION 7.
FIGURE 39. - CONCLUDED.

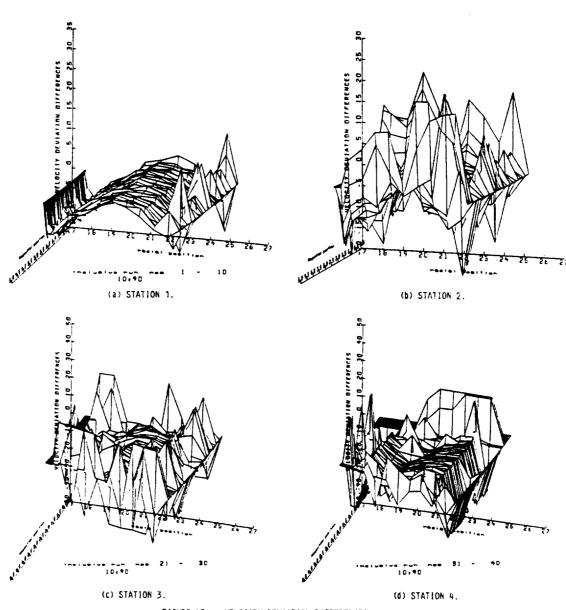


FIGURE 40. - VELOCITY DEVIATION DIFFERENCES, MPS.

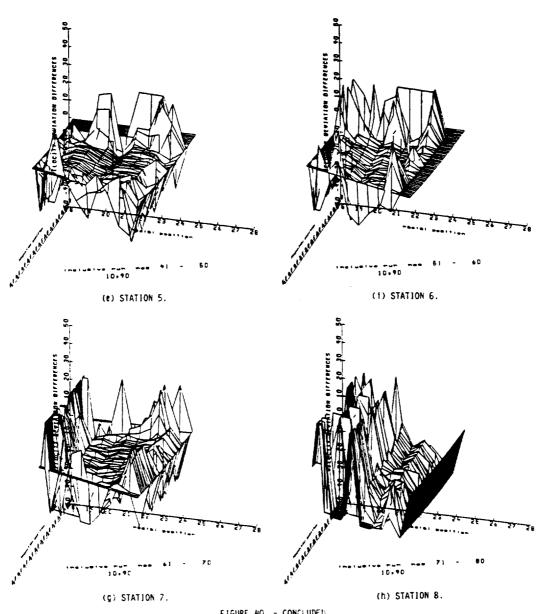


FIGURE 40. - CONCLUDED.

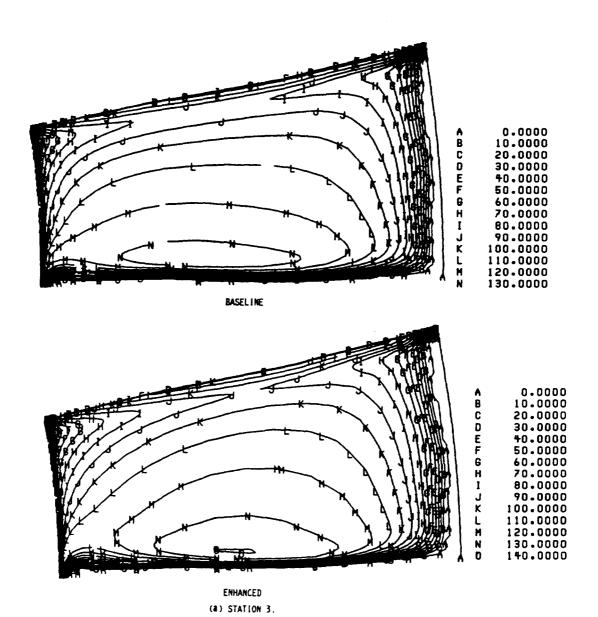
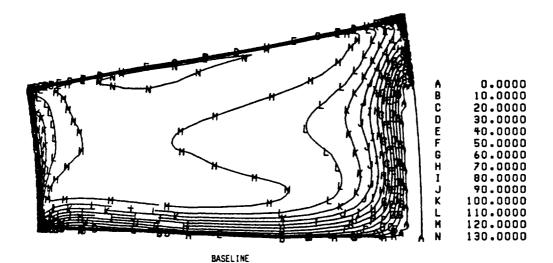
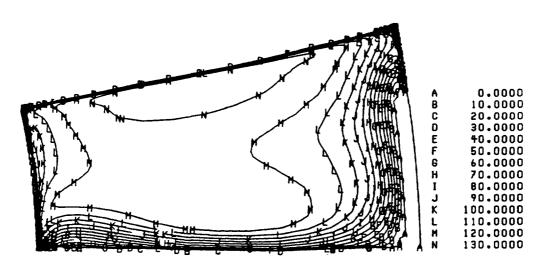
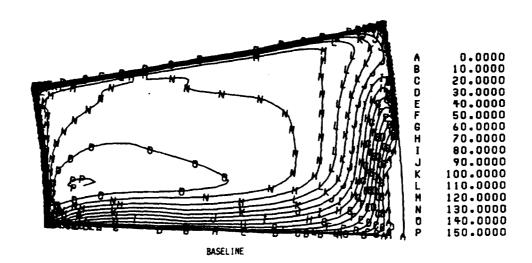


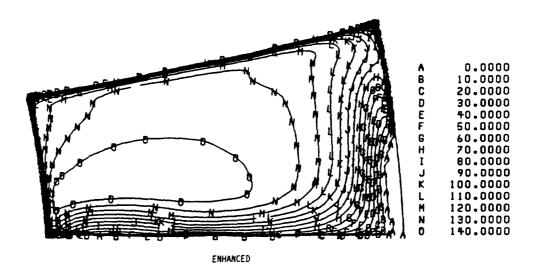
FIGURE 41. - COMPUTED AXIAL VELOCITY, LINE PLOT.



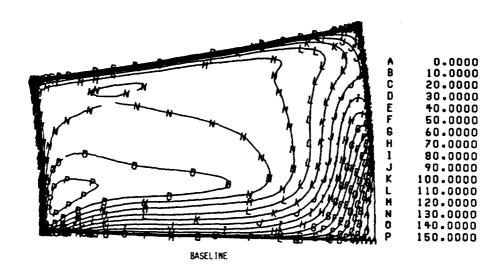


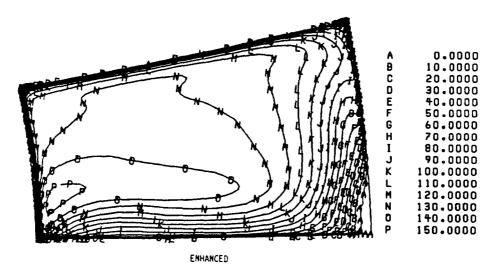
ENHANCED
(b) STATION 4.
FIGURE 41. - CONTINUED.





(c) STATION 5.
FIGURE 41. - CONTINUED.





(d) STATION 6.
FIGURE 41. - CONTINUED.

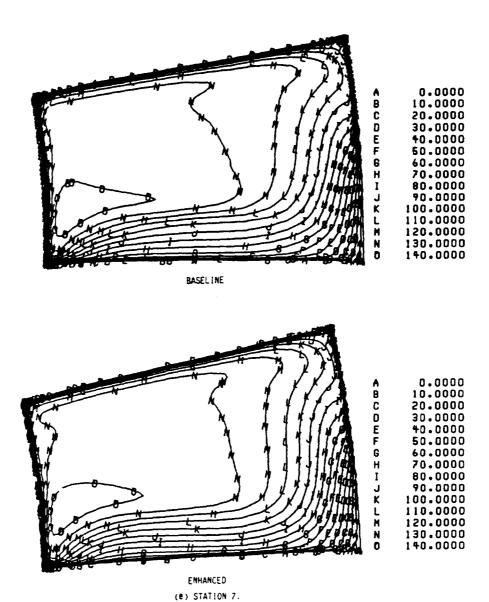


FIGURE 41. - CONCLUDED.

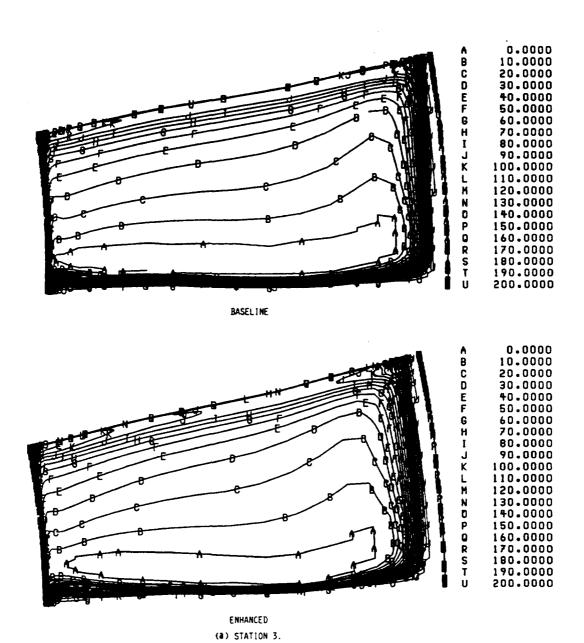
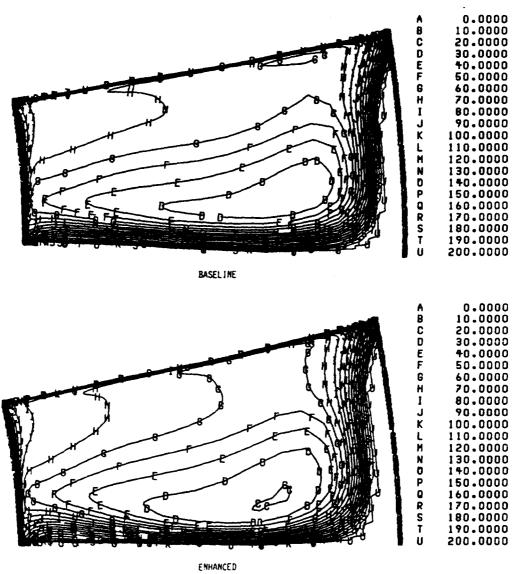


FIGURE 42. - COMPUTED ABSOLUTE TANGENTIAL VELOCITY, LINE PLOTS.



(b) STATION 4.
FIGURE 42. - CONTINUED.

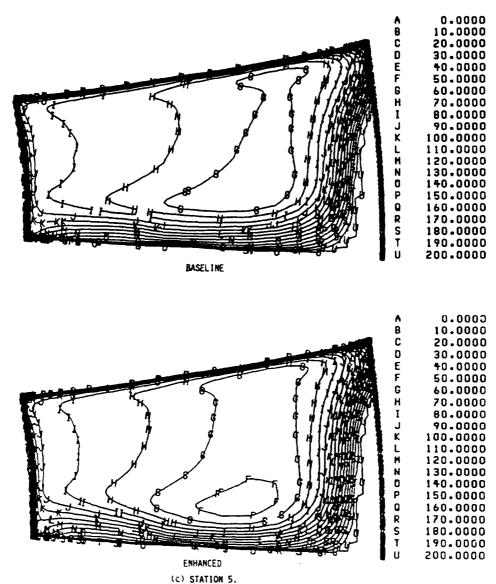


FIGURE 42. - CONTINUED.

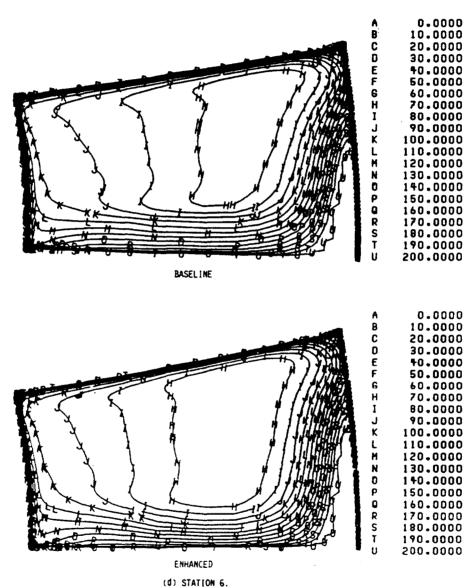
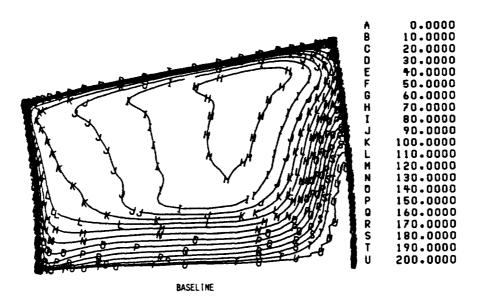
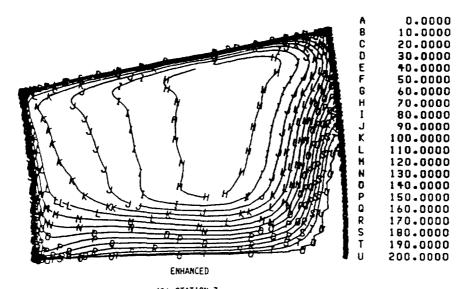


FIGURE 42. - CONTINUED.





(e) STATION 7. FIGURE 42. - CONCLUDED.

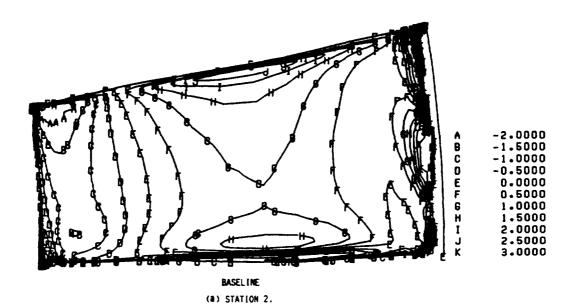
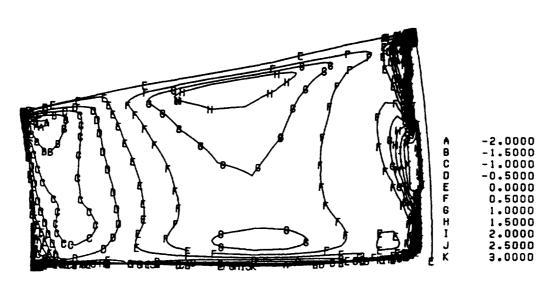
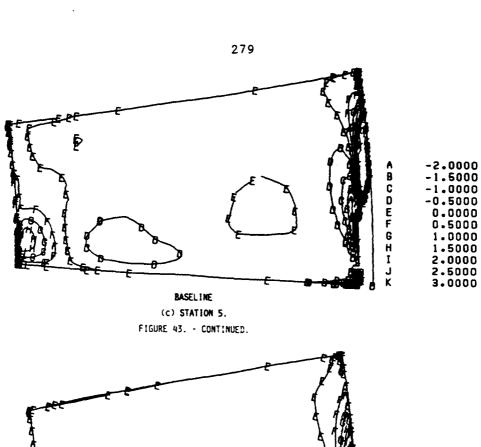
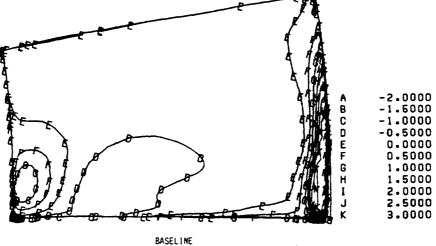


FIGURE 43. - DIFFERENCES BETWEEN COMPUTED TEST CASE RELATIVE FLOW ANGLE DIFFERENCES β_{BASE} - $\beta_{ENHANCED}$



BASELINE
(D) STATION 3.
FIGURE 43. - CONTINUED.





(d) STATION 7.
FIGURE 43. - CONTINUED.

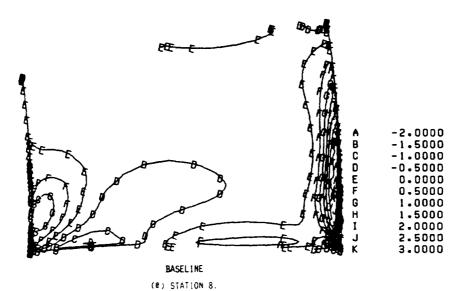


FIGURE 43. - CONCLUDED.

APPENDIX A

Appendix A contains tables of the acquired data and of calculated values of that data that has been used to develop the plots used in this report. Information provided is:

TABLE DATA PROVIDED

- Al Axial Velocities
- A2 Absolute Tangential Velocities
- A3 Number of Axial Measurements
- A4 Number of Tangential Measurements
- A5 Calculated Axial Uncertainties
- A6 Calculated Tangential Uncertainties

TABLE A1. -

(a) AXIAL VELOCITY (M/S)

CP	BASI VA1	ELINE INI VA2	LET, STA	TION 1 VA4	VA5	VA6	VA7	VAS	VA9	VA10-AXIA	L VEL
12345678901234567890123456789012345678901234567890	81.190 78.157 79.440 80.617 79.450 81.472 80.617 79.538 79.62 79.79.79 79.450 80.62 79.79.79 80.27 79.450 80.27 79.450 80.27 79.450 80.27 79.450 80.27 79.450 80.27 80.27 80.27 80.27	92.291 92.391 92.1770 92.1770 92.1770 92.1770 92.153744 92.15374 92.15374 92.15374 92.15374 92.15374 92.15374 92.15374 92.15374 92.1537 92.153	87.79.94.06.20.31.79.88.77.79.51.21.24.42.86.72.01.24.42.86.77.66.20.12.44.57.79.51.24.57.79.51.24.57.79.51.24.57.79.51.25.26.26.26.26.26.26.26.26.26.26.26.26.26.	87.657.32.467.6888.888.8888.8888.8888.8888.8888.8	87.33140 87.33140 86.00	85.414686666666666666666666666666666666666	86.5971885.84478886.5951886.59	85.5.7760 85.4.706 86.4.706 86.5.7760 86	85.545 86.49221	79.33 63.32 74.78 65.63 87.99 81.61 82.51 72.51 72.70 45.05 81.65 81.65 81.65 81.65 81.65 81.65 81.65 81.74 66.22 68.75 74.78 75.06 68.75 74.78 76.05 68.75 74.78 77.38 85.79 66.22 68.75 74.78 77.38 85.79 66.22 66.22 66.23 77.38 85.36 66.23 77.38 85.36 85.37	
CP	BASEL VAl	INE INLE	VA3	VAG	VA5	VA6	VA7	VAB	VA9	VAIO-AXIAL	VEL
125456789012545678901254567890125345678901254567890	71.38 912.39 913.39	61.876.689 92.5089 92.5089 92.5089 1004.6339 1009.5364 1	59.09 72.06 80.39 72.08 80.39 1037.66 23.29 1037.66 21.112.66 88 7.77 11.126 11.126 11.127 11	51.02 59.46 67.77 70.88 59.94 85.32 103.90 111.14 108.16 119.95 119.95 119.95 119.95 119.95 119.95 119.24.98 110.3.99 11	47.44 53.04 69.87 73.32 84.80 89.98 89.98 100.73 101.74 111.84 111.84 111.84 111.84 111.84 111.84 111.84 111.94 111.84 111.9	58, 404 233 66 23 3 2 4 3 0 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	38, 27, 76, 66, 27, 66, 27, 66, 27, 66, 27, 66, 27, 66, 27, 66, 27, 67, 67, 67, 67, 67, 67, 67, 67, 67, 6	32.810 53.863 540.8663 77.872 944.057 84.7057 84.7057 103.688 105.999 103.688 105.999 103.688 105.999 103.688 105.999 103.688 114.778 114.778 114.778 114.778 114.778 115.808 116.8	36.640 39.381 58.864 58.867 58.867 58.867 59.33 90.77 100.550 100.67 100	55. 29 33. 184 37. 64 49. 10 62. 94 57. 80 57. 80 58. 52 58. 52 58. 52 58. 52 66. 61 77. 80 66. 61 77. 76. 64 66. 61 77. 76. 64 67. 76. 64 68. 65 69. 76 69. 76 60. 64	

CP	BASEI VA1	INE IN	LET, STA	TION 3 VA4	VA5	VA6	VA7	VAS	VA9	VA10-AXIAL	VEL
123456789011123456789901123456789901123456789901222222222333356789444444444444444444444444444444444444	į	75550 31 600 32 7 6550 00 000 00 0	32. 81 29. 82 74. 00 0. 00 0. 00 0. 00 0. 00 100. 15 111. 01 132. 81 113. 52 113. 53 1123. 73 1123. 73 1123. 75 1123. 75 1121. 05 1121. 05	7.97 8.149 19.31 0.00 0.00 0.00 121.322 115.369 119.45 1127.66.94 1127.63 1127	27. 06 0.00 71.75 0.00 0.00 0.00 84.59 97.44.55 121.24.45 121.24.10 123.13 120.20 123.13 120.20 123.13 120.20 123.13 120.20 123.13 120.20 123.13 124.48 124.49 125.39 116.45 111.02 111.03 101.	0.00 0.00 54.98 1605.62 0.00 0.00 59.62 74.38 114.82 112.12 112.12 112.31 121.62 122.31 122.31 122.31 122.31 123.10 124.47 124.43 121.62 122.73 121.62 122.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 123.73 121.62 122.73 121.62 123.73 121.62 123.73 121.62 123.73 124.43 125.73 127.63	30. 22 30. 22 30. 07 47. 50 0. 00 10. 00 11. 53 123. 50 124. 61 124. 61 124. 61 125. 49 125. 49 127. 04 119. 34 119. 34 110. 41 110. 41 109. 90 107. 95 107. 96 109. 90 107. 95 109. 90 109. 90 10	0.00 0.00 24.17 46.95 0.00 0.00 121.47 117.54 117.54 117.54 117.54 112.22 119.80 119.80 119.80 119.18 116.77 113.87 110.17 111.80 119.18 110.77 111.80 110.77 111.80 110.77 111.80 110.77 110.77 100.75 100.7	15.00 31.71 37.150 0.00 0.00 58.90 97.26 97.26 97.26 97.85.77 91.12 91.02 97.85.77 91.12 92.58 97.64 9	0.723 16.2289 13.2899 10.000 10.00	
СР	BASEL1 VA1	NE INL	ET, STAT	ION 4 VA4	VA5	VA6	VA7	VAS	VA9	VA10-AXIAL	VEL
12345678901234567890123456789012345678901234567890	1016.5991 1016.5	00.000 00.0000 00.0000 00.0000 00000 0000 0000 00000 0000 00000 0000	0.00 122.36 138.49 138.49 132.68 132.68 132.68 132.19 132.59 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 132.19 133.19 134.22 133.11 135.69 136.17 137.10 137.	0.00 118.66 127.66 118.66 125.29 127.67 138.34 1132.41 133.41 133.41 134	0.00 132.61 131.85 132.160 132.89 123.160 123.	117. 405 113. 082 125. 856 1125. 856 1125. 664 1125. 664 1125. 664 1123. 137 1121. 277 1118. 766 1117. 884 1117. 887 1118. 916 1117. 887 1118. 916 1117. 887 1118. 916 1119. 018 1119. 018 119	134.36 130.109 1128.74 128.74 128.74 128.74 128.74 128.74 128.74 128.74 128.74 128.77	117. 43 129. 44 129. 41 129. 41 129. 41 129. 41 129. 41 129. 41 129. 41 129. 41 129. 42 121. 42 123. 24 123. 24 123. 24 123. 24 124. 42 125. 42 126. 42 127. 42 127	99.86 119.30 119.30 119.30 119.30 110 115.08 1121.65 1112.73 1112.73 1112.30 112.30 112.30 112.30 112.30 112.30 112.30 112.30 112.30 1	52.32 70.46 117.22 117.22 118.88 107.32 94.25 100.09 96.02 112.95 88.40 1100.75 108.05 86.07 93.95 88.17 93.95 93.95 93.95 93.95 93.95 93.95 93.95 93.95 93.95 93.95 93.95 93.	

СР	BASE VA1	LINE INL VA2	ET, STAT	TION 5 VA4	VA5	VA6	VA7	VAB	VA9	VA10-AXIAL	VEL
12345678901234567890123456789012345678901234567890	0.00 0.00 0.00 114,41 113,39 115,74 105,64 107,49 116,77 117,64 112,13 123,88 112,13 127,51 1	0.00 0.00 0.00 142.357 140.91 138.39 137.497 134.97 134.97 134.53 134.45 135.45 135.45 135.25 136.20 137.25	0.00 0.00 79.38 143.22 141.93 139.98 138.21 137.76 135.68 137.76 135.68 136.27 135.69 135.69 135.69 135.69 135.69 135.69 135.69 135.69 136.81	0.00 0.00 14.29 145.31 145.36 141.81	0.00 119.30 132.84 135.63 137.50 136.139 137.78 137.78 138.85 139.79 138.86 135.86 137.86 138.85	0.00 103.461 1158.01 124.632 140.07 1136.752 140.637 1136.752 140.638 1137.03 1138.454 141.533 140.383 1137.405 1137.405 1137.405 1137.405 1137.405 1137.405 1137.405 1137.406 1134.487 1132.77 1133.277 1133.277 1134.487 1135.496 1154.487 1135.496 1154.487 1135.496 1154.487 1135.496 1154.487	0.00 132.10 126.88 131.46 132.23 134.61 132.29 134.61 133.53 134.41 135.49 133.53 135.86 131.28 130.86 131.38 130.86 131.38 130.86 131.68 131.	127.56 115.76 115.76 115.76 129.54 129.54 129.54 133.55 131.37 133.59 133.57 133.59 133.57 133.59 133.57 133.59 133.57 133.59 133.57 133.59 133.57 133.59 13	128.58 91.36 95.99 130.99 127.13 123.69 127.13 126.15 126.15 128.59 131.89 131.89 131.89 132.69 133.42 132.69 133.42 129.65 133.60 129.65 127.79 126.06 127.73 127.12 127.12 127.12 127.12 127.12 128.59 129.65 129.	88.39 83.49 75.24 93.59 80.94 93.59 80.27 108.14 80.27 108.14 80.03 92.73 92.87 84.03 92.73 92.87 77.5.38 84.03 92.73 92.87 77.5.38 92.73 92.87 77.75 85.63 92.73 95.00 101.81 77.86 96.50 97.79 96.50 97.79 96.50 97.79 96.50 97.79 96.50 97.79 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 96.50 97.79 97.89 96.50 97.79 97.89 96.50 97.79 97.89	
CP	BASE!	INE INLE	T, STAT	ION 6 VA4	VAS	VA6	VA7	VA8	VA9	VAIO-AXIAL V	EL
12345678901234567890123456789012345678901234567890	0.000 127,497 1262,7695 11326,252 11326,252 11336,720 11	0.00 1508.568 1444.568 1444.742 1442.742 1442.742 1441.03 1442.94 11441.03 1442.94 11441.03 1442.03 1443.03 14	0.00 1521.865 1150.825 1150.825 1150.825 1147.71 1146.864 1147.71 1145.45 1147.71 1145.45 1147.71 1145.45 1147.71 1145.45 1147.48 1147.71 1147.48 1147.71 1147.48 1147.71 1147.48 1147.48 1147.48 1147.48 1147.48 1147.48 1147.49 1147	0.01 146.734 147.133 146.722 147.133 146.722 147.232 147.232 1447.332 1447.	0.09 1340.491 140.491 140.5647 140.5647 142.2739 1442.473 1442.173 1443.173	0.00 131.84 132.84 135.87 135.	0.00 132.294 132.294 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 133.22.494 123.23.294 12	0.06 136.49 136.89 1386.81 137.51 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.61 1387.72 1387.72 1387.72 1387.72 1387.73 1387	0.00 114.45 114.45 114.45 115.27 115.	0.00 79.75 80.35 89.37 99.77 98.64 99.87 98.64 98.62 98.62 98.71 98.62 88.71 98.62 98.85 88.85 98.86 98.85 88.85 1001.81 108.88 88.85 1001.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 101.83 105.86 100.80 100.80 100.80 100.80 100.80	

BASELINE INLET, STATION 7
CP VAL VAZ VAS VA4 VA5 VA6 VA7 VAB VA9 VA10-AXIAL VEL

12345678901234567890123456789012345678901234567890	0.00 0.00 1.04 66.48 7.4.51 22.2.40 128.33 24.50 24.50 24.50 25.50 26.50	0 00 119 47 118 75 120 94 118 75 120 94 118 75 127 84 1137 92 127 84 1139 97 140 25 139 97 140 25 139 97 140 25 139 97 140 85 139 97 140 85 139 97 140 85 150 85 15	0.00 0.00 150.16 147.32 146.43 144.40 144.41 145.79 143.98 141.20 143.98 141.20 143.98 141.20 143.98 141.39 143.98 141.39 143.98 141.39 143.98	0.00 145.09 143.50 145.37 145.10 142.43 143.54 143.54 142.94 141.62 142.53 159.64 1139.73 159.75 159	0.00 192.58 180.89 134.94 134.53 135.84 135.85 137.99 137.81 136.31 136.	0.00 124.57 124.07 124.07 124.07 129.56 129.57 133.74 132.57 133.74 132.57 133.39 133.74 135.57 132.59 133.39 132.59 133.39 132.59 133.39 132.59 133.39 132.59 133.39 132.59 132.59 132.59 132.59 132.59 132.59 132.59 132.59 132.59 132.59 132.59 133.13 132.59 132.59 132.59 133.13 132.59 132.59 133.13 132.59 132.59 133.13 132.59 132.59 133.13 132.59 132.59 133.13 134.59 129.59 129.50 129.	0.00 114.94 117.657 128.23 128.25 129.33 130.83 130.83 130.83 130.83 130.83 130.83 130.83 130.83 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.62 129.33 128.63 129.33 128.63 129.33 128.63 129.33 128.63 129.33 128.63 129.33 129	8.80 194.00 119.79 119.79 119.33 124.87 127.34 127.34 128.04 128.04 128.15 128.89 127.48 128.15 128.89 127.48 128.15 128.89 127.48 128.15 128.69 127.56 127.56 127.56 128.15	0.00 0.00 96.46 97.08 119.59 112.00 121.54 122.123 124.124 126.52 126.52 126.52 126.53 127.72 127.73 12	0.00 0.00 76.29 74.17 78.82 84.90 98.40 99.56 92.35 92.47 102.49 100.50 102.50 96.16 105.50 104.63 102.29 97.66 105.07 106.09 110.63 107.73 107.
СР	BASE VA1	LINE INL	ET, STAT	ION 8 VA4	VAS	VA6	VA7	VAB	VA9	VAIO-AXIAL VEL
12345678901234567890123456789012345678901234567890	34.77 11.77 11.77 12.77 13.29 18.31 0.000 90.39 59.200 0.000	128.647 38.96.97 51.70668 52.70667 53.917 53.96.97 53.97 54.9	39.87 42.29 40.48 24.76 34.08 48.11 51.87	44.48 50.265.372 65.372 65.873 65.873 66.873 67.764 67.774 67.775 68.772 68.772 68.772 68.772 68.773 68.	112.69 80.45 77.80 92.58 132.63 136.42 131.34 140.13 1	63.10 66.26 77.89 66.95 86.15 108.47 95.09 95.09 1123.60 1113.46 1123.69 127.02 113.47 113.48 122.94 123.48 122.94 123.47 123.48 122.94 123.47 123.47 124.49 125.94 127.80	84.24 93.46 97.14 104.03 117.71 125.43 120.21 125.43 130.29 134.02 134.02 134.02 134.02 134.02 134.02 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 134.03 135.03 136.03 137.0	58.69 80.05 95.69 88.60 95.73 117.87 120.52 116.04 127.25 121.130.94 129.125.84 129.427 131.131.131.131.132.65 131.131.95	63.03 72.23 77.23 91.367 102.65 107.67 102.65 107.67 1124.93 1124.93 1124.93 1124.93 1124.93 1125.58 1127.58 1127.58 1128.55 1128.55 1128.55 1128.56 1	65.64 66.93 71.70 76.60 76.06 82.93 84.13 89.01 93.26 88.32 97.73 88.71 99.23 88.71 105.28 101.61 105.28 102.76 110.77 112.33 1102.76 113.82 114.96 115.78 115.78 115.78 116.55 117.84 116.77 117.85 117.86 115.78 116.10 117.86 117.86 117.86 117.86 117.87 110.63

	ENHA	NCED INL	ET, STAT	IDN 1					NAO	WATE-AVIAL VEL
CP 1	VAI	VAZ	VAS	PAV	VA5 90.10	VA6 94.02	VA7 95.44	VA8 90.02	VA9 82.96	VAIO-AXIAL VEL
12345678901234567890123456789012345678901234567890	65. 66. 85. 67. 68. 65. 69. 99. 64. 85. 65. 65. 65. 65. 65. 65. 65. 65. 65. 6	76.09 76.09 76.73 76.74 76.12 76.12 76.18 76.18 77.5 77.5 77.5 77.5 77.5 77.5 77.5 77.	80.403 80.504 80.504 80.504 80.504 80.504 80.779.784 80.8779.784 80.8779.884 81.779.884 81.779.884 81.779.884 81.779.884 81.779.884 81.779.884 81.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.884 81.779.779.779.884	86.00714.458.866.487.500714.458.866.488.866.4888.866.4888.866.4888.866.48888.866.488888888	90.175 90.265 90.265 99.268 89.268 89.277 89.347 88.802 90.177 89.347 88.802 90.135 88.802 90.135 88.802 90.135 89.91 89	94.08 94.08 94.57 94.146 97.14	954.492 954.691 954.691 954.691 954.691 954.691 954.691 955.86	90.89 91.615 88.82 92.019 92.109 92.109 92.109 92.109 91.147 92.204 91.147 92.204 92.37 92.204 92.38 99.75 92.38 99.75 99.85 99.85 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87 99.87	82.960 83.711 83.711 83.996 85.17.881 84.295 84.295 84.295 84.295 84.295 84.295 84.295 84.295 84.295 84.295 84.295 84.295 82.355 84.83 82.494 83.188 84.646 84.295 84.295 85.188 86.466 86.6666 86.666 8666	73.72 75.76 75.47 75.47 76.69 76.69 74.61 74.61 74.61 75.20 75.20 76.67 76.67 77.67 78.99 78.99 78.99 78.99 78.99 78.99 78.99 78.75 78.99 78.75 78.99 78.75 78.99 78.75 78.99 78.75 78.75 78.99 78.75 78.75 78.75 78.75 78.75 78.75 78.75 78.75 78.75 78.75 79.75 78.75 79.75
CP	ENHA VA1	NCED INLI VA2	ET, STAT VA3	ION 2 VA4	VA5	VA6	VA7	VA8	VA9	VA10-AXIAL VEL
123456789012345678901234567890123456789012345678901234567890	49.43 56.93 71.82 82.88 95.87 99.87 89.01 104.18 105.10 104.18 105.10 104.18 105.10 104.18 105.10 104.18 105.10 104.18 105.10 104.18 105.10 104.18 105.10 105.10 106.10 107.77 107.87 107.87 108.87 109.8	47. 26 43. 80 59. 76 66. 85 76. 85 106. 87 110. 93 110. 93 110. 93 110. 93 110. 93 110. 83 110. 85 103. 35 104. 83 105. 85 106. 82 107. 85 108. 85 109. 76 103. 85 104. 85 105. 85 107. 85 108. 85 109. 76 109. 76	50.47 44.75 44.75 55.58 59.58 66.60 77.32 88.58 91.19 97.82 101.45 105.45 107.83 111.97 111.28 110.27 111.28 110.27 111.28 110.36	54.55 56.22 55.62 55.63 57.80 57	58.70 42.65 42.65 42.65 43.75 100.81 100.83 110.36 1113.69 1121.87 122.87 123.82 124.91 123.82 124.83 121.23.82 127.28 128.83 121.24 123.82 127.28 128.83 121.29 127.73 128.66 124.69 128.87 12	63.89 44.65 45.57 74.34 82.78 98.494 119.28 1115.06 1121.21 121.22 122.26 122.26 122.36 123.61 123.61 117.51 118.83 1123.46 1121.88 1121.19 118.83 1123.61 117.51 118.83 1123.61 117.51 118.83 119.99 108.99 108.99 108.99 87.99 87.99 87.29	68.05 47.67 63.00 72.43 80.80 90.86 90.86 101.03 112.25 116.87 118.72 122.86 122.86 122.86 122.86 122.15 122.15 122.15 122.15 122.15 122.15 123.69 123.69 124.123.69 125.107 1	69.15 51.94 44.38 60.92 69.59 96.95 107.46 111.58 110.68 110.68 110.68 110.70 119.39 118.10 118.10 118.10 118.10 118.10 118.20 119.39 1	63.59 54.08 53.68 73.62 73.62 74.73 75.66 75.66 75.75 77.66 77.75 77.66 77.76 77.76	55. 35 47. 74 40. 39 118. 77 35. 35 35. 35 36. 36 37 45. 38 45. 38 46. 49 57. 23 87. 81 98. 49 97. 23 98. 49 99. 38 99. 38 9

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		NCED INL	ET, STAT	10N 3	VAE	WAA		VAR	UAO	VAI N-AYTAI	VEI
CP 12345678901234567890122322222222223333333555544444444445	VAI 0.00 0.0	VA2 0.00 4.29 0.00 0.00 0.00 0.00 130.47 122.94 121.42 121.42 121.42 121.39 112.39 11	VA3 0 000 0 000 0 000 0 000 0 000 0 000 0 000 0 107 499 119 988 120 488 124 568 124 568 127 491 128 499 129 497 128 688 111 349 111 141 110 899 110 305 99 41 88 668 85 541 88 551 88 65 54	VA4 0.00 7.20 3.01 0.00 0.00 0.00 0.00 0.00 78.84 99.20 114.825 115.34 115.34 112.47 122.47 122.47 122.47 123.71 121.42 123.33 115.91 121.42 122.33 115.91 121.42 122.33 115.91 121.42 122.93 123.71 121.42 122.93 123.71 121.42 122.93 123.71 121.42 122.93 123.71 121.42 122.93 123.71 121.42 122.93 123.71 121.42 122.93 123.91 124.62 125.93	VA5 0.00 0.00 17.213 0.00 0.00 17.213 0.00 0.00 121.30 127.17 120.97 127.43 126.64 137.75 128.16 139.85 129.67 128.57 129.57 128.57 129.57 128.57 129.57 128.57 129.57 129.57 129.57 129.57 129.57 129.57 129.70 0.00	VA6 0.00 91.71 2.52 0.00 99.16 100.00 128.03 130.53 133.24 133.2	VA7 0.00 0.00 0.56 0.00 0.88.70 131.33.68 134.86.23 135.73 135.73 135.74 135.36 134.86 134.86 134.86 134.86 134.86 135.73 136.87 127.87 127.87 127.87 121.33.69 125.54 126.56 127.87 121.33.69 125.54 126.56 127.87 127.87 121.38 126.36 127.87 127.87 127.87 128.55 129.55 129.55 129.55 130.87 130.	VAB 0.000 0.000 0.000 2.553 8.07 0.000 110.18 913.59 117.02 116.513 116.91 117.18 117	VA9 0.02 1.02 0.00 2.12 0.00 0.00 43.46 48.13 78.79 113.78 109.69 114.56 111.72 116.50 111.00 115.69 111.00 110.07 158 110.08 105.26 105.26 85.55	VAIO-AXX 18.86 10.52 10.400 27.747 00.000 148.01 1532.132 460.930 148.158 461.991 148.158 461.991 148.158 172.757 172.758 188.490 172.758 188.490 172.828	YEL
СР	ENHA VA1	NCED INL	ET, STAT VA3	ION 4 VA4	VA5	VA6	VA7	VAB	VA9	VA10-AXIAL	VEL
12345678901234567890123456789012345678901234567890	0.000 0.000	0.00 0.00 157.74 38.27 74.13 99.26.65 55.59 66.67.08 66.67.08 67.08 67.08 67.08 67.08 68.37 68.37 69.20 69.2	0.00 132.63 131.03 127.76 133.25 127.71 131.20 132.35 126.34 122.38 126.05 125.68 127.71 131.50 126.05 125.68 126.05 127.16 132.38 126.05 125.68 126.05 127.17 14.89 115.68 116.33 116.57 117.04 118.90 116.33 116.3	0.00 52.45 49.47 120.22 95.88 97.116.25 122.25 122.25 122.26 123.27 124.88 117.58 117.58 117.58 117.59 116.49 116.49 116.49 116.49 116.49 116.49 116.49 117.59 117.	0.00 151.55 0.00 109.57 123.24 119.41 128.35 127.37 121.99 121.62 124.72 121.99 121.62 124.77 118.76 118.76 118.35 119.41 114.97 116.38 118.35 119.46 118.35 119.46 118.35 119.46 118.35 119.46 118.35 119.46 118.35 119.46 118.35 119.46 118.35	96.51 133.58 136.51 134.42 129.62 129.62 129.61 122.78 124.78 123.78 124.78 123.78 124.78 123.78 124.78 124.78 125.49 127.78 129.61 129.78 129.61 129.78 129.61 129.78 129	124 - 52 134 - 51 136 - 26 135 - 89 136 - 85 128 - 859 129 - 84 129 - 84 129 - 84 129 - 125 125 - 52 122 - 23 121 - 23 122 - 24 123 - 25 124 - 85 125 - 69 126 - 69 127 - 69 128 - 69 129 - 69 121 - 69 121 - 70 122 - 70 123 - 70 124 - 86 125 - 70 126 - 70 127 - 70 128 - 70 129 - 70 120 - 70 120 - 70 120 - 70 121 - 70 122 - 70 123 - 70 124 - 80 125 - 70 126 - 70 127 - 70 128 - 70 129 - 70 120	106.39 135.97 140.04 121.89 123.42 121.95 124.69 127.09 126.86 127.09 126.86 127.09 121.19 121.19 121.29 121.19 123.23 12	45.50 68.25 68.25 91.02.31 91.03 105.32 119.34 123.20 119.34 121.07 121.07 114.16 117.39 114.29 117.39 114.29 117.39 114.30 113.61 113.	0.00 93.00 0.00	IRL PAGE IS OR QUALITY

CP	ENHA VA]	NCED INL	ET, STAT VAS	ION 5 VA4	VA5	VA6	VA7	VAS	VA9	VA10-AXIAL VEL
1234567890123456789012345678901234567890123345678901234567890	0.00 0.00 92.60 99.03 78.54 86.94 86.94 86.94 87.77.74 86.95 87.51 88.67 101.97 97.86 102.56 104.67 103.56 104.67 107.96 10	0.00 0.00 0.00 0.00 127.00 112.65 130.20 129.16 118.32 130.20 129.16 114.42 119.00 114.43 114	0.00 118.54 113.89 129.37 137.22 134.74 140.71 135.64 136.61 137.80 136.96 136.74 137.80 136.96 136.76 137.80 137.	0.00 113.47 123.42 128.46 137.04 133.12 138.02 140.49 141.40.49 140.16 138.21 138.36 140.26 138.37 123.37 133.74 135.25 133.31 135.26 135.26 135.26 135.26 135.27 136.27 137.42 137.42 137.42 137.42 138.37 106.25 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0	8.00 84.27 95.16.20 117.492 115.05 125.21 125.521 130.33 131.59 133.20 133.20 133.20 132.62 132.62 132.62 132.62 132.62 127.95 127.128 128.129 129.29	0.00 122.87 132.285 134.65 134.65 134.46 135.89 134.25 134.46 135.89 134.25 134.25 134.25 134.25 134.25 134.25 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 134.26 135.283 132.	128 . 28 125 . 28 125 . 30 129 . 89 135 . 86 133 . 13 135 . 86 133 . 13 135 . 87 135 . 16 134 . 95 134 . 97 135 . 16 134 . 97 135 . 87 134 . 97 135 . 87 134 . 97 135 . 87 137 . 97 137 . 97 128 . 62 128 . 60 128	124, 70 129, 34 134, 77 135, 22 135, 92 132, 23 135, 90 136, 12 135, 90 136, 12 135, 90 136, 12 135, 90 136, 12 135, 90 136, 12 135, 90 136, 12 135, 90 136, 12 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 135, 56 137, 77 137, 83 130, 23 131, 83 130, 23 131, 83 130, 23 127, 83 120, 65 121, 72 120, 65 121, 72 120, 65 121, 72 120, 65 121, 73 120, 65 121, 73 120, 65 121, 73 120, 60 120, 7	93. 19 100.22 97. 10 110. 89 1122. 94 122. 94 122. 95 125. 68 135. 18 135. 18 135. 19 133. 53 135. 10 133. 53 135. 10 131. 54 131. 55 130. 98 131. 34 127. 98 131. 12 129. 62 131. 16 127. 98 121. 18 127. 98 121. 18 127. 98 121. 18 127. 99 127. 18 128. 18 129. 52 131. 19 121. 18	73. 29 75. 4.0 80. 391 74. 75. 28 80. 391 74. 75. 28 68. 62 68. 62 68. 62 82. 92 73. 56 76. 04 81. 40 82. 92 73. 56 778. 88 9. 24 775. 88 12 76. 69 10 10 10 10 10 10 10 10 10 10 10 10 10
CP	ENHA VA)	NCED INL VA2	ET, STAT VA3	10N 6 VA4	VAS	VA6	VA7	VAB	VA9	VA10-AXIAL VEL
12345678901234567890123456789012335533344444444445	0.00 96.82 94.99 98.19.7 90.261 107.621 108.848 107.621 1122.123.33 124.420 1120.051 124.30 127.33 128.81 126.78 128.81 126.78 1126.78	0 000 0 000 12 89 133 80 125 348 131 57 132 43 133 26 133 37 133 26 133 30 133	0 00 0 00 131 53 138 73 139 .59 139 .69 142 .75 139 .90 137 .75 137 .75 138 .61 131 .86 131 .86 131 .86 131 .86 131 .86 131 .86 131 .86 131 .86 131 .86 132 .86 131 .86 132 .86 131 .86 132 .86 133 .86 134 .86 138 .86 128 .88 126 .61 127 .72 126 .61 127 .73 126 .61 127 .73 128 .85 129 .65 121 .65 121 .60 128 .65 129 .65 120 .60 120 .6	0 .00 92 .84 115 .23 127 .23 127 .34 129 .34 129 .35 136 .93 137 .75 138 .86 137 .75 137 .87 135 .88 136 .50 135 .53 135 .53 135 .71 135 .88 136 .60 127 .92 128 .66 127 .92 128 .66 127 .92 128 .56 127 .92 128 .56 127 .92 128 .56 129 .03 127 .92 128 .56 128 .56 129 .03 127 .92 128 .56 129 .03 127 .92 128 .86 10 .80 0 .00 0 .00 0 .00 0 .00 0 .00	0.00 110.23 126.02 127.26 131.28 134.94 131.28 134.94 131.38 134.94 131.38 134.94 131.38 134.94 131.38 134.94 131.28 134.94 131.28 134.98 134.98 134.98 127.97 126.99 127.77 126.99 127.99 128.99 129.	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 116.62 125.67 128.89 131.26.58 130.38 130.38 130.29 130.20 129.21 130.05 129.21 130.15 126.87 126.87 127.98 127.98 127.99 120.82 121.31 120.82 1	0.00 102.72 107.97 118.02 126.08 121.26.08 121.25.126.61 123.79 128.02 1	

CP			ET, STAT VAS		VA5	VA6	VA7	BAV	VA9	VA10-AXIAL
123456789012345678901233456789012345678901234567890	85.411 0.0000 0.00	0.00 15.70 168.94 116.01 15.49 115.49 117.38	0.00 27.69 121.76 57.00 139.43 141.95 189.89 137.65 1121.36.42 1130.91 1134.64 1130.91 1134.64 1130.91 1134.64 1132.80	0.00 0.00 41.08 89.91 115.528 69.80 72.83 115.73 128.78 126.52 138.83 120.63 135.80 120.63 135.06 128.48 129.69 135.06 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 127.46 128.48 128.48 129.48 129.48 129.48 129.48 120.00 0.00 0.00	0.00 51.84 58.19 58.19 75.72 64.14 72.36 88.87 118.01 1129.36 129.20 118.01 129.36 129.38 124.97 130.03 126.22 132.81 129.36 129.38 124.97 120.20 128.54 120.20 128.54 120.20 128.54 128.54 128.54 128.54 128.54 128.54 128.54 128.54 128.55 128.66 128.56 128.66 128.56 128.56 128.56 128.56 128.56 128.56 128.56 128.56 128.66 128.56 128	0.00 73.08 93.19 91.39 1164.35 124.42 121.24 122.48 122.41 127.81 126.90 129.17 129.85 129.52 128.67 129.57 129.47 128.69 128.76 128.76 128.76 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69 128.76 128.69	0 00 86 49 90 78 91 78 115 .64 118 .85 120 .99 124 .89 127 .51 126 .64 127 .91 127 .91 127 .91 127 .91 127 .91 128 .69 127 .91 128 .92 129 .92 129 .92 129 .92 120 .92 121 .92 122 .92 123 .99 124 .69 125 .93 127 .92 128 .92 129 .92 129 .92 120 .92 121 .92 122 .92 123 .93 124 .69 125 .93 126 .92 127 .92 128 .92 128 .92 129 .92 120 .92 121 .92 122 .92 123 .93 124 .69 125 .93 126 .92 127 .92 128 .92 129 .92 129 .92 120 .92 121 .92 122 .92 123 .93 124 .99 127 .92 128 .92 129 .92 120 .92 121 .92 122 .92 123 .93 124 .93 125 .93 126 .92 127 .92 128 .92 129 .92 120 .92 121 .92 122 .92 123 .93 124 .93 125 .93 126 .92 127 .92 128 .92 129 .92 120 .92 121 .92 122 .93 123 .93 124 .93 125 .93 127 .93 128 .93 129 .93 120 .93 120 .93 121 .93 122 .93 123 .93 124 .93 125 .93 127 .93 127 .93 128 .93 129 .93 120	0.00 93.81 98.02 100.35 111.35 121.78 121.78 122.78 122.78 123.79 124.88 125.61 124.88 125.65 124.99 125.64 126.21 126.82 126.82 126.82 126.83 125.58 126.83 125.58 126.83 125.58 126.83 127.36 128.83 125.58 126.83 127.36 128.83 125.67 128.83 125.67 129.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83 125.67 120.83	0 00 0 00 0 00 88 76 98 35 97 36 100 08 111 - 43 116 - 12 120 96 121 - 43 121 - 54 121 - 54 121 - 54 122 - 54 121 - 54 122 - 54 123 - 54 124 - 76 125 - 54 127 - 54 128 - 54 129 - 54 1	0.00 0.00 0.00 113.97 0.00 49.23 82.81 78.59 90.468 92.27 89.31 89.31 89.31 89.31 89.31 89.31 89.31 89.31 10.36 89.3
СР	ENHA VA1	NCED INL	ET, STAT VAS	ION B VA4	VA5	VA6	VA7	VAS	VA9	VAIO-AXIAL N
12345678901234567890123456789012345678901234567890	0.00 4.14 0.00 0.00 0.00 0.00 0.00 0.00	274.76 277.59 79.64 127.53 194.49 195.85 139.37 162.85 139.53 104.55 139.53 105.86 48.50 60.61 60.61 60.67 79.55 105.86 60.67 79.55 113.85	59.97 42.518 38.75 35.61 39.61 11.11 209.61 12.14.35 22.4.35 22.4.35 22.4.35 22.4.36 22.5.77 230.71 233.61 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 249.21 259.88 274.00 263.61 260.22 260.32 276.89 271.55 251.24 260.32 276.89 271.35 251.24 260.32 276.89 271.35 251.24 261.25 261.26 262.46 262.262.46 262.27 263.35 263.35 272.35 263.35 272.35 273.59 271.35 273.59 271.35 273.59 271.35 273.59 271.35 273.59 271.35 272.35 273.59 273.59 273.59 273.59 273.59 274.00 263.61 274.00 263.61 274.00 263.61 274.00 273.59 274.00 263.61 274.00 263.61 274.00 263.61 274.00 263.61 274.00 263.61 274.00 263.61 274.00 274.00 274.00 275.00 275.00 275.00 276	47.04 42.55 42.96 42.42.42.42.42.42.43.44 47.108 44.88 44.88 45.89 47.08 46.88 47.08 46.88 47.08	45.80 0.085 4.407 14.002 28.04 12.79 12.79 12.79 12.79 14.02 12.79 14.02 12.79 14.02 12.79 14.02 12.79 13.03 13.03 13.03 13.03 13.03 14.03 15.03 16.03 17.03 17.03 18.03 1	36.17 47.65 35.72 52.56 44.66 106.93 77.25 96.40 1125.35 124.03 124.03 124.03 125.35 127.21 130.15 131.35 132.35 133.35 135.	20.20 33.31 37.64 30.59 2.08 35.29 2.08 36.07 79.14 105.95 101.35 117.49 123.78 125.78 125.16 127.30 131.28 128.22 129.440 130.27 126.50 127.46 128.22 129.460 130.27 125.76 125.97 125.97 125.97 125.97 125.97 125.97 125.97 125.97 125.97 126.97 127.97 127.97 128.22 129.97	54.63 65.58 69.69 80.99 80.99 92.31 105.69 123.59 122.70 127.78 129.22.70 127.78 129.22.70 127.78 128.99 128.36 128.99 128.36 12	41.50 29.60 74.52 47.59 64.78 49.60 84.00 75.47 86.19 93.03 98.75 93.31 101.51 101.03 98.73 102.44 107.77 114.33 106.03 113.41 113.31 106.03 113.41 115.31 107.65 112.09 113.63 114.63 115.63 116.75 117.63 1	26.65 89.65 89.84 1.20 250.76 17.40 87.40 13.05 129.78 87.5.57 31.86 21.30 97.12 29.34 29.34 29.30 21.30 21.30 21.30 22.31 28.34 29.34 29.34 29.34 20.38 20.38 20.66 21.50 20.66 21.50 20.66 20.

TABLE A2. -

(b) ABSOLUTE TANGENTIAL VELOCITY (M/S)

СР	BASE VT1	LINE INL VT2	ET, STAT	ION 1 VT4	VT5	VT6	VT7	VTB	VT9	VT10- ABS TA	N
123456789012345678901234567890123456789012345678901234567890	4.5799 4.5799 4.644.839 5.444.839 5.444.839 5.444.839 5.444.839 5.444.839 6.1053 6.105	3.666 646 3.132 3.134 3.132 3.	4.0873626998989898989899999999999999999999999	4.275939495554986564445535546566666552555666665525556566666552555656666655255565666666	4655554578645655565556555655565556555565	759196426465738999859797978686164469979757859804562687	6.2872444166.3924664166.3924664166.3924664166.3924664166.39246641665.39346661.9934665.3934665.393466555665556655566555665556655566666666	4.367.52144.65.55.55.55.1.2888.299.887.65.55.55.55.55.55.55.65.56.55.55.65.66.55.66.65.65	45.09992217626161616161797979797979797979797979797979	5 7 0 0 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	
CP 1	BASE VT1 -40.00	LINE INL VT2 -30.85	VT3	ION 2 VT4 -3.44	VT5	VT6 38.54	VT7 6.83	VTB 18.60	VT9 25.72	VT10- ABS TA	AH.
12345678901234567890123456789012345678901234567890	-44.937 -38.93.93.93.93.93.93.93.93.93.93.93.93.93.	-39.29 -412.49 -42.49 -42.49 -53.75 -53.75 -62.74 -153.95 -10.57 -274.56 -274.56 -274.56 -274.56 -274.56 -274.56 -274.56 -274.56 -274.56 -274.56 -275.57 -276.56 -	-13.36 -20.69 -31.52 -31.52 -31.52 -31.52 -31.85 -27.73 -26.38 -27.73 -26.38 -14.79 -21.35 -21.47 -20.81 -21.55 -22.55 -23.55 -23.55 -24.55 -25.55	-15.32 -128.49 -312.55 -32.55 -32.55 -32.55 -32.49 -19.57 -32.48 -19.57 -28.89 -28.49 -18.55 -28.49 -19.57 -20.57	-10.508 -120.985 -229.0145 -229.029.145 -229.0	27.31 -7.000 -16.24 -26.27 -29.50 -29.603 -29.603 -29.603 -29.603 -29.77 -25.50 -22.57 -27.79 -27.79 -17.87 -11.38 -21.21 -21.38 -21.21 -21.38 -21.21	-17.3 -24.3 -37.0 -38.7 -40.6 -36.36.6 -36.36.86 -36.36.86 -28.2 -28.86 -14.7 -18.6 -14.7 -10.1 -14.7 -10.1 -10.	-3.61 -3.22 -3.7.25 -3.7.25 -4.0.56 -3.8.02 -3.8.02 -3.8.57 -3.8.66 -3.8.57 -1.7.57 -1.6.99 -1.4.91 -1.5.44 -1.7.57 -1.5.44 -1.7.57 -1.5.44 -1.7.57 -1.5.44 -1.7.57 -1.5.44 -1.7.57 -1.5.44 -1.7.57 -1.5.44 -1	19.84 19.84 19.84 19.29 19	-20.10 -16.18 -16.18 -16.142 -19.844 -17.64 -10.88 -10.88 -10.88 -10.88 -10.88 -10.22 -12.13 -12.13 -22.86 -6.98 -6.98 -14.33 -14.33 -14.60 -10.27 -15.61 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.80 -10.27 -10.82 -1	

CP	BASELINE INU VT1 VT2	ET, STATION 3	V15	VT6	VT7	VTB	VT9	VT10- ABS TAN
12345678901123456789011234567890112345678901234444444667890	MHHHHHH -20.30 0.00 0.00 0.00 -19.96 -6.52 -3.95 -6.57 -3.95 6.65 7.07 14.62 17.68 17.68 17.98 1	14.49 -6.36 14.82 -4.69 14.82 -4.69 14.82 -4.69 16.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 -33.25 -24.92 -19.49 -17.09 -11.46 -22.46 -7.39 -12.31 -11.46 -22.46 -7.39 -12.31 -11.46 -22.46 -7.39 -12.31 -11.46 -22.46 -7.12.31 -11.76 -22.46 -7.12.31 -11.76 -22.46 -7.12 -11.76 -22.46 -7.12 -11.76 -22.46 -7.12 -11.76 -22.46 -7.12 -11.76 -22.46 -7.12	6.23 10.20 112.24 187.61 0.000 0.000 -11.58 -28.61 -20.85 -20.87 -20.87 -2.	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	-20.85 2.11 189.18 8.00 0.00 -31.18 8.46 9.46 -24.93 -18.87 -12.86 4.63 -18.96 -2.36 -18.07 -18.07 -18.07 -19.07 -	0.00 0.00 158.04 158.04 0.00 0.	0.00 11.42 23.59 39.73 184.44 0.00 0.00 37.20 -20.87 -28.01 -28.07 -31.69 -26.60	0.00 1.755 -1.25 -3.59 14.17 -28.43 0.00 0.00 0.00 26.97 -31.79 -24.69 -3.2.69 -106.43 -107.58 -
CP	BASELINE INL	ET, STATION 4 VT3 VT4	VTS	VT6	VT7	VT8	VT9	VT10- ABS TAN
123456789012345678901233456789012334567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	49.59 43.97 45.78 45.78 35.61 35.69 41.46 41.46 41.46 41.46 62.77 65.61 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000	54.90 38.75 34.86 32.61 38.86 32.61 38.86 37.72 40.05 42.97 40.07 40.55 47.72 40.55 51.16 55.57 66.57 77.86 60.50 62.28 65.57 77.86 60.50 62.28 65.57 77.86 60.50 62.28 65.57 66.86 60.50 60.00 60.00 60.00	24.533 24.533 250.914 250.914 250.914 333.451 333.121	41.76 41.70	91.469 91.219 91	111.08 102.46 102.46 103.47 109.79 109.79 109.109 109.

CP	BASEL VT1	.INE INL	ET, STAT VT3	ION 5 VT4	VT5	VT6	VT7	VTS	VT9	VT10-	ABS	TAN
12345678901234567890123456789012345678901234567890	0.00 0.00 97.560 111.09 1106.89 1107.89 98.57 104.00 98.57 104.00 99.27 89.1.33 85.52 84.59 80.22 80.80 80.99 80.22 80.22 80.22 80.00 80.0	0.00 0.02 0.02 0.02 0.04 0.04 0.04 0.05 0.05 0.05 0.05 0.05	0.00 0.00 20.98 69.65 69.65 69.68 69.68 69.86 770.31 772.74 75.09 772.74 775.09 777.75 78.86 777.78 78.86 777.78 78.86 777.78 779.86 77	0.00 0.00 0.00 1.98 73.451 18.458 77.4.58 77.75.48 18.85 77.76.57 78.85 18.82 78.83 78.83 78.83 78.83 78.83 77.75 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.77 78.63 78.	0.00 92.29 86.00 76.86 85.26 72.08 71.46 72.27 73.27 74.32 77.56 81.42 77.56 78.55 78.55 78.75 7	0.00 113.65 108.65 196.68 89.99 84.22 775.58 775.75.76 80.15 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.20 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66 80.35 80.66	0.00 66.43 67.54 67.54 66.53 66.53 66.51 71.83 77.73 77.19 77.73 77.19 77.73 77.74 9	80.70 81.06 75.12 75.12 64.05 67.05	114 .71 911.75 97.79 95.59 97.79 95.59 84.92 95.59 84.92 77.25 77.26 84.92 77.25 66.28 77.69 66.28 67.61 68.63 67.61 68.75 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.94 70.85 70.85 70.94 70.85 70.87 70.	0.00 140.87 159.374 159.374 161.461 197.364 19		
CP			ET, STAT VT3		VT5	VT6	VT7	VT8	VT9	VT10-	ABS	TAN
12345678901234567890123456789012345678901234567890	0.00 137.64 123.65 124.84 127.69 117.94 118.36 118.36 119.69 117.94 115.08 115.08 116.99 117.94 1109.10 101.69	00077883735578884992372700006546460000009943.8894992373600000000000000000000000000000000000	0.00 97.65 96.62 96.62 96.85 96.95 95.16 95.16 95.36 94.99 95.36 94.99 95.36 94.99 95.36 94.99 95.36 94.99 95.36 94.99 95.36 96.86 97.	0.00 105.03 105.91 101.03 98.05 97.23 98.45 97.23 98.45 97.23 98.45 97.24 93.16	0.00 114.73 112.97 104.82 101.44 89 101.44 99.42 99.42 97.05 97.75 96.11 93.30 96.41 97.05	0.00 110.86 110.82 198.82 96.59 95.262 96.59 91.50 88.85.71 88.44 85.46 85.71 84.48 85.46 85.71 84.48 85.46 85.71 81.92 87.83 80.79 81.59 80.79 81.59 80.79 81.59 80.79 81.59	0.09 95.37 95.37 95.76 98.36 85.963 86.95 87.19 82.63 87.19 82.63 87.19 82.63 87.19 82.79 82.83 80.65 80.05 80.07 79.26 80.07 79.26 79.18 80.07 79.26 79.18 80.07 79.26 79.18 80.07 79.26 79.18	0.00 97.642 97.642 87.573 87.573 83.1123 82.666 80.99 779.06 81.666 80.99 779.06 81.666 80.99 776.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 87.76.98 88.99 89.90 89.90 80 80 80 80 80 80 80 80 80 80 80 80 80	0.009 140.37 77.62 69.72 55.22 69.73 57.52 69.73 61.68 77.61 61.68 77.48 61.68 71.99 58.88 64.68 65.89 66.60 60.31 74.35 61.62 62.30 64.08 62.31 64.35 64.08 62.31 64.35 64.08 62.31 64.35 64.35 64.35 64.35 64.35 64.35 64.35	03.464 110.456 1110.251 111.351 193.864 192.459 192.459 192.459 192.459 192.459 192.459 192.459 193.864 194.459 195.858 197.65		

Company Winds

BASELINE CP VT1 V	INLET, STATI	ON 7 VT4	VT5	VT6	¥T7	VT8	VT9	VT10- ABS TA	AN
1 0.00 0 2 0.00 112 3 128.37 112 4 -131.68 109 5 -123.18 108 6 115.85 107 7 -126.99 107 8 -121.27 105 9 -123.94 106 10 115.86 105 11 115.86 105 11 115.86 105 11 115.86 105 12 115.89 105 13 112.24 104 15 113.82 104 15 113.82 104 15 113.82 104 16 112.02 104 17 115.25 104 18 115.18 104 19 108.75 103 22 106.44 103 22 102.45 103 23 101.86 103 24 101.74 102 25 101.78 102 25 101.78 102 25 101.78 102 25 101.78 102 25 101.78 102 27 100.69 103 30 107.35 101 31 -111.01 100 33 102.97 100 33 102.97 100 33 102.97 100 33 102.97 100 33 102.97 100 33 102.97 100 34 106.57 93 35 109.72 93 36 106.57 93 37 108.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.72 93 38 106.57 99 36 109.72 93 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 36 106.57 99 37 109.50 98 37 109.50 98 38 106.57 99	.86 107.42 23 107.13 08 105.78 83 104.05 87 104.76 67 104.76 16 103.11 16 103.52 29 101.58 96 102.11 46 101.45 48 101.92 38 101.92 38 101.92 38 100.51 25 102.59 46 102.11 46 101.92 38 101.92 38 100.52 38 100.51 38 100.51 38 100.51 39 100.52 38 98.80 59 98.80 59 100.52 88 98.80 91 98.88 94.70 95.97 96.98 97.23 15 96.51 16 96.58 47 797.85 96.98 47 97.98 48 106.33 19 115.09 70 00.00 71 105.62	0.00 120.088 12 115.088 12 116.489 11 110.749 11 110.74	37520752201755568867088795220065997549118751136412578201146	0.00 124.4856 1114.4856 1114.4856 1108.3466 11	0.00774226765610.0000774282887465886746821883104312299955526610.00077889855266748880888104312299663910000077886888213667368000000000000000000000000000000000	0.00 103.09 103.09 99.37 102.49 99.37 102.49 99.37 103.37	0.000 109.341 1002.32	0.00 0.00 0.00 125.943 116.544 113.717 107.564 114.564 116.564	
BASELINE CP VT1 V	INLET, STATI	ON B VT4	VT5	VT6	VT7	VT8	VT9	VT10- ABS TA	AN
2 26.28 65 3 17.52 70 4 147.06 -48 5 0.00 63 6 -12.94 129 8 0.00 87 9 0.00 97 11 0.00 86 13 0.00 105 14 0.00 12 15 0.00 97 16 0.00 105 17 0.00 86 18 0.00 105 17 0.00 86 20 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 21 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 21 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 22 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 22 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00 98 21 0.00 98 21 0.00 98 22 0.00 98 23 0.00 98 24 0.00 98 25 0.00 98 26 0.00 98 27 0.00 98 26 0.00 98 27 0.00 98 27 0.00 98 28 0.00 98 29 0.00 98 20 0.00	04 106.29 28 92.97 67 114.43 40 111.30 68 113.36 68 113.37 107.72 110.57 15 110.57 15 110.57 15 110.57 15 107.89 164 104.63 17 107.89 17 107.89 18 107.89 18 107.89 18 107.89 18 108.37 18 107.89 18 108.37 18 108.37 18 108.38 18 108.39 18	130.059 13 119.97 12 1119.881 12 1118.881 12 1118.266 11 112.366 11 112.743 11 110.525 11 1108.950 100 109.30 100 100 100 100 100 100 100 100 100 10	5.99 9.28 95.62 7.62 7.19 7.19 9.30 9.30 9.30 9.30 9.30 9.30 9.30 9.3	144. 88 1 1132 8. 71 1145. 89 1 127. 252 29 1 104. 207 116. 207 117. 252 29 1 104. 207 117. 252 29 1 104. 207 117. 252 29 1 104. 207 117.	114431212111111111111111111111111111111	1556-9-35-2-4-6-11-14-1-3-3-5-2-4-6-3-2-4-6-3-3-5-2-4-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1	1441.45.72.9.3.14.45.14.	164. 17 148. 10 149. 91 142. 97 140. 18 142. 97 140. 18 121. 25 146. 71 121. 63 129. 64 129. 121 121. 63 111. 25 111. 25 111. 25 111. 26 110. 47 110.	

CP	ENHA VT1	NCED INL VT2	ET, STAT VT3	IDN 1 VT4	VT5	VT6	VT7	VT6	VT9	VT10- ABS TAN	
123456789012345678901234567890123456789012345678901234567890	923714697746798991116734694554682545596116774677467746774677467746774677467746	443365987758117460669740992227433853152310911663698564 44536598775238653665366674099286971857388653364718688723914	5.941835397122345373779662247722166753256756455776657766666677665554555445475436459022	4.55.55.55.55.57.59.21.48.88.24.43.29.55.55.55.55.55.55.55.55.55.55.55.55.55	55.34204949313527564.8351314284649981672496.577774496.8529377765.8885.8481812846499816777765566665775666545655655565556655756655665	5.0029 6.00361	4.55.299035.84777727.869.89707.21.10.62.899.835.84777.27.869.89707.21.10.18.28554.57.49.25.29.29.5.45.65.55.65.55.55.55.55.55.4.4.74.55.4.2.9.9.5.55.55.55.55.55.54.55.55.4.4.4.7.4.55.4.2.3.55.4.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.	6.8568 65.8576233313699328337000554.63.69.99.96.75.215188 65.85766.67.55808337000554.63.89.99.97.98.7.22.6 66.65.766.766.55.45.63.89.99.97.22.50.26.65.56.65.56.65.56.65.56.65.56.65.56.65.56.45.44.44.45.45.45.45.45.44.44.45.45.45.	44.551.69 44.1502583.189 44.1502583 44.1502583 44.1502583 44.150258 44.	7.49 5.67 5.67 5.67 6.77 7.39 6.07 6.07 6.07 6.07 6.07 6.07 6.07 6.07	
CP		NCED INL VT2	ET, STAT VT3		VTS	VT6	VT7	VTB	VT9	VT10- ABS TAN	
12345678901234567890123456789012345678901234567890	-11.35 -29.48 -29.98 -29.98 -28.46 -20.57 -13.76 -20.98 -13.76 -21.3.76 -21	6.307 -18.307 -7.80.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.20.053 -7.47 -7.4	24.760 0.059 -11.811 -20.28 -20.18 -20.18 -21.72 -20.48 -21.18 -2	32.199 -3.209 -3	42.320 6.709 -24.286 -32.889 -34.417 -35.6467 -35.739 -32.8817 -26.297 -113.891 -14.309 -6.794 -10.349 -6.794 -10.349	41.05 9.37 -9.81 -20.14 -34.35 -34.35 -34.41 -52.45 -27.24 -11.20 -14.36 -11.20 -14.36 -11.20 -14.36 -11.20 -14.36 -11.20 -14.36 -11.20 -14.36 -11.20 -14.36 -11.20 -27.77 -10.22 -27.77 -10.22 -27.77 -10.22 -27.77 -2	38.620 11.78 4.78 12.74 15.76 10.77 10.78	43.71 45.05 28.66 -19.32 -27.39.36 -40.65 -39.86 -39.87 -30.81 -33.81 -29.02 -11.90 -15.15 -17.90 -11.90 -1	44.19 50.37 2-0.167 2-0.167 2-24.553 2-24.653 3-329.28 3-329.29 3-38.361 3-329.21 3-38.361 3-329.21 3-38.361 3-329.21 3-38.361 3-329.21 3-38.361 3-329.22 3-38.361 3-329.22 3-38.361 3-329.22 3-38.361 3-329.22 3-38.361 3-329.22 3-38.361 3-329.22 3-38.361 3-329.22 3-329.361 3-329.361 3-329.363 3-329.363 3-329.363 3-329.363 3-329.363 3-329.363 3-329.363 3-329.363	36.01 44.17 44.17 44.17 44.17 46.30 46.30 47.41	

СР	ENHA VT1	NCED INL VT2	ET, STAT VT3	ION 3 VT4	VT5	VT6	V T7	VTB	VT9	VT10- ABS TAN
123456789012345678901234567890123456789012344567890	0.000 0.000 0.000 0.000 0.000 -8.53 -7.192 -0.368 8.304 -7.39 -14.41 -7.39 -7.	0.00 0.00 189.40 0.00 0.00 0.00 0.00 0.00 0.00 -13.69 -17.56 -15.24 -21.69 -17.56 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17 -21.88 -4.17	0.00 0.00 0.00	0.00 0.00 0.00 197.87 0.00 0.00 10.05	0.00 0.00 0.00 42.70 0.00	0.00 0.00 -6.95 -218.20 0.00 0.	0.00 0.00 32.94 -231.64 0.00 0.00 -20.86 -28.86 -28.23 -18.45 -13.94 -23.23 -13.45 -13.96 -24.23 -25.24 -27.01 -2.88 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -1.34 -2.32 -2.32 -2.32 -2.34 -3.32 -3.34 -3.32 -3.34 -3.32 -3.34 -3.32 -3.34 -3.32 -3.34 -3.33 -3.33 -3.33 -3.33 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.35 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.34 -3.35 -3.35 -3.36 -3.	0.00 0.00 143.21 -143.21 -0.00 0.00 0.00 4.24.25 -23.71 -16.23 -16.2	0.00 0.00 0.00 0.00 123.25 0.00 0.00 164.55 75.65 -15.96 -17.00 -17	75. 26 12. 78 0.00 0.00 -7. 61 -13. 71 0.00 15. 83 -55. 53 -64. 51 51. 36 55. 52 -7. 64 -1. 27 73. 50 30. 55 -7. 69 -46 -15. 34 -1. 27 77. 93 -1. 27 77. 93 -1. 27 77. 93 -1. 27 -7. 1. 80 -1. 67 -7. 1. 80 -1. 67 -7. 1. 80 -1. 67 -7. 1. 80 -1. 67 -7. 1. 80 -1. 67 -7. 1. 80 -7.
CP	ENHA VT1	NCED INL	ET, STAT VT3	ION 4 VT4	V15	VT6	VT7	VTB	VT9	VT10- ABS TAR
12345678901234567890123456789012345678901234567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 0.00 -5.55 -31.55 -31.73 -35.00 -35.28 -37.55 -37.55 -37.55 -38.60 -37.55 -38.60 -38.6	0.00 0.00 0.00 70.37 53.00 37.28 37.28 443.24 443.24 443.24 443.25 57.15 449.05 57.15 62.14 65.62 62.55 77.55 62.14 65.62 68.47 77.50 80.00	0.00 80.94 120.43 120.43 120.43 120.43 120.43 120.28 41.43 12.28 41.43 13.39.52 42.80 46.81 49.05 55.39 56.46 67.86 67.86 67.86 70.92 66.66 0.00 0.00 0.00 0.00 0.00 0.00 0.	0.000 0.005 28.311 39.35 3	47.28 48.28 49.27 49.27 49.27 49.28 49	35.93 28.12 30.05 29.720 30.07 3	48. 485 34. 85 32. 16 35. 49 32. 17 34. 532 35. 98 41. 17 34. 85 37. 18 48. 32 48. 32 49.	162.59 199.35 995.99 61.35 95.99 61.35 95.99 61.93 62.50 63.85 64.35 64.37 64.37 64.37 64.37 64.37 64.37 64.37 65.47 67.77 68.97	0.00 88.20 0.00 64.38 16.47 41.85 328.19 250.75 39.747 300.64 337 45.23 400.75 400.77 41.05 57.77 468.77 40.65 57.77 68.67 59.67 59.69 69.67 59.69 69.69

CP	ENHA VT1	NCED INL VT2	ET, STAT	10N 5 VT4	VT5	VT6	V17	VTB	VT9	VT10- ARS TAN
123456789011234567890123456789012345678901234567890	0.00 0.00 0.00 32.05 126.82 7.92.94 123.52 101.08 50.85 69.30 82.92 777.34 63.45 69.30 82.92 777.34 63.45 69.30 82.92 777.50.46 67.30 82.92 82 82 82 82 82 82 82 82 82 82 82 82 82	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 49.85 56.45 56.45 56.45 76.68 77.55 68.59 77.56 68.59 77.56 81.55 79.21 81.55 79.21 80.00 0.0	0.00 0.00 95.364 77.3.55 74.39 77.1.35	0.00 107.18 86.262 79.203 86.262 79.21 78.004 77.47 74.48 87.5.25 77.44 75.02 77.44 75.02 77.44 75.02 77.44 75.02 77.44 76.02 77.44 77.50 77.	0.00 106.82 99.39 99.31 83.15 67.74.31 774.52 71.85 775.50 77.89 772.89 774.82 774.82 774.82 774.82 774.82 774.82 774.82 774.82 774.82 774.82 775.91 775.92	0.00 72.49 68.89 68.89 65.56 64.55.60 64.65.60 64.90 66.08 66.79 70.11 70.87 71.16 77.12 87 77.12 87 77.13 69.49 69.49 69.69 6	74.70 69.03 62.92 65.15 62.95 62.86 62.86 63.75 64.31 66.37 67.86 67.86 67.87 67.97 68.78 69.88 69.14 69.88 69.14 69.88 69.14 69.88 69.14 69.88 69.14 69.88	79.90 74.87 71.70 67.66 66.66 66.67 67.67 64.63 66.24 64.61 64.16 66.38 66.33 67.39 68.67 69.68 67.10 68.10 68.10 69.66	106.00 113.754 91.68 86.168 88.06 87.305 74.851 77.701.69 68.27 70.350 69.27 68.27 67.74 68.64 66.25 68.51	\$1.62 \$9,505 \$44.755 \$44.755 \$44.755 \$45.756 \$45.757 \$45.75
СР	ENHA VT1	NCED INL VT2	ET, STAT	ION 6 VT4	VTS	VT6	VT 7	VT8	VT9	VT10- ABS TAN
12345678901234567890123456789012345678901234567890	0.00 25.37 104.752 114.89 117.172 108.09 61.455 114.557 61.469 14.57 14.94 77.95 105.57 14.94 97.70 105.57 14.94 105.57 1	0.00 0.00 0.00 1.29 87.09 88.09 89.30 88.33 88.33 88.33 88.33 88.33 88.33 88.33 88.33 88.33 89.33 89.33 89.37 89.33	0.00 0.00 110.21 95.485 105.85	0.00 171.22 1124.80 11	0.00 1106.50 1106.50 195.27 194.04 87.10 87.10 87.10 87.79 87.79 87.85 78.53 78.53 77.28 77.79 77.28 7	0.00 95.96 91.07 92.12 91.07 82.63 77.8.97 76.16 77.23 76.21 77.24 78.58 76.26 77.67 77.67 77.67 77.67 77.67 77.67 77.62 77.62 77.58 77.62	0.00 94.69 82.584 79.865 77.605 77.89 772.09 772.13 771.866 772.73 771.80 771.80 771.80 771.80 771.80 771.80 771.80 771.80 771.80 771.80	0.00 89.39 85.724 85.733 78.045 775.08 773.852 771.90 773.87 771.90 770.91 770.	0.00 111.29 111.	

CP	ENHANCED VT1 VI	INLET, STA	TIDN 7 VT4	VT5	VT6	VT7	VT8	VT9	VT10-	ABS	TAN	
123456789101112345678910111213456789101112134567899101112134567910111211111111111111111111111111111111	0.00	00 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	132.52 -9.892 -19.3861 103.331 101.3331 1	0.00 119.22 131.739 114.39 115.83 112.39 115.83 112.39 26.46 98.46 89.38 87.88 87.88 87.88 88.49 84.42 88.31 83.46 83.31 83.26 83.31 83.27 84.22 85.15 85.16 85.16 85.17 85.27 86.00 0.00	0.00 119.67 1115.553 116.553 116.853 91.27 84.264 87.27 88.29 1.27 88.29 1.27 88.29 1.27 88.29 1.27 88.29 1.27 88.29 1.27 77.52 77.52 78.29 77.52 77.52 77.77 77.7	0.00 119.29 100.63 104.67 99.05 56.43 85.11 75.59 77.19 76.81 75.59 75.06 75.07 74.24 74.19 74.29 74.29 74.24 74.19 73.60 73.60 73.71 73.71 73.71 73.71 73.71 73.71 73.71 73.71 73.71 73.71 74.64 75.93 75.9	0.00 0.00 10.58 99.64 98.25 94.35 88.76 88.76 88.76 77.78 77.78 77.78 77.28 77.39 77	0.00 0.01 119.50 110.50 110.60 1101.62 99.19 69.63 88.07 89.29 89.29 80.48 76.41 77.20 80.00 80.	0.00 0.00			
CP	ENHANCED VT1 VT	INLET, STA	TIOH 8 VT4	VTS	VT6	VT7	VT8	VT9	VT10-	ABS	HAT	
1 2 3 4 5 6 7 8 8 9 10 11 12 3 14 5 6 7 7 8 8 9 10 11 12 3 14 5 16 7 18 8 12 0 12 22 24 5 26 7 28 8 3 3 3 3 5 5 3 7 8 6 12 2 3 3 4 6 5 6 6 7 6 6 8 8 9 5 0	0.00 43. 0.00 15. 0.00 22. 0.00 26. 0.00 27. 0.00 28. 0.00 28. 0.00 29. 0.00 20. 0.00 41. 0.00 41. 0.00 42. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 34. 0.00 35. 0.00 35. 0.00 36. 0.00 37. 0.0	12 33.585 96 27.3167 714 36.367 728.1774 36.367 912 0.000 863 113.59 11	0.01 -96.13 -90.88 -91.63 -89.63 -87.98 -87.98 -87.98 -85.92 -85.92 -81.97 -81.97 -81.97 -81.97 -81.14 -81.97 -81.14 -81.	0.00 0.00 0.00 0.00 0.00 -74.71 0.00	75.05 80.77 103.29 119.82	-131.58 -128.68 -128.68 -123.65 -123.65 -113.28 -113.28 -113.28 -113.28 -123.65 -123.6	127.157 179.281 1111.85 1104.85 1104.87 1105.86 1105.8	137-14 137-56 137-56 137-56 137-56 137-56 137-76 138-37-154 137-76 138-37-154 1101-38-38 1102-38-38-38-38-38-38-38-38-38-38-38-38-38-	4483244610862874808748698888888888888888888888888888888888			

TABLE A3.
(C) NUMBER OF AXIAL MEASUREMENTS

СР	BASELIN MA1	E INLET,	STATION MA3	1 MA4	MA5	HA6	MA7	MAB	MA9	MA10-AXIAL HEAS
1234567890123456789012345678901234567890	99 119 101 101 101 101 101 102 103 112 112 113 114 115 116 117 117 117 117 117 117 117 117 117	249 245 245 303 303 284 425 257 287 287 287 287 287 287 287 287 287 28	1760 1760 1874 1185 1185 1185 1185 1186 1187 1188 1189 1188 1189 1188 1189 1188 1	1438 4458 458 463 463 463 463 463 463 463 463 463 463	200 1932 1975 1990 1200 1200 1200 1200 1200 1200 1200	315 316 318 358 358 353 353 353 353 353 353 353 35	217 207 229 229 2213 2213 2214 2213 2214 2216 2216 2217 2217 2218	219 194 207 215 2215 2216 2216 2216 2217 2217 2217 2217 2217	145 1118 128 128 133 1134 1135 1136 1137 1138 1139 1139 1139 1139 1139 1139 1139	13 77 15 8 6 9 6 77 12 9 11 18 12 8 8 11 10 12 6 9 7 10 12 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6
CP	BASELIN MA1	E INLET,	STATION MA3	2 MA4	MA5	MA6	MA7	MA8	MA9	MA10-AXIAL MEAS
12345678901234567890123456789012345678901234567890	200 210 221 228 228 228 228 228 228 228 228 228	3444586914329964477660803268871770740139577032102475	11947759550583110707778700711159765204624211094774264358	113630921034181782678299982619513600209791762585229	146 146 147 151 160 120 120 121 108 116 108 117 108 1	174 157 1008 1138 1106 1355 1157 147 1557 147 158 1157 1158 1160 1153 1160 1153 1160 1153 1160 1153 1160 1160 1160 1160 1160 1160 1160 116	91 78 86 106 99 96 103 87 88 96 103 87 98 98 98 98 98 98 99 98 100 100 100 100 100 100 100 10	104 702 1005 892 1146 951 1018 777 108 977 108 1018 1018 1018 1018 1018 1018 101	1702 990 1019 999 1019 999 1019 999 1019 999 1019 999 1019 999 1019 853 107 944 858 107 999 108 108 999 108 108 108 108 108 108 108 108 108 108	43 24 11 9 8 10 12 11 6 10 12 16 10 7 7 8 6 8 8 11 12 11 6 11 17 7 16 11 9 10 11 9 11 11 10 11 10 11 11 10 11 11 10 11 11

CP	BASELIN MA1	E INLET,	STATION MA3	3 MA4	MA5	MA6	HA7	MAB	MA9	MA10-AXIAL MEAS
12345678901234567890123456789012345678901234567890		924000010575324283373295125967687008258292109200015	54 100000114053602627140876222338867778055339146923202	1 14400001518911634115044984492265113810239103561000	301500012253511194844452222121111112112215121111111111111	0 0 3 1 6 8 4 0 3 8 1 1 0 0 0 3 1 6 8 4 0 3 8 1 1 0 8 9 9 7 8 6 2 8 9 5 1 1 0 1 1 1 1 2 1 2 1 1 1 1 2 1 1 1 1 2 1 1 1 1 1 1 1 2 1	90 225 00 1324 496 566 577 675 1765 1871 1871 1871 1871 1871 1871 1871 187	0 0 7 7 3 0 0 0 31288335533448256557411582865777791989342316667556556574115829999611910101666	123100000352649774394563575725399597210475773645560011233111311312222122222222222333322	06041200002322632247283722864617432755544525643300
СР	BASELIN Mal	E INLET, MA2	STATION MA3	4 MA4	MA5	MA6	MA7	BAM	MA9	MA10-AXIAL MEAS
123456789012345678901234567890123456789012345678901234567890	0004604598079011183420531836264424511000000000000000000000000000000000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 9 5 8 0 1 0 5 5 9 5 6 7 9 3 9 2 2 2 1 1 1 5 5 5 6 6 7 9 3 9 2 2 2 1 1 1 5 7 8 9 1 2 1 1 5 7 8 9 1 2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	07 160 177 1745 180 1114 1115 1115 1115 1115 1115 1115 111	1205973239856659727171255010100000242114000000000000000000000	109 275 4567 1375 1257 1257 1257 1257 1257 1257 1257 12	72 156 255 353 353 456 459 457 468 467 467 467 467 467 467 467 467 467 467	26 44 45 43 119 66 77 70 81 107 107 107 107 107 107 107 107 107 10	33365209269437236670996403149046101817138487690893111135531490461018171384831000000000000000000000000000000000	397911864728155019667843744916655025941300058000000000000000000000000000000000

CP	BASELIN MA1	E INLET, MAZ	STATION MA3	MA4	MA5	MA6	HA7	MAS	MA9	MA10-AXIAL MEAS
12345678901234567890123456789012345678901234567890	0 0 0 106 1158 1137 1137 1137 1137 1137 1137 121 106 110 98 85 84 85 85 86 87 84 44 41 37 8 29 26 11 7 3 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 0 0 98 109 118 105 115 114 133 114 135 114 125 117 133 116 117 133 117 133 117 133 117 133 117 133 117 133 117 133 133	00118095282094751736497084710001000000000000000000000000000000	0 0 17 133 444 4 53 8 84 8 8 6 8 9 5 5 1 10 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	0 215 219 229 35 55 68 99 102 1114 1113 1129 1114 1129 1113 1129 1114 1129 1117 1129 1119 1119 1119 1119 1119	0 277 111 277 1127 488 450 1553 484 487 756 488 983 1111 988 1111 1131 1129 1129 1129 1129 1129 1129	85 447 758 764 666 679 859 857 774 859 857 859 857 859 857 859 857 859 857 859 859 859 859 859 859 859 859 859 859	32 34 40 35 48 46 46 47 77 11 48 48 48 48 48 48 48 48 48 48 48 48 48	198833571754919880781632614801414325586474570000000	6 8 5 12 2 11 1 3 9 11 1 1 4 6 8 1 1 1 1 1 1 6 8 1 2 7 2 2 1 1 2 2 1 1 2 2 2 1 1 2 2 2 1 1 3 3 7 3 6 1 2 5 5 2 2 1 2 2 2 2 2 8 1 3 3 7 3 6 1 2 5 5 2 2 1 2 2 2 2 2 8 1 6 4 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
CP	BASELIN MA1	E INLET, MA2	STATION MA3	6 MA4	MA5	MA6	MA7	MAB	MA9	MAIO-AXIAL MEAS
1234567890123456789012234567890123456789012345678901234567890	0 61 57 66 65 67 86 61 27 86 10 11 13 10 11 13 10 11 10 11 10 10 10 10 10 10 10 10 10	0	00 7440 998 9933339 89933339 81010 11011 11111 8829555144695 81120014660000000000000000000000000000000	07 47 47 47 47 47 47 47 47 47 47 47 47 47	0 15 29 36 36 36 36 36 36 36 36 36 36 36 36 36	00 51 646 657 73 846 573 884 99 99 112 100 100 115 889 762 588 317 15 889 762 588 317 15 889 762 588 317 15 889 762 588 317 15 889 889 889 889 889 889 889 889 889 88	09 81 95 94 1104 1126 1127 1129 1121 1129 1121 1123 1135 1135 1135 1146 1159 1166 1178 1179 1164 1179 1164 1179 1164 1179 1164 1179 1164 1179 1164 1179 1179 1180 1180 1180 1180 1180 1180 1180 118	0 81 82 106 106 1103 1111 1121 1121 1123 1121 1123 1124 1125 1124 1125 1126 1127 1127 1128 1129 1129 1120 1121 1121 1121 1122 1123 1124 1127 1127 1127 1127 1127 1127 1127	100 119 1224661122114668423311221111211274894723113500000	00 37 37 37 31 12 34 34 34 34 34 34 34 34 34 34

and the state of the section of the

CP	BASELI MA1	NE INLET, MA2	STATIC MAS	N 7 MA4	MA5	MA6	MA7	BAH	HA9	MA10-AXIAL MEAS
123456789012345678901234567890123444444444444444444444444444444444444	00911533244823864642294726094858443693896399641600 1111 324977600 123217	0 23472394439443944394439443944394439443944394	00806629873865761215014811036796000057401835010200	00692848258644129105614251539861308538567107300	0 43 57 59 84 85 103 128 113 127 127 127 127 127 127 127 127 127 127	0 24 44 51 52 80 77 96 109 110 112 128 114 145 129 159 159 159 159 169 179 189 189 189 189 189 189 189 18	0 47 64 67 64 79 99 83 104 1104	0 52 45 78 74 75 80 101 92 125 116 125 116 127 128 129 129 125 116 127 128 129 129 129 129 129 129 129 129	0 52 49 52 60 80 81 1124 1135 1131 1124 1135 1138 1138 1138 1138 1138 1138 1138	0002491051439794599873257436206540333318466635876789875454466635876789876454666358767898767635876778987666358767
CP	BASELI MAl	NE INLET, MA2	STATIO MA3	N 8 MA4	MA5	MA6	MA7	MA8	MA9	MAIO-AXIAL MEAS
123456789012345678901234567890123456789012345678901234567890	276310111021000000000000000000001458619855645842	16796676015714611415776227080206320622331442883210658193306563	117988102135555687777988710747535950647777988109437437439651556477776949	13 16 15 37 53 61 66 106 110 110 110 125 1125 1125 1126 113 126 127 128 129 129 129 129 129 129 129 129 129 129	71381922534665115519572572588848487971779861105710571057105710571057105710571057105	103 233 169 272 322 327 358 450 770 627 677 133 986 776 982 982 982 982 984 984 985 981 985 981 1171 1148 1148 1148 1148 1148 1148 11	27 41 34 66 60 68 62 61 75 70 78 106 128 88 173 93 106 112 113 121 112 122 112 123 124 112 124 125 127 128 129 129 129 129 129 129 129 129 129 129	29 34 42 53 44 41 42 68 79 91 92 79 166 106 101 119 1115 102 1121 141 1121 141 152 142 148 141 155 135 146 66 31	41 52 527 677 60 55 64 101 95 66 63 79 93 101 92 84 98 107 107 108 108 109 109 114 1137 125 116 117 125 126 127 127 127 127 127 127 127 127 127 127	128 134 1107 1131 1168 907 1034 890 1003 890 1003 890 1003 890 1003 890 1003 1003 1003 1003 1003 1003 1003 10

CP	ENHANC MA1	ED INLET, MA2	STATIC MA3	ON 1 MAG	MA5	MA6	MA7	BAM	MA9	MA10-AXI	AL MEAS
12345678901234567890123456789012345678901234567890	13396543919149044564663599963342603526822649108459	5446537253010293961916932569939772640655436778190541465555555555555555555555555555555555	04004720956895463500854603253276358998968905149307	56428108347699888577726478818694520948884246637565200	953296654687777958661315855554761236226609352768805475555555555555555555555555555555555	65775553661548020455504377979312021106463776537665372665	56777623861995539112756366671655160829955251889244382304	017555120843367331688735348885747287412396675596576576576576576576576576576576576576576	5651 5676 6669 56769 6669 56769 6765 6776 6775 6775	3322334234323233322233233322233332223323199433	
CP	ENHANC MA1	ED INLET, MA2	STATIO MA3	N 2 MA4	MAS	MA6	MA7	MA8	MA9	MA10-AXI	AL MEAS
12345678901234567890123456789012345678901234567890	4344555454373932452406349052032558121779033403354524565857657077677867768999981918	646656756454455575544555555555555775775778878899901114	185577676685647656551252794953677666688947078332 11855776766856476565566664555564527666688947078332	9856773558590207110587840268146214631200424769461199	103717489 6575996662559965625599656555956657 77589866625599656559965657 77589849	68654445555445545658987757575478156937379868671	11069247620117745101436373819088550666785787803	84 10942085574 118442085575455696470199339918841857042277952008817831	1046246268857866466001566647456149190854173396988050007759799	222 1962236202818831308549191173181128384208324546	

OF STARS MINISTY

CP	ENHANCI MA1	MAZ	STATIC MA3	MA4	MA5	MA6	HA7	MAS	MA9	MAID-AXIAL HEAS
123456789012345678901234567890123456789012345678901234567890	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 2 0 0 0 0 0 0 1 2 2 6 3 8 11 3 11 1 1 2 4 7 7 1 1 0 9 6 8 4 3 7 8 1 1 1 1 1 2 5 1 0 8 2 1 1 1 1 1 2 5 1 8 2 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 6 5 5 5 0 6 7 2 7 7 0 7 16 2 0 0 0 2 6 1 9 2 9 2 9 3 3 3 3 3 3 3 3 3 3 3 3 3 3 3	0 0 1 7 0 0 0 0 0 1 1 1 3 3 9 6 9 7 8 7 8 4 3 1 7 1 9 2 9 5 9 8 8 7 5 7 3 1 5 1 5 3 8 2 4 4 0 3 4 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	00410001479963593840517112499959791256334061555555564999597778899999993	0012000344069546530940819186473036358602888130920671	0 0 1 35 0 0 7 6 45 7 6 107 111 112 113 107 111 113 113 113 114 114 117 119 119 119 119 119 119 119 119 119	0 0 0 0 2 1 1 3 2 8 3 3 5 1 5 3 2 2 3 3 5 2 1 5 3 2 2 3 3 5 2 1 5 3 2 2 3 3 5 2 3 5 2 3 5 2 3 5 2 4 4 5 5 7 2 4 4 5 5 6 5 6 5 6 5 6 5 6 5 6 5 6 5 6 5	0100700081647538920041209294144824753180572979321 2 1124244546644456545957777668769887918113	831007700033566631487454535419027354235861491887987
CP	ENHANC MA1	ED INLET, MA2	STATI	DN 4 MA4	MA5	MA6	MA7	MAB	MA9	MA10-AXIAL MEAS
1234567890123456789012345678901233533333333333444444444444444444444444	0000010204215547425545324132103021130258000000000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	00311120704943282730293795929447983040000071000000000000000000000000000000	11023511466114732129955114663332855599982000000000000000000000000000000	010393113758376843129167885932974244200000000000000000000000000000000	19 20 38 48 514 77 100 100 100 100 100 100 100 100 100	17 345 560 518 663 668 752 789 796 771 1112 1118 1118 1117 1100 1117 1117 1117 1117	7 36 35 35 35 46 46 46 46 46 46 46 46 46 46 46 46 46	15077683716699072420710054738091333694432465305480000000000000000000000000000000000	ORIGINA:

MAS

MA9 MA10-AXIAL MEAS

CP MA1 MA2 MA3 MA4

ENHANCED INLET, STATION 6 MAI MAZ MAS MAS MAS MAS MAS MAS MAS	67 88 328 329 321 125 167 18 68 120 111 123 124 125 127 128 129 129 129 129 129 129 129 129 129 129	77 188 199 114 165 117 177 186 187 199 195 195 195 195 195 195 195 195 195	21 21 21 21 21 21 21 21 21 21 21 21 21 2	12723007558269155272428009614589134486838777667686986177468868387770641988333666	6818968110687846620963690000000000000000000000000000000	233455505118801838243242943232255576000000000000000000000000000000000	424261872770284436878745932308051468000000000000000000000000000000000000	48 493 541 541 558 641 658 658 658 658 658 658 658 658 658 658	10 11 8 16 20 18 19 20 18 19 20 15 20 15 20 15 21 16 10 11 12 13 9 16 10 10 10 10 10 10 10 10 10 10 10 10 10	
27 17 83 179 154 225 164 262 248 116 29 9 37 79 81 113 95 167 181 194 30 7 42 84 76 116 97 181 181 75 31 2 35 79 55 98 100 198 166 75 32 5 42 65 43 83 81 1206 180 70 33 1 23 48 38 65 69 177 200 86 34 2 26 38 38 24 68 76 169 179 88 35 2 22 27 13 46 75 185 174 84 36 2 11 30 17 62 66 155 167 184 194 184	1 0 2 0 3 24 4 15 5 18	0 0 0 0 4 0 5 10 8 24 9 25	0 0 15 24 24 30 30	0 0 8 20 1 20 3 31 2 42	0 41 31 40 37	0 68 86 110 102	0 118 85 118 110	0 60 51 64 49		

ONSUMED PARK IS ON POOR CUALITY

MA7 MA8 MA9 MA10-AXIAL MEAS

ENHANCED INLET, STATION 7
CP MA1 MA2 MA3 MA4 MA5 MA6

47 9 48 2 49 50	0 1 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	14 25 0 0	10 115 9 15 8 8 10 10 10 11 11 11 11 11 11 11 11 11 11	11 22777557155530490302421173267869663380100200	162 112 123 107 115 244 197 216 224 197 226 231 468 336 502 247 410 252 441 452 244 410 252 441 452 263 263 263 274 464 464 464 465 274 465 274 465 274 465 274 465 275 275 275 275 275 275 275 275 275 27	641620416870881089949899999999999999999999999999999	94 95 91 107 107 107 110 110 1110 1117 110 1117 110 1117 110 1117 110 1117 1	5171244625115593856688846025596881451176660	65574706845796739705268771128988147463570	6345477787780491869764556810809013508210
1 2 3 4 5 6 7 8 9 10 1 12 13 4 5 6 7 8 9 10 1 12 13 4 5 6 7 8 9 10 1 12 22 23 4 22 5 22 7 8 29 3 3 1 3 2 3 3 3 5 5 3 6 6 6 1	MA 1 0 1 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	A 141332343545223374457960556593270280004149	STATION MA3 3-00 5-65 5-03 3-69 2-05 1-50 2-7 2-1 2-7 2-7 2-8 2-9 2-7 2-8 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-1 2-9 2-7 2-5 2-8 3-0 2-7 2-5 2-8 3-0 2-7 2-7 2-5 2-8 3-0 2-7 2-7 2-7 2-7 2-7 2-7 2-7 2	MA 446240530735101122332102223122231211732223231	MA 4036668778979584185145315275827485735333333333333333333333333333333333	MA 1 9658696164932247319673405917686020215362858	MA7 1386 1028 848 1028 848 586 638 148 586 638 1886 638 1886 649 649 649 649 649 649 649 649 649 64	MA8 178 177 181 177 181 183 187 184 185 187 144 124 124 124 124 124 125 140 137 147 115 167 181 177 163 181 177 163 181 177 175 177 175 177 175	MA9 178488710251086932604029010035522619468330922642	MAIO-AXIAL MEAS 7 6 4 6 11 22 4 6 6 7 7 7 6 8 13 6 16 6 9 8 10 13 11 12 14 14 14 13 8 10 8 5 17 15 15 15 15 15 15

TABLE A4.
(d) NUMBER OF TANGENTIAL MEASUREMENTS

СР	BASEL MT1	INE INLE	T, STATI	ON 1 MT4	MTS	MT6	MT7	MTS	MT9	MT10-TANG MEAS
1	75	259 253	203 159	279 280	225 226	321 343	233 247	197 196	232 226	40 46
2 3 4	61 67 54	259 244	164 170	293 307	196 212	316 319	232 245	226 215	236 231	41 46
	66 62	268 251	178 170	293 276	202 211	336 326	218 227	203 195	243 242	42 46
7	53 48	255 255	184 184	297 304	212 204	319 320	227 224	227 208	223 234	44 51
9 10	58 72	251 258	181 198	317 291	208 202	324 339	219 241	205 213	227 261	45 42
11 12	56 59	242 268	188 171	286 288	226 212	336 326	207 234	212 210	269 259	50 47
13 14	61 77	264 248 239	187 177	292 2 77	205 211	337 294 362	263 228	245 230 223	250 241	37 40
15 16	78 60	256	177 206	279 317	241 213	328	217 228	205	244 256	55 42
17 18	58 81	265 250	192 185	284 274	212 202	323 319	223 213	214 193	277 232	49 48
19 20	61 61	252 264	160 194	292 318	229 221 237	309 322 337	248 183 217	218 217 224	238 236 278	34 41 42
21 22	58 64	248 264 247	176 175 181	273 275	204 219	341 314	246 224	206 191	258 244	60 56
23 24 25	59 63 58	244 259	173 181	319 282	207 196	334 335	225 236 222	218 233	247 241	46 57
26 27	66	252 252	181 181	321 320 303	231 213	322 325	222 219	217 223	265 247	39 48
28 29	64	260	183 183	302 305	213	356 286	210 215	212 214	259 263	41 44
30 31	47 59	266 279 258	197 179	298 312	217 223 191	311 325	22B 239	192 248	256 265	47 43
32 33	64 67	249 228	180 180	320 269	220 223	319 335	199 220	205 211	236 219	52 49
34 35	50 51	249 266	180 196	318 287	225 198	332 312	230 233 212	218 234	231 231	62 53
36 37	59 65	248 272	188 185	289 286	218 237	344 318	221	211 198	226 231	43 43
38 39	72 61	229 263	197 177	285 301	209 221	332 332	232 230	217 198	248 261	43 46
40 41	56 59	241 248	168 165	283 278	198 246	309 321	217 228	201 189	234 241	40 38
42 43	52 66	263 268	180 189	291 309	201 231	315 333	220 220	214 214	242 257	46 36
44 45	68 52	270 267	177 171 208	291 278 305	212 194	358 333 323	236 226 236	207 221 200	229 259 219	41 47 39
46 47 48	77 77 64	250 255 257	187 191	262 278	235 231 198	330 349	251 210	219 202	239 224	51 40
		259	175	256	216	332	221	215	265	49
49 50	61					537	329	352	490	77
49 50	61 98	436	285	550	337	537	329	352	490	71
50 CP	61 98 BASEL MT1	436 INE INLET MT2	205 T, STATI MT3	550 ON 2 MT4	337 MT5	537 MT6	HT7	MTB	MT9	MT10-TANG MEAS
50 CP	61 98 BASEL MT1 246 263	436 INE INLET MT2 70 58	285 T, STATI: MT3 127 140	550 ON 2 MT4 14	337 MT5 178 170	537 MT6 131 131	HT7 81 81	MTB 75 84	MT9 141 157	MT10-TANG MEAS 99 89
50 CP 1 2 3	61 98 BASEL MT1 246 263 264 293	436 INE INLET MT2 70 58 76 61	205 T, STATI- MT3 127 140 135 132	550 ON 2 MT4 15 15 15	337 MT5 178 170 209 192	537 MT6 131 131 117 136	MT7 81 81 88 86	MTB 75 84 82 80	MT9 141 157 105 88	MT10-TANG HEAS 99 89 25 11
50 CP 1 2 3 4 5	61 98 BASEL MT1 246 263 264 293 284 331	436 INE INLET MT2 70 58 76 61 63 62	285 T, STATI: MT3 127 140 135 132 106 127	550 ON 2 MT4 15 15 12 14 11	337 MT5 178 170 209 192 200 227	537 MT6 131 131 117 136 162 187	MT7 81 81 88 86 91	MTB 75 84 82 80 97 47	MT9 141 157 105 88 90 74	MT10-TANG MEAS 99 89 25 11 14
50 CP 1 23 45 67 8	61 98 BASEL MT1 246 263 264 293 284 331 353 302	436 INE INLET MT2 70 58 76 61 63 62 68 89	285 T, STATI: MT3 127 140 135 132 106 127 129 127	550 ON 2 MT4 15 15 12 14 11 14	337 MT5 178 170 209 192 200 227 199 224	537 MT6 131 131 117 136 142 187 159	MT7 81 88 88 91 104 98	HTB 75 842 80 97 47 77 87	MT9 141 157 105 88 90 74 74	MT10-TANG MEAS 99 89 25 11 14 7 13
50 CP 12345678910	61 98 BASEL MT1 2463 2643 284 3313 3502 378 3671	436 INE INLET MT2 70 58 76 61 63 62 68	285 T, STATI MT3 127 140 135 132 106 127 129	550 ON 2 MT4 15 12 14 11 14 11 13 10 12	337 MT 5 178 170 209 190 227 199 227 227 2256	537 MT 6 131 117 136 162 185 165 166	MT7 81 88 86 91 104 99 94 98	HTB 75 84 82 80 97 47 77 70 86	MT9 1417 1058 90 744 778 655	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10
CP 1 2 3 4 5 6 7 8 9 10 11 12 13	61 98 BASEL MT1 246 263 264 293 284 353 353 302 378	436 INE INLET MT2 70 58 76 61 63 62 68 89 78	285 T, STATI: MT3 127 140 135 135 132 106 127 129 127 118 111	550 ON 2 14 15 12 11 14 11 13 10 13 13	337 MT5 178 170 209 199 220 227 199 224 227	537 MT 6 131 137 1362 187 1555 1676 1754 1758	MT7 81 886 91 1049 984 984 112	HTB 7542 820 897 7787 86 87 400 100 100 100 100 100 100 100 100 100	MT 9 147580 4448550 47 8550 47 8550 47	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8
CP 1 2 3 4 5 6 7 8 9 10 11 12 13 14 15	61 98 BASEL MT1 246 263 264 293 264 351 352 378 367 367 367 367 370 370 380	436 INE INLET MT2 70 58 76 63 662 68 978 78 78 81 81	285 T, STATI- MT3 127 140 135 135 129 127 118 111 132 116 139 141 108	550 ON 2 MT4 155 122 141 143 100 113 133 133 141	NT5 178 170 209 192 200 227 129 1224 227 202 256 216 217 202 256 217 202 256 217 202 256 217 227 228	557 HT6 131 137 136 162 187 155 163 176 154 178 178 178	MT7 811 888 861 1049 999 860 11257	MTB 75428877777888897477878899477889947789986	MT 9 141758890 74485664477885	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9
CP 12 3 4 5 6 7 8 9 10 112 13 14 15 16 17	61 98 BASEL MT1 246 264 293 264 351 352 378 367 359 370 380 367 366	436 INE INLE MT2 70 58 76 61 63 62 68 76 86 70 81 82 88 91	285 T, STATI: MT3 127 135 135 137 129 129 118 131 132 118 139 144 108 115 123	550 ON 2 MT4 14 15 12 14 14 13 10 12 13 13 13 13 13 13	337 MT5 178 170 209 192 207 192 227 227 227 227 226 219 225 219 225 219 225 219 225 219 227 227 227 227 227 227 227 227 227 22	557 MT6 131 131 137 136 162 187 159 163 178 178 178 178 178 178	MT7 81 88 86 91 104 69 98 99 86 90 112 105 117 123 109	MTB 75 84 82 80 77 77 87 70 86 87 100 98 86 109	MT 14175880 444855047873188974	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9 11
CP 12 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18	61 98 BASEL MT1 246 263 229 284 351 352 352 358 367 351 359 367 374 370 366 356	436 INE INLET 70 58 76 61 63 68 69 78 76 61 82 88 91 80 81 75	285 7, STATI: MT3 127 135 135 137 127 127 118 131 132 116 139 144 108 115 123 123 133	550 ON 2 MT4 14 15 12 14 14 13 10 12 13 13 18 12 19 14 14 14 14	337 MT5 178 170 209 192 207 199 227 227 227 227 227 226 219 227 227 227 227 227 227 227 227 227 22	557 MT6 131 131 137 136 162 187 155 163 178 178 178 178 163 164 163 163 164 167 178	MT7 81 81 88 86 91 104 69 98 94 112 1157 1123 1109 116	MTB 75 84 82 80 97 77 87 78 86 87 100 98 100 112 130	MT 9 1417 105 88 974 48 58 674 78 58 674 78 58 674 78 58 674 78 78 18 78 674 78 78 18 78 674 78 78 18 78 674 78 78 78 78 78 78 78 78 78 78 78 78 78	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9 11 7 13
50 CP 1 2 3 4 5 6 7 8 9 11 11 12 13 14 15 16 17 18 19 20 20 20 21 20 21 21 21 21 21 21 21 21 21 21 21 21 21	61 98 BASEL MT1 246 263 2293 284 351 352 378 351 352 374 374 370 380 366 355 366 353 367	436 INE INLET 70 58 76 61 63 68 69 76 70 81 82 88 91 90 81 75 105	285 T, STATI MT3 127 140 135 132 106 127 118 111 132 116 117 118 111 132 116 113 117 118 111 132 116 117 118 117 118 117 118 117 118 118 118	550 ON 2 HT4 15 12 14 13 13 13 12 12 12 19 14 12 14 17	NT5 178 178 1209 190 220 227 202 227 202 219 224 267 267 213 214 225 213 214 225 225 225 227 227 227 227 227 227 227	HT6 131 131 137 1362 1862 1859 1553 1764 178 178 178 178 177 178	MT7 81 81 88 86 91 104 69 98 60 112 1057 1123 1109 116 1139	75 84 82 80 97 77 78 78 78 86 86 109 98 86 112 112 112 113 113 1143	MT 9 141 157 1058 90 744 748 655 600 749 781 910 891	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 9 3 8 5 9 11 7 15 11 13 13
50 CP 12345678911121341561771189022223	61 98 BASEL MT1 2463 263 2293 284 351 352 378 351 357 367 370 380 366 350 366 350 366 350 366 366 366 366 366 366 366 366 366 36	436 INE INLE: MT2 70 58 76 61 63 62 689 78 80 81 82 89 90 81 75 105 97 97	285 T, STATI: MT3 127 140 135 132 104 127 128 127 128 139 140 131 132 134 144 144 144 155 123 133 144 147 117 117	550 ON 2 MT4 14 15 12 14 14 15 12 13 10 12 13 18 12 11 12 14 17 16 17	337 HT5 178 1709 192 200 227 199 224 227 225 226 237 245 245 245 221 221	HT6 131 131 137 1362 1862 1563 1764 178 178 178 1643 1643 1643 167 178 178 178 178 178	MT7 81 81 88 88 89 99 94 99 91 105 117 123 109 116 113 109 126	MTB 75 84 4 82 80 97 67 77 87 86 87 98 86 112 112 112 112 112 112 113 115 115	MT 9 141 157 1058 90 744 748 655 600 749 781 1106	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9 11 7 13 11 13 13 13
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50 CP 12345678901123145167189222222222222222222222222222222222222	61 98 BASEL 263 263 224 353 353 353 353 353 353 353 353 353 35	436 INE INLET 70 58 67 61 63 62 68 67 78 78 78 79 79 71 91 91 92 92 92 92	285 T, STATI MIS 127 140 135 132 129 127 129 127 129 127 140 101 132 116 117 123 135 135 135 135 135 135 135 135 135 13	550 ON 2 HT4 15 12 14 13 12 13 8 12 11 12 13 16 17 16 17 20 21 17 22 28	337 MT5 178 1709 192 192 207 202 219 2247 205 215 225 215 227 221 227 227 227 227 227 227	557 MT6 131 131 137 1362 187 155 176 167 178 171 186 164 164 165 177 178 179 179 179 179 179 179 179 179 179 179	MT7 81 81 88 86 91 104 69 98 86 91 105 115 112 1109 116 113 119 115 119 1195	75 84 82 80 97 77 77 87 86 87 98 86 109 112 112 112 112 112 112 112 112 112 11	MT9 141 157 105 80 744 748 655 60 748 978 978 1106 891 106 992 1097 11162 1121 1121	MT10-TANG MEAS 99 89 255 11 14 77 13 14 10 9 3 8 5 9 11 7 13 11 13 13 13 13 13 13 13 13 13 13 13
50 CP 1234567890112314516718922222455678901223455678901223245567890122334	61 98 BASEL 263 263 224 355 352 358 357 351 357 351 357 351 357 367 367 367 367 367 367 367 367 367 36	436 INE INLET 70 58 67 61 63 62 68 69 78 78 78 86 79 105 91 105 93 93 93 93 91 101 85	285 T, STATI MIS 127 140 135 132 129 127 129 127 129 127 129 127 129 127 129 127 136 137 140 140 140 140 140 140 140 14	550 ON 2 HT4 15 12 14 13 12 13 8 12 11 12 13 16 17 16 17 16 17 17 16 17 17	337 MT 5 178 1709 192 192 207 202 219 2247 205 225 227 227 227 227 227 227 22	557 MT6 131 131 1362 187 155 176 167 178 171 186 164 164 165 177 178 179 179 179 179 179 179 179 179 179 179	MT7 81 88 88 91 104 69 98 86 91 115 115 117 1123 116 113 116 113 116 113 115 119 115 119 115 119 115 119	75 84 82 80 97 77 87 86 87 98 86 109 98 86 112 112 112 112 112 112 112 112 112 11	MT 9 141 157 105 80 74 748 655 660 748 781 89 1106 89 1106 89 1106 1132 1121 1121	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9 11 17 13 13 13 11 15 20 9 18 12 13 14 19 9 18
50 CP 12345678901123145167189012222455678901233456	61 98 BASEL 263 263 284 355 352 378 351 351 351 351 366 353 367 367 368 369 369 369 369 369 369 369 369 369 369	436 INE INLET 70 58 67 61 63 68 69 78 78 68 80 91 81 91 91 91 91 91 91 91 91 91 91 91 91 91	285 T, STATI MI3 127 140 135 132 129 127 129 127 140 118 111 121 132 116 123 135 135 135 135 135 135 135	550 ON 2 HT4 15 12 14 13 13 12 13 13 12 11 14 17 17 17 17 17 17 17 17 17 17 17 17 17	337 HT5 178 1709 192 192 207 207 207 207 207 207 207 20	557 HT6 131 131 1362 187 1553 1764 169 178 1643 1644 1662 1778 178 178 178 178 178 178 178 179 179 179 179 179 179 179 179 179 179	MT7 811 886 91 104 698 94 990 1125 1177 1223 1173 1240 109 1140 109 1159 1159 1159 1159 1159 1159 1159	75 84 82 80 97 77 77 87 86 87 98 86 91 109 86 91 112 112 112 112 112 112 112 112 112	MT 9 141 157 1058 974 748 655 660 774 891 1068 992 107 1135 121 121 1253	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 9 11 17 13 13 13 11 15 20 9 18 12 13 13 13 13 13 13 13 13 13 13 13 13 13
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50 CP 1234567890112345678901223456789012335567890	61 98 BASEL 263 263 263 353 352 378 351 357 351 356 357 367 367 369 369 369 369 369 369 369 369 369 369	436 INE INIZ 70 58 612 70 58 68 69 78 78 68 79 78 82 91 81 75 105 95 97 110 81 82 91 81 101 85 87 84	285 T, STATI M13 127 140 135 132 129 127 129 127 129 127 136 131 131 132 136 135 136 137 140 123 135 136 137 127 128 139 140 125 126 139 111 126 139 111 126 127 128 128 129 120 120 120 120 120 120 120	550 ON 2 HT4 15 12 14 13 12 13 12 14 17 16 17 18 17 18 17 18 17 18 18 18	337 HT 5 1709 1709 1920 1	557 HT6 131 137 1362 187 1553 178 178 178 164 164 164 164 164 164 169 1159 164 169 1159 164 165 173 170	MT7 81 88 88 88 91 104 69 94 99 86 112 117 122 117 123 124 109 114 109 124 109 125 115 129 121 145 118 129 128	#T8	MT 9 1477 1058 974 4855 660 477 873 189 168 992 974 132 221 112 122 112 122 112 122 112 12	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 9 3 8 5 5 9 11 17 13 13 13 11 15 20 9 18 12 13 13 13 11 15 20 9 18 12 13 14 17 20
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50 CP 123456789012345678901234567890123456789012345678901234567890123456789012345678901234567890123456	61 98 BASEL 243 263 2293 351 352 351 351 351 365 365 365 365 365 365 365 365	436 INE INIZ 70 586 878 766 63 688 776 688 891 808 815 105 877 110 933 907 822 141 855 877 110 855 877 843 884 877 878 884 884 877 884	285 T, STATI MI3 127 140 135 132 129 127 129 127 129 1216 1313 1316 115 135 140 117 125 114 125 115 125 126 127 127 127 127 127 127 127 127 127 127	550 ON 2 HT4 15 12 14 13 12 14 13 18 12 19 14 17 16 17 18 17 18 18 19 19 19 19 19 19 19 19	337 HT 5 1780992077 12099207221947 12099207221947 120	557 HT6 131 131 137 1362 1859 1553 1764 1678 1771 178 1641 1649 1641 1651 1651 1651 1671 1671 1671 1671 167	MT7 81 81 88 86 91 104 99 94 96 112 117 1123 1196 1198 1129 1195 1188 129 1195 1189 1295 1189 1295 1296 1297 127 127 127 127 1299	#T8 75 84 82 80 97 77 77 78 86 87 100 86 112 130 112 123 121 123 124 123 124 123 124 123 124 123 124 123 124 125 126 127 127 128 129 129 129 129 129 129 129 129	MT 9 1477655 807748 855 860 47748 87	MT10-TANG MEAS 99 89 25 11 14 7 13 14 10 10 10 9 3 8 5 9 11 13 13 13 13 13 13 13 13 13 13 13 13
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CP	BASELINE INLET, MT1 MT2	STATION 3 MT3 MT	4 HT5	MT6	HT7	MT8	HT9	MT10-TANG MEAS	
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СР	BASELINE INLET, MT1 MT2	MT3 MT		MT6	MT7	MT8	MT9	HT10-TANG MEAS	
123456789012345678901234567890123456789012345678901234567890	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	144 41 2 56 40 139 17 922 7 9101 9 108 9 118 13 112 126 110 118 13 117 112 117 116 118 13 139 14 130 139 14 100 64 9 77 9 91 10 64 9 77 9 10 64 9 77 1 30 4 11 10 00 00 00 00 00 00 00 00 00 00 00	24 5563 638 6565 6365 6468 7.566 648 83 667 7.592 1.592 1.592 1.592 1.592 1.592 1.592 1.592 1.592 1.593 1.59	167 466 8746 1748 1748 1798 1797 1797 1797 1797 1797 1797 179	210627 17798 100627 17798 100629 100629 100629 10088 1	20 338 452 75 76 89 89 89 81 1120 1121 1121 1121 1121 1121 113 97 1125 1149 95 115 126 127 128 129 129 129 129 129 129 129 129 129 129	13 14 14 15 16 17 16 16 17 17 18 18 18 19 19 10 10 11 11 10 10 11 11 11 11 11 11 11	115 124 136 149 136 136 137 138 135 138 131 139 136 137 148 149 157 177 188 188 188 188 188 188 188 188 18	
								OF POOR	QUALITY

CP	BASELI MT1	NE INLET, MT2	STATIO MT3	N 5 MT4	NT5	MT6	MT7	MT8	MT9	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	00077748333374815478616630224881252188300000000000000000000000000000000000	000765438701641382919920302783611949213221020000000111111111111111111111111	0 0 9 5 5 5 5 1 2 5 6 5 1 2 5 6 5 6 7 9 5 5 5 6 7 9 5 5 5 6 7 9 5 6 5 6 7 9 6 5 6 7 9 6 7 9 6 7 9 6 7 9 7 9 7 9 7 9 7 9	00 31 324 47 145 166 1111 102 100 100 100 100 100 100 100 10	04462344956888699101111111111111111111111111111111	08332414447605292590652479093088777555441158210000000000000000000000000000000000	29809358651342100715586935869358693586935869358693586935869	25359052532667753266839759008839858250022735101239888888977985888889759000000000000000000	952268369167513840616705442898604549121027000000000000000000000000000000000	02116453736770775123677281197792426713351842518210500000000000000000000000000000000
C 123456789012345678901234567890123456789012345678901234567890	BAT 00339726520978889289232052851 2181111111111111111111111111111111	INE TNLET MT2 0 0 929 1188 0 1449 1153 1157 1167 1153 1164 1139 1170 543 1170 1103 7 117	NT3 0 0 176511588222212658512121212222222222222222	ON 6 MT 4 0 6 6 7 8 6 7 9 7 4 9 9 8 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	MT 4353287991536903579911006667797110066677991100897915564889666019221000000000000000000000000000000	MT	MT 09998399144304374525057465783505745582111580465940500000000000000000000000000000000	MT 03460226974917295142386222697491729514238622111423862211142386221111111111111111111111111111111111	MT9 01110985140222936651944173147559861884488822000000000000000000000000000000	MT10-TANG MEAS 0 27 27 32 15 30 37 15 32 16 20 24 27 29 24 25 20 26 27 50 53 44 29 56 47 55 29 64 29 64 27 79 44 29 41 41 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0

CP	BASELI MT1	NE INLET, MT2	STATIO	IN 7 MT4	NT5	HT6	HT7	MT8	MT9	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	0006180729541530289997803273582337398079978608 8111516981302899978032735823373980799786060	0904534047756189511930266191630607595873020863700 11245340477561895119930265191529587302011 1122224222232222222222222222222222222	0067599179979159251878622226577348211625367326200	0525777040179254727779903438548901235612833434136600	0 0 74 67 83 87 93 1146 125 147 146 152 147 165 150 150 150 150 150 165 165 165 165 165 165 165 165	0 0 0 54 49 62 69 96 97 1121 1001 131 145 157 138 164 165 166 168 160 165 166 167 653 161 12 67 653 10 12 11 13 12 29 0	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	001779105733060148477777777985776889405666421188390	00 399 508 488 464 668 880 951 104 104 104 116 117 116 117 117 118 118 119 119 119 119 119 119	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
CP 1	BASEL! HT1	INE INLET MT2	, STATII MT3	DN 8 MT4 17	MT5 20	MT6 13	MT7 21	MT8 30	MT9	MT10-TANG MEAS
2345678901234567890123456789012345678901234567890	103102220110000000000011100113202377367305443969	1 11 12433314635455335534342433344444689676242	1166778912878950896280773911933163771698418163571667789508962807773911191333163716986283	144 233 333 539 698 734 1244 1351 1198 1199 1199 1199 1199 1199 1199 11	145 177 21 188 499 388 554 664 870 102 1126 1126 1127 1126 1127 1128 1129 1137 1129 1137 1129 1137 1129 1137 1149 1159 1159 1169 1179 1179 1179 1179 1179 1179 117	181 355 259 355 553 769 63 708 1103 1112 1127 1127 1127 1127 1127 1127 112	30 41 26 58 58 57 77 64 107 111 152 114 115 115 127 114 115 127 114 115 127 115 115 115 115 115 115 115 115 115 11	376 379 488 488 775 775 775 781 883 892 1062 1169 1179 132 1109 132 1109 1118 1143 1150 1151 1151 1151 1151 1151 1151 115	232 350 455 465 570 577 657 657 657 657 657 657 657	71 67 79 66 71 88 93 95 94 95 77 98 88 94 80 105 70 144 94 72 100 67 82 104 71 103 101 103 101 103 101 103 101 114 90 98 88 114 96 98 114 153 1153 127 1222 276 356 445 159 98

CP	ENHANG MT1	CED INLET	, STATIC	N 1 MT4	MT5	MT6	MT7	MTB	MT9	MT10-TANG MEAS
123456789012345678901234567890123456789012345678901234567890	4444465544595555564545456555645564455554554554554554	461164820648877774384182318874364667156539356907761	656527986821750673431589161718770751155268867934662	675566665758676863322369782942783139194581634628491275	578465772169433738677687655788756775694377887557788755778875577887657776602	75848831055558441564488865677145514917665539974706282	648497626539403071899711985669159653717446053018583430	83 94 104 99 91 100 89 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 107 99 108 88 89 99 109 109 109 109 109 109 109 109 109	683654106294286938070685756541062946856541066294286938076667666755555555655278798730	2497 3237 3237 3237 3237 3237 3237 3237 32
CP	ENHANI MT1	CED INLET	, STATIO	N 2 MT4	MT5	MT6	MT7	HT8	MT9	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	90759808220584157161927647196621233528689034193142	106 107 111 111 1199 857 995 847 107 1187 1103 1103 1110 1110 1110 1110 1110 111	91261165145513599455335778865727294498498419969899889988990847888990849981094788883374	926203B705309527956320109623B330018229029618252193	655565577656666666598040792929290355320328071770602	111 933 185 86 867 987 1022 651 677 776 800 801 874 82 932 857 89 97 108 89 113 89 114 89 115 115 116 117 117 118 118 119 119 119 119 119 119 119 119	54752129400606077779586619577662222040877849884408	5510416075836534734845378344108407800717676675666	84480674866690491243434122732428644874876238114556 458543546456564464355455555685566555566645555546	17 16 123 123 124 164 17 120 130 131 131 131 131 131 131 131 131 13

CP	ENHANCI MT1	ED INLET, MT2	STATION MT3	3 HT4	MTS	MT6	нт7	MT8	MT9	MT10-TANG	MEAS
12345678901234567890123456789012345678901234567890	0 0 3 0 0 0 0 0 5 14 2 19 4 2 2 2 2 2 2 2 3 3 2 3 2 3 2 3 3 3 3 3	128 0 0 0 0 0 0 2 2 0 10 0 1 1 3 5 1 1 4 2 1 1 1 1 3 7 1 1 4 4 1 1 1 1 1 3 7 1 1 4 4 4 1 1 1 1 1 3 7 1 1 4 4 4 1 1 0 0 0 0 0 0 0 0 0 0 0 0 0	0000000234573391175458414262440276526167074851500	00090000188692618271966782268948967364075363172210	0 0 12 0 0 12 0 0 12 0 0 12 0 0 12 0 12	0 0 1 4 0 0 0 5 4 5 1 4 6 9 3 1 1 2 3 2 3 5 4 6 6 6 7 7 7 7 6 7 6 7 7 7 7 8 3 8 8 9 9 3 4 7 4 8 1 9 2 2 7 2 3 8 8 9 9 9 1 3 2 8 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 1 3 2 8 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	00360003987931052446525902291742446167779580537307425	00058000711074189697777779877983794979852452177886584993555	0 0 0 121 0 0 0 121 0 0 0 121 0 0 0 123 2 19 4 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6	51012 158000012113142431242223541211131437133457464366	
СР	ENHANC MT1	ED INLET, MT2	STATION MT3	1 4 HT4	MT5	MT6	HT7	мт8	MT9	MT10-TANG	MEAS
12345678901234567890123456789012345678901234567890	0	0 0 0 6 8 5 6 9 0 1 1 2 6 9 9 3 2 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 0 3 2 0 8 1 0 4 4 2 7 9 2 7 1 7 1 1 2 1 4 4 1 4 9 5 3 7 6 5 2 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0123650025085271917276044768274231000000000000000000000000000000000000	0 2 3 8 12 45 41 51 67 68 88 116 7 117 117 117 117 117 117 117 117 1	163 242 666 1723 993 993 103 993 1017 1117 1117 1117 1117 1117 1117 111	542 623 720 807 777 889 1043 1111 1112 1112 1112 1112 1112 1112 11	922716681778940824127788208477894040837708208478855290000000000000000000000000000000000	203382222445173556457777889788997869710654468220000000000000000000000000000000000	0 POC	IAL PAGE IS OR QUALITY

CP	ENHANC MT1	ED INLET	, STATIC	ON 5 MT4	MT5	MT6	MT7	MTB	MT9	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	0001425430212522556790804727551920143986000000000000000000000000000000000000	00000000000000000000000000000000000000	000229812353064851147424976700188225176000000000000000000000000000000000000	0 9 30 410 577 785 1252 1454 1458 1459 1460 1150 1160 1170 8150 981 1100 1100 1100 1100 1100 1100 1100	0715 115 125 125 125 125 125 125 125 125 1	0 39 45 57 77 88 101 101 91 104 114 115 117 115 117 115 117 48 55 46 23 33 10 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	18 433 482 59 59 58 87 718 681 99 102 1105 1105 1103 1105 1101 1100 112 91 837 760 00 00 00 00 00 00 00 00 00 00 00	29 344 61 539 707 675 777 938 886 810 99 104 1126 1126 1126 1138 109 109 100 100 100 100 100 100 100 100	26 367 446 468 468 468 774 573 573 605 749 749 749 749 749 749 749 749 749 749	25 44 10 41 23 55 44 12 39 47 34 45 57 52 59 52 48 57 61 69 76 69 76 69 71 80 75 80 80 75 80 80 80 80 80 80 80 80 80 80 80 80 80
CP 1	MT1 O	ED INLET MT2 0	MT3 0	MT4 0	MT 5	MT 6	MT7	MT8	MT9 0	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	0410857367489895981035232111588408646512000000000000000000000000000000000000	03651 1121036180817922204237 24237325116811224961000000000000000000000000000000000000	01201212886968022215686176941750088483460000000000	946579757189992680151657922864207033149773800000000	125 126 227 356 277 790 101 119 119 119 119 119 119 127 125 125 125 125 126 130 130 141 151 161 173 173 173 173 173 173 173 174 175 175 175 175 175 175 175 175 175 175	1543891795867743788873399100295149495867645887339910029514945880000000000000000000000000000000000	634 634 1011 1123 1440 1580 1580 167 168 167 178 1207 179 179 179 179 179 179 179 179 179 17	952 1010 127 1140 127 1140 127 1150 1150 1150 1150 1150 1150 1160 1172 1172 1173 1173 1174 1175 1175 1175 1175 1175 1175 1175	54589099772280923646446988341867412444247400000000000000000000000000000	C:514

CP	ENHANO MT1	CED INLET	, STATIO	DN 7 MT4	MTS	MT6	MT7	MT8	нт9	MT10-TANG MEAS
12345678901234567890123456789012345678901234567890	0 0 2 4 1 4 2 5 4 7 6 5 1 1 1 8 1 8 4 4 7 2 3 2 2 2 4 3 1 0 1 2 2 5 7 2 3 1 8 7 2 2 1 2 1 5 1 2 2 5 7 2 3 1 8 7 2 2 2 3 1 8 6 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	07959916743759033844421129959535916743759403384444214959533929884709021000	0 0 141 19 123 8 6 5 8 73 8 122 1127 1126 5 5 8 122 127 1120 121 125 9 1227 1140 141 145 8 145 8 145 8 145 8 145 8 145 8 145 8 15 15 15 15 15 15 15 15 15 15 15 15 15	0 0 0 1 2 3 3 5 4 9 9 1 1 0 3 3 4 9 1 2 2 2 2 2 2 2 2 2 2 2 2 3 3 3 5 5 6 6 1 1 2 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0003127116297236855199235559664177726768841997988899999999999999999999999999999	0 0 12 15 29 35 35 76 78 99 125 130 147 138 147 153 140 147 149 153 140 149 160 175 180 192 192 193 193 194 195 195 195 195 195 195 195 195 195 195	0 0 39 49 70 104 115 118 115 118 112 123 121 123 124 127 128 128 129 129 129 129 120 121 121 123 141 145 146 147 146 147 146 147 146 147 146 147 146 147 146 147 147 148 149 149 149 149 149 149 149 149	0 45 45 72 81 77 77 117 1109 1109 1116 1118 115 125 125 125 125 125 126 127 127 128 129 120 120 121 120 120 120 120 120	00606130449713276866788678875927786322980	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
CP 1	ENHANO HT1 2	CED INLET MT2 5	MT3	DN B MT4 7	MT5 0	MT6	HT7 13	MTB	MT9	MT10-TANG MEAS
2345678901234567890123456789012345678901234567890	00000000000000000000000000000000000000	111122103160732852071898880898192451066093053795252322323233134434457053798622541924510660930525252525	244524673240001000010002100001238911200010001000110001238911200010010011000110001100011000110001	128779752719358888682020294130057875721343060752 119358884686220294130057875721343060752	0000010001111111310012101011222000244488161740	733652558918144695147688997756889977568899977568899977588918112568832	126 144 219 238 461 76 89 96 122 145 123 123 123 123 123 123 123 123 123 123	23 15 17 20 39 40 51 86 91 86 91 107 107 119 103 1126 1127 114 1121 1148 117 1168 117 1168 117 1188 117 1188 1170 1188 1170 1188 1170 1188 1170 1188 1170 1188 1170 1188 1170 1188 1170 1188 1170 1188 1188	142 102 112 113 114 115 115 116 117 117 117 118 118 118 118 118 118 118	102882688181180666718462240055132129900432540941735575

TABLE A5.
(e) CALCULATED AXIAL UNCERTAINTY (PERCENT)

СР	BASEL EA1	INE INL	ET, STAT	ION 1 EA4	EA5	EA6	EA7	EAB	EA9	EA10-X AXIAL ERROR
1234567890123456789012345678901234567890	11111111111111111111111111111111111111	0.66759762516003667500000667500000066750000000000000	0.985 0.955 0.955 0.951 0.897 0.895 0.997	1.13 1.123 1.089 1.194 1.428 1.095 1.127 1.386 1.127 1.386 1.345 1.080 1.335 1.128 1.233 1.245 1.268 1.268 1.268 1.271 1.288 1.295 1.208 1	0.0.000 0.0.0000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.0000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.0000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.000 0.0.0000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0	247760602397791302981601186173080145968886686388810019 	0.8772518545067847058239779087695959807537878900.88877853545067888778579778798778787878787878787878787	8.88844 6.77765 8.88884 6.77765 8.677776 8.677776 8.6777778 8.677778 8.677778 8.677778 8.677778 8.677778 8.677778 8.677778 8.67778 8.677778 8.677778 8.67778 8	1.0401 1.0401 1.1829 1.1229 1.1227 1.	3 93 9 9 1 9 3 9 1 9 3 1 2 3 5 1 2 2 5 1 2 2 6 3 7 7 2 8 1 2 2 6 1 3 8 1 9 1 6 1 6 6 9 1 7 4 9 1 7 7 8 1 8 9 9 4 9 8 1 9 6 9 6 9 6 9 6 9 6 9 6 9 6 9 6 9 6 9
СР	BASEL Eal	INE INLI EA2	ET, STATI	ION 2 EA4	EA5	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
123456789012345678901234567890123456789012345678901234567890	0.6777759243333151127738111836170554733345977735924333315151127738111836170554733345979882778392	3.3343539861567473058705660188888813469554022516153 4.4614213301357909688545904209417946878677202516153 5.534353455554345343444459413333222222222222222	11.820.526.653.612.66.653.653.653.653.653.653.653.653.653.	10.343 8.550 117.6136 117.6136 117.6136 117.6136 117.6136 117.6136 118.793 116.617 121.6617 121.6617 121.6507 121.6507 121.6507 121.6507 121.6507 121.6507 122.7553 110.553	0.6747377685315966193095536918336663360208347337568531509553691833666236208347337587587587587587587587587587587587587587	0.65329861666421067033346893191111111111111111111111111111111111	0111112122223223222222222222222222222111111	011111112222222222322223322222222222222	0.01111112223222222222222332222212222111111	2.12 3.50 8.60 11.75 15.57 10.77 6.71 8.00 24.23 12.35 8.18 1.10.36 11.36 11.10.36 12.88 15.09 13.38 13.76 8.80 10.00 9.02 20.98 15.13 16.70 11.39 17.28 8.89 12.80 11.39 14.65 7.33 8.99 12.80 11.76 9.95 11.34 11.35 7.33 8.99 12.80 11.76 9.95 10.34 11.35 7.33 8.89 12.80 11.76 9.95 10.34 11.35 7.38 8.75 10.31 6.30 6.63 6.63

CP	BASELIH EA1	NE INLET EA2	, STATI EA3	OH 3 EA4	EA5	EA6	EA7	EA8	EA9	EA10-X	AXIAL	ERROR
1234547890123456789012345678901234567890	1	65.6880000000000000000000000000000000000	1.80 0.00 0.00 0.00 0.00 0.00 0.00 0.00	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 110.25 111.31 123.30	33.00.000 21.660 00.000 13.156 14.7.486 27.660 13.156 14.7.486 15.156 15.156 16.156 17.7.219 1	0.00 55.04 284,940 0.00 0.130 5.87 4.94 12.18 22.63 22.64 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 22.46 23.09 1.80 1.50 1.75 1.66 4.95 1.75 1.66 1.76 1.	13.44.8140000 11.44.81400 10.0000 11.44.81400 11.44.81400 11.44.81400 11.44.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400 11.48.81400	0.004 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.000 0.004 169.0000 169.000 169.000 169.000 169.000 169.000 169.000 169.0	0.660249000112291716952218062071976187333650360 0.660489000112291716953673091140937197653973091147333650360 0.00118269177663.28068719706339650360 1104476623769117796788662177888662173633650360	0.00 20.99 553.60 67.88 0.00 0.00 0.00 109.28 51.28.52 63.00 120.52 63.00 120.52 63.00 140.00 151.85 1120.52 63.00 151.85 120.52 63.00 151.85 120.52 63.00 151.85 120.52 63.00 151.85 120.52 63.00 151.85 120.52 63.00 151.85 120.52 63.00 151.85 120.52 120.5		
СР	BASEL I Eal	NE INLE	T, STAT	ION 4 EA4	EAS	EA6	EA7	EA8	EA9	EA10-	% AXIAL	ERROR
12345678901234567890123456789012345678901234567890	0.000 0.000	0.0008894571644518944515894451886656854455894451876878787878787878787878787878787878787	0.007799.617732893.661799.65.28745.339.849612.15.5700.141.53.39.849.33.664.857.5700.000.000.000.000.000.000.000.000.	0.007 9.801 13.485 6.665 6.758 6.675 6.675 6.675 6.757 7.777 1.608 1.777 1.778 1	0.9726900000000000000000000000000000000000	24.801 7.881 15.467 10.5.467 1	41. 43 12. 39 17. 90 9. 84 4. 30 6. 41 6. 7. 95 6. 41 6. 7. 95 7.	9.74 8.78 4.84 5.89 4.84 5.89 5.25 5.42 5.35 5.42 5.35 5.42 5.35 5.42 5.35 5.42 5.35 5.42 5.42 5.42 5.43 6.43	6.943.4649.11.41.41.41.41.41.41.41.41.41.41.41.41.	45.48 46.65 25.65 27.57 9.27 16.37 9.27 16.38 16.38 9.06 43 8.85 9.06 6.71 6	SINAL	- PAGE IS

CP	BASELI Eal	NE INLE	ET, STATI EAS	ON 5	EA5	EA6	EA7	EA8	EA9	EA10-2	AXIAL ERROR		
12345678901123415678901123456789001233456789001234567890012334567890012344444444444444444444444444444444444	0000027110005000000000000000000000000000	000056784489396202764494288675437444557800630000000000000000000000000000000000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0.00842663736266437375266673334285468522222222222222222222222222222222	0.95771814458899497148924338446131835434343949714892243384491300000000000000000000000000000000000	0.007 0.170 0.	0.0553934.2097749262.21812.22.234.34.851712.000.000.000.000.000.000.000.000.000.0	8647.6437.6827.735.64480827.54444333333333333333333333333333333333	14.1703 1703 1703 1703 1219 1013 1290 1152 1290 1290 1290 1290 1290 1290 1290 129	20 5794 10 5794 110 5797 12 9 9 110 5797 12 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9			
СР	BASEL EA1	INE INI EA2	LET, STAT	TION 6 EA4	EAS	EA6	EA7	EAB	EA9	EA10-%	AXIAL ERRO	ıR	
123456789011234567189011234567890112345678901123456789011234567890	7.80 69.45 0.00 0.00 0.00	0.000 0.0000 0.0000	0.004 0.004	0.09 6.99 5.88 5.78 4.83 5.78 4.29 6.77 5.90 6.22 6.22 6.23 7.72 7.72 7.72 7.72 7.72 7.72 7.72 7	0.986350772869356959599995777455257772455507772869355557774555077728777745550745000000000000000000000000000	0.924 6.1433 6.1	0.00 3.07 2.46 2.75 2.04 1.90 2.10 1.87 2.03 1.87 2.03 1.87 1.57 1.37 1.37 1.37 1.37 1.37 1.37 1.37 1.3	0.00 3.13 2.33 2.33 2.33 2.33 2.33 2.33 2.33	0.00 19.48 8.78 6.88 11.86 6.88 11.99 10.99 11.14 19.33 17.64 11.33 12.64 12.6	071026755203960896029587747142849900773945183000000000000000000000000000000000000			PAGE IS

CP	BASE! EA1	LINE INL	ET, STAT EAS	ION 7 EA4	EA5	EA6	EA7	EAB	EA9	EA10-X AXIAL ERROR
12345678901234567890123456789012345678901234567890	0.00 0.00	0.00 11.84 6.23 5.088 2.981 2.57 2.2.11 1.97 1.73 0.49 2.22 1.97 1.73 1.73 1.74 1.67 1.74 1.67 1.74 1.68 1.67 1.74 1.68 1.67 1.74 1.74 1.68 1.74 1.74 1.74 1.74 1.74 1.74 1.74 1.74	0.00 0.00 10.70 14.59 10.90 16.63 17.30 17.69 17.55 18.77 18.7	0.00 0.034 14.32 64.627	0.00 0.00 6.713 6.71	0.007 0.007 10.402 4.463 2.99 4.4563 2.532 2.332	0002482455307775123180000024824553535232222222212111111111111111236954470	0007908239183778335782522222121121221111111111111111111111	0.000 0.000	0.00 0.00 3.24 2.89 2.89 2.89 3.90 3.50 3.19 3.19 3.19 3.50 3.19 3.50 3.19 3.19 3.50 3.19 3.50 3.19 3.50 3.79 3.50 3.79 3.70
CP	BASEI Eal	INE INL EA2	ET, STAT EA3	ION B EA4	EA5	EA6	EA7	EAB	EA9	EA10-% AXIAL ERROR
12345678901234567890123456789012345678901234567890	2 827 32 510 0 000 0 0 000 0	0.010000000000000000000000000000000000	6.21 3.84 1.407 1.077 1.077 1.087 1.186 1.181 0.744 1.186 1.029 0.42 0.50 1.055 0.78 0.67 0.67 0.67 0.67 0.67 0.67 0.67 0.67	7.19 5.13 5.23 7.44 2.136 3.805 1.810 1.280 1.181 1.280 1.181 1.280 1.181 1.280 1.181 1.280 1.181 1.280 1.181 1.280 1.181 1.280 1.28	36, 316, 316, 316, 316, 316, 316, 316, 3	12.8704309375339607188706666532548710993785753960733785733960733785733602477737666653254871000000000000000000000000000000000000	6.4653 5.353 5.4658 6.4	4443455342225322212222222222222222222222	5.2552.45.52.245.6799.941.620.535.564.5177.3635.245.56.775.535.56.7775.355.245.56.7775.355.245.56.7775.355.245.61.79.25.79.25.79.	1.00 1.22 1.04 1.78 1.52 1.41 1.51 1.41 1.93 1.62 1.72 1.95 2.16 2.18 9 1.72 1.95 2.18 9 1.72 2.38 2.11 2.46 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.31 2.476 2.55 2.38 2.38 2.31 2.476 2.55 2.38 2.38 2.31 2.476 2.55 2.38 2.38 2.38 2.38 2.38 2.38 2.38 2.38

 $\gamma_{ij} = (i - N_i) = 0$

		CED_INLE	T, STATI	ON 1						
CP 123456789100112	ENHA1 2.696707073559609222322232223222322322322322322322322322	EA 0575644749357922.2222222222222222222222222222222222	EA3 3.27 3.04 2.71 2.72 2.41 3.20 2.78 2.47 2.46 2.80	EA4 3.01 2.690 2.488 2.778 2.451 2.530 2.758 2.5162	E A 5 3.08 3.98 3.72 5.72 5.13 4.31 4.31 4.31 4.31 4.31 4.31 4.31 4	E A 6 2.880 2.880 2.889 2.889 2.587 2.658 2.658 2.658	E A 7 8.48243339 6824.5539 6.5539 6.738 7.6738 7.88498	EA8 2.59 2.176 2.475 2.475 2.727 2.926 2.737 2.994 2.772	EA9 2.99 2.57 3.34 2.516 2.466 2.464 2.457 3.051	EA10-X AXIAL ERROR 4.28 5.14 6.43 5.41 4.49 6.43 6.53 4.70 3.78 6.35 4.78 3.78 6.35 4.78 5.15
111111222222222233333333333444444 5678901234567890123456789012345	2.2.2.3.4.169 2.2.2.3.4.169 2.2.2.3.5.169 2.2.2.3.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.	823444409713452462252988344921344539735452466275598837529227272727272727272727272727272727272	463569922596251762356731827466609910234997517623564905833338898739911	22222222222222222222222222222222222222	83353 0453 4483 4495 4495 1525	922222223552525222223522222222225555	2333222232222242323445537227424 61111156571561627622246537 6227423222423233452232227423	22223222232223222222223322222222222222	23.22.22.22.23.22.22.23.23.23.23.23.23.2	4.799 4.483 5.465 5.415 5.455 6.572 5.569 6.572 6.572 6.572 6.775 6.775 6.775 7.325
46 47 48 49 50	3.37 2.71 3.26 3.74 1.21	2.20 2.35 2.91 2.99 1.29	3.05 2.37 2.66 2.68 1.79	2.61 2.31 2.33 2.50 1.68	3.14 3.15 3.75 3.31 2.45	3.56 2.81 3.00 2.85 1.78	2.79 3.10 3.01 2.70 2.04	2.59 2.92 2.69 2.42 1.57	3.29 3.12 3.07 2.70 1.33	8.28 8.37 6.30 4.71 2.73
CP	ENHAN EA1	CED INLE EA2	T, STATI EA3	ON 2 EA4	EA5	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
1234567890111234567111111111111111111111111111111111111	235525343432534435353334333333223212222222222	11111222524545152469888899178884855444992292221111111111092252524545958888888554449922922221111111110	011111122222354235455251660446887545145522526693249528099181536 01776881247568654525166044688764459266693249528099181536	1-2-1-2-2-2-3-2-4-4-4-3-3-4-3-4-4-4-5-4-2-4-3-3-3-2-2-2-1-1-1-2-1-2-2-2-3-2-4-4-4-3-3-4-3-3-4-3-4-4-5-4-4-5-4-2-9-3-3-3-2-2-2-1-1-1-1-1-1-1-1-1-1-1-1-1-1	1012223333334454524685186224133850742233333333333333333333333333333333333	1012223555345444444444455555445442443435533552222111	1012323444444435554454564434324444333333322222221	1.0.81253200000166692771777031787505512555275486551253378750317879855527548651806320244999555275486552754865527522222211	10011112112242343346660009872358294694443973795951738	4.67 4.69 3.63 7.91 5.69 8.65 5.12 8.21 8.10 10.99 11.99 14.48 15.98 8.70 25.64 13.05 14.88 13.05 14.88 13.05 14.88 13.05 14.88 13.05 14.88 15.98 16.88 17.10 18.21 18.21 18.21 18.21 19.21
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CP	ENHA EA1	NCED INL	ET, STAT EA3	ION 3 EA4	EA5	EA6	EA7	EAB	EA9	EA10-X AXIAL ERROR
123456789011234 11123456789011234 111234567890112334567890112334444444444444444444444444444444444	00000000000000000000000000000000000000	0.00 0.00	0.00 0.00	8.00 9.00 9.00 9.00 9.76 9.00 9.00 9.00 9.00 18.33 17.01 128.83 17.01 128.83 17.01 18.51 12.67 13.92 14.50 14.50 15.50 16.32 17.01 17.01 18.51 18.51 19.50 18.51 19.50 19	0.0047 0.00314.0000 0.3470000 0.005.64.1513	0.000 0.000	9.00 9.00 9.00 9.00 9.00 9.812 9.812 9.814	8.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 19.71 7.00 10.64 9.7.71 7.03 13.44 29.41 88.745 94.22 94.23 94.23 97.50 10.64 13.44 14.65 16.67 17.59 17.59 17.59 17.59 18.66 18.	0000097630940303750830119622667000009976131966967788564694033555544599473341896222332222211115	5 26 27 .06 0 .00 1.30 3.43 0 .00 1.30 0.00 0.00 0.00 62 .16 125 .05 18 .92 27 .46 14 .51 41 .37 0.00 25 .76 10 .81 9 .85 13 .20 38 .05 11 .81 22 .85 13 .29 28 .18 18 .18 18 .18 18 .19 18 .19 18 .19 18 .19 18 .19 18 .19 18 .18
CP	ENHA EA1	NCED INL	ET, STAT EA3	ION 4 EA4	EA5	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
1234567890111234567890112345678901233456789012345678901234567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 7.80 67.28 7.80 10.15 11.23 7.58 8.05 5.71 6.16 10.15 11.23 7.58 8.05 5.71 6.16 10.15 11.23 13.59 8.05 13.59 10.15 11.23 13.59 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 11.23 10.15 1	0.00 0.00 131.40 23.21 229.38 9.94 6.08 4.113 5.08 5.08 5.08 5.08 1.70 2.47 2.02 1.70 1.54 1.42 2.31 1.42 2.31 1.42 2.31 1.42 2.31 2.31 2.31 2.31 2.31 2.31 2.31 2.3	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 127.599 200.698 214.620 7.245 11.178 7.056 7.405 11.178 7.056 7.405 11.178 7.056 7.405 11.178 7.056 7.405 11.178 7.056 7.405 11.178 7.056 7.405 11.178 7.056 7.405 7	10.775515886674441390344621105755237188667444290344221034221034221000000000000000000000	14.84 4.72.155 4.47.155 4.47.155 4.47.155 4.57.106 7.37.1	36.69 7.221 8.2367 9.214 6.997 3.237 9.3281 3.778 2.977 3.281 3.276 2.217 2.21	7.53 3.847 4.7.761 3.7.	0.00 0.00 0.00 20.17 24.68 0.00 25.17 24.68 0.00 25.15 30.25 9.83 20.91 5.44 7.11 10.81 10.77 11.21 5.74 7.15 5.89 4.5 6.1 11.21 5.74 5.89 62.75 61 18.49 12.39 62.75 61 18.49 12.39 62.75 61 18.49 12.39 62.75 61 18.49 12.39 62.75 61 18.49 12.39 62.75 61 18.34 01 9.27 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0

CP	ENHA EA1	NCED INL EA2	ET, STAT EAS	ION 5 EA4	EAS	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
123456789901112134567789901112134567789901112134567789901112133456778990111213345677899011121334564789011121345647890112134564789000000000000000000000000000000000000	0.00 0.00 0.00 10.378 11.66 9.154 9.154 9.375 12.88 14.055 31.23 22.126 24.412 27.726 125.136 125.136 127.726 125.136 127.726 125.136 127.726 125.136 127.726 127.726 128.726 129.726 12	0.00 0.00	0.00 0.00 23 5.95 24 9.53 15 7.55 6.76 77 .55 6.76 77 .55 6.76 6.94 4.10 5.24 4.10 5.25 4.10 5.24 4.10 5.25 6.20	0.00 0.00 4.77188 3.65443.3654 4.3.6453.3122 2.2119 2.21	0.8494295360175585764754543434333333333333333333333333333	0.007 3.571 3.015 22.577 21.952 21.533 11.648 11.631 11.60	9.866 9.870	5.40 14.31 4.80 5.49 5.49 5.49 5.89 5.19 4.72 4.72 4.72 4.72 3.96 4.72 3.19 2.10 2.67 3.19 2.67 3.19 2.67 3.19 3.19 3.19 3.19 3.19 3.19 3.19 3.19	3.688722066469828787878787878787878787878787878787878	5.180 5.180
CP	ENHAN EA1	CED INLE EA2	T, STATI EA3	DN 6 EA4	EA5	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
12345678901123456789012234567890123345678901234567890	0.000 7.393 110.592 111.552 111.1552 11	0.000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.00000 0.00000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.00000 0.0000 0.0000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.000000	0.000 18.386794001172000000000000000000000000000000000	0026651384512134696644565644491930388253138451214512468253138333333333333333333333333333333333	006427266666752785944720000117555533333333333333222222222223333546603772000000000000000000000000000000000	0.447.600447.520219904.60035.775.22.537.52.5.23.775	0.9998888888888888888888888888888888888	0.00 1.989 898 898 292 242 252 272 272 272 272 273 273 273 27	0.748875745245305444455754452472727272727272727272727272	original.

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CP	ENHA!	NCED INLE	T, STATI	ON 7 EA4	EAS	EA6	EA7	EA8	EA9	EA10-% AXIAL ERROR
12345678911123456789112334567891100000000000000000000000000000000000	0.000 0.0000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.00000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.	0.00 0.00 332.19 9.63 1.89 0.38 0.44 0.79 0.00 0.00 0.00 0.00 0.00 0.00 0.00	0.00 0.00 0.00 0.00 0.00 0.00 0.00 135.52 137.06 1347.06 1347.06 135.52 137.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06 137.06 138.55 147.06	0.00 0.007 71.560 36.290 36.290 35.201 35.201 35.201 44.31 45.689 22.299 24.437 10.364 11.207 11	0.001 0.001 95.534 64.880 95.534	0.007 0.	00005384 00005386 000056	00001123.1284 000001123.1284 000001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 000001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 00001123.1284 000000000000000000000000000000000000	0022228994349808748266441285717999608887441985556144443002222123333343444444444444444334333333333	0.00 0.00 0.00 0.00 0.00 0.00 37.77 25.39 0.80 38.42 22.76 10.21 22.81 10.21 17.72 17.90 14.56 17.58 17.57 14.55 10.20 35.34 12.81 14.55 16.83 17.72 17.90 14.56 17.72 17.90 14.56 17.72 17.90 14.56 17.58 17.58 17.58 17.58 17.58 17.58 17.58 17.58 17.64 17.58 17.59 18.54 24.28 18.64 25.64 27.64 23.31 10.43 10.18 11.10 10.18 12.11 10.28 4.43 2.16 182.84 0.00 0.00 0.00
CP 1 2 3 4 5 6 7 8 9 10 11 2 3 3 4 5 6 7 8 10 11 2 3 14 5 16 7 18 19 2 2 2 2 5 6 2 7 8 2 9 9 3 3 1 2 2 3 3 4 5 6 4 4 6 4 6 4 6 4 6 4 6 4 6 6 4 7 8 9 4 6 6 6 4 7 8 9 5 10 10 10 10 10 10 10 10 10 10 10 10 10	EA1 0 0000000000000000000000000000000000	120.63 1.20.63 1.30.63 1.30.63 1.30.63 1.50.63 1.75.66 7.3.71 3.80.63 1.75.66 7.3.71 3.80.63 1.50.79 96.65 7.3.80.63 1.50.79 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 96.10 9	EA3 0.39 0.39 0.39 0.37 0.454 0.357 19.36 12.47 19.36 16.47 19.36 16.47 17.40 18.46 17.47 19.36 16.57 17.40 18.46 18.47	EA4 1.27 0.692 0.981 1.35 1.955 57.82 0.000 0.000 0.12 1.45 1.75 1.61 1.75 1.75 1.61 1.75 1.75 1.75 1.75 1.75 1.75 1.75 1.7	EA5 26.01 0.235 26.07 25.90 27.02 27.02 27.02 27.02 27.02 27.02 27.03 27	EA6 4.68 42.7.89 164.45 227.83 110.83 110.25 110.83 110.25 110.83 110.25 110.83	EA 7 14.55770.600.85770.600.87770.600.87770.600.87770.600.8577.8699.8889773.688.64275.54.53.54.53.54.53.57.3.68.64275.54.54.65.3.54.54.65.54.65.54.65.54.65.65.65.65.65.65.65.65.65.65.65.65.65.	EA8 0.622 0.79 0.884 1.20 1.63 1.163 1.63 1.63 1.63 1.63 1.63 1.6	EA9 22414155.852677.95566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.86959566.8695959595959595959595959595959595959595	EA10-% AXIAL ERROR 15.13 59.31 62.86 26.07 00 89.10 41.05 30.83 104.22 46.03 37.37 42.56 46.83 14.51 24.79 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 24.72 13.65 26.39 16.20 8.61 11.33 15.26 10.45 16.31 8.61 11.33 15.26 10.45 16.31 8.61 11.33 15.26 10.45 16.31 8.61 17.56 10.45 16.21 7.58 17.69 10.93 9.96 7.93 7.64 25.91 6.21 7.58 8.50 2.47 8.50 2.47 8.50 2.47 8.50 2.47 8.50 2.47 8.50 2.47

TABLE A6. -

(f) CALCULATED TANGENTIAL UNCERTAINTIES (PERCENT)

CP	BASELINE INLE ET1 ET2	T, STATION 1 ET3 ET4	ET5	ET6	ET7	ET8	ET9	ET10-X TANG	ERROR
1234567890112345678901233456789012334567890	0 12 0 03 0 02 0 15 0 003 0 17 0 003 0 19 0 003 0 11 0 003 0 0 003 0 0 11 0 003 0 0 0 0	0.04 0.03 0.05 0.03 0.05 0.03 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.04 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.03 0.05 0.05	0.04 0.05 0.05 0.05 0.05 0.05 0.05 0.05	0.033333333333333333333333333333333333	0.054 0.054 0.054 0.066 0.	• 066 • 005 • 006 •	04442445444223244233443323244443325444232122344444 0000000000000000000000000000000	0.28 0.394 0.372 0.3	
CP	BASELINE INLE	T, STATION 2 ET3 ET4	ET5	ET6	ET7	ET8	ET9	ET10-% TANG	ERROR
123456789011234567890112345678901123456789012345678901234567890	0.32	2.2 2.85 0.240 0.440 0.522 0.550	0.11 0.09 0.28 0.32 0.32 0.34 0.29 0.21 0.12 0.10 0.10 0.10 0.11 0.10 0.10	0.58830000000000000000000000000000000000	0.17 0.460 0.853 0.747 0.774 0.538 0.053 0.0	0.498 0.498 0.775 1.240 1.917 1.083 0.783	122888235360829431683404951171069488482255005274521211111111111110000000000000000000000	1.08 1.229 8.1229 15.635 15.635 16.775 16.795 12.201 11.211 16.358 16.573 17.220 11.211 16.358 16.358 16.358 17.35	

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СР	BASEL!	INE INL ET2	ET, STAT ET3	ION 3 ET4	ET5	ET6	ET7	ETS	ET9	ET10-X TANG ERROR
123456789012345678901234567890123456789012345678901234567890		00000000000000000000000000000000000000	0.92 0.92 282.43 0.00 0.00 0.00 0.00 0.00 0.00 1.30 1.32 1.32 1.12 0.86 0.51 0.66 0.74 0.67 0.77	2.91 42.763 21.731.400 0.000 0.000 3.259 0.000 3.259 0.000 0.000 3.259 0.0000 0.000	2.50 102.68 179.06 0.00 0.00 0.00 0.544 9.69 4.69 4.69 2.771 3.57 2.457 3.457 2.457 3.58 12.170 1.382 1.701 1.259 1.629 1	0.00 0.02 15.283 0.00 0.245 15.779 0.375 0	4.50 0.00	0.00 0.00 4.243.547 113.554.67 1.269.00 0.000 3.107 1.269.00 0.469.00 0.510.00 0.510.00 0.699.00 0.555.00 0.666.0	8.00 3.174 4.901 6.0000 6.00000 6.0000 6.0000 6.0000 6.0000 6.0000 6.00000 6.00000	0.00 0.20 1.21 0.51 1.51 0.51 0.00 0.00 1.91 3.47 4.96 0.00 0.05 0.05 0.05 0.05 0.05 0.05 0.0
CP	BASEL:	INE INL ET2	ET, STAT	ION 4 ET4	ET5	ET6	ET7	ETB	E19	ET10-2 TANG ERROR
123456789012345678901234567890123456789012345678901234567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00000444429590818575705554272320313540000000000000000000000000000000000	0.00 5.12 1.33 0.32 0.55 0.55 0.55 0.55 0.55 0.57 0.77 0.77 0.77 0.77 1.05 0.57 1.76 4.35 2.95 0.00	0.000 0.0000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.0	0.00 1.09 0.57 0.26 0.18 0.18 0.18 0.18 0.19 0.18 0.19 0.25 0.25 0.25 0.25 0.25 0.25 0.25 0.25	5.83 21.017 00.58 22.24 00.227 00.227 00.335 00.335 00.335 00.355	1.824 0.891 1.077 0.577 0.655 0.985 0.844 0.845 1.023 1.114 1.023 1.112 1.124	3.42 1.78 0.69 0.34 0.35 0.40 0.35 0.58 0.58 0.58 0.64 0.70 0.78 0.66 0.66 0.91 0.99 1.56 0.99 1.56 0.99 1.56 0.99 1.56 0.99 1.56 0.99 0.99 0.99 0.99 0.99 0.99 0.99 0.9	8.7357 1.7557 1.	20.53 16.87 83.63 48.81 48.81 45.04 48.07 17.17 23.87 22.70 40.71 23.87 22.70 40.71 23.95 22.70 40.71 33.04 40.51

СР	BASE ET1	LINE INL ET2	ET, STAT ET3	ION 5 ET4	ET5	ET6	ET7	ETB	ET9	ET10-2 TANG ERROR
12345678901234567890123456789012345678901234567890	0.00 0.00 0.00 2.35 1.20 1.30 1.20 1.30	0.00 0.00 1.57 1.12 1.14 1.03 1.03 1.03 1.03 1.24 1.32 1.12 1.32 1.12 1.32 1.12 1.32	0.00 0.00 4.75 2.31 1.64 1.50 1.20 1.20 1.21 1.20 1.21 1.20 1.20 1.2	0.00 0.00 4.75 4.99 2.79 2.79 2.1.42 1.51 2.20 1.51 1.52 1.20 1.51 1.52 1.53 1.55 1.55 1.55 1.55 1.55 1.55 1.55	0.00 12.997 4.43 7.57 2.81 2.24 11.55 11.32 11.3	0.00 12.364 7.943 5.455 21.652	0.00 0.20 1.00	5.15 2.580 2.13 1.89 1.646 1.461 1.461 1.461 1.461 1.461 1.401 1.4	21.54 107.79 13.86 10.48 110.45 10.45 110.45	0.00 213.81 0.00 0.00 41.34 73.43 57.54 61.80 22.35 88.11 36.82 34.84 20.57 45.78 11.41 25.87 11.41 25.87 18.97 22.95 26.48 19.29 18.97 22.95 25.56 11.57 11.57 9.73 10.40 10.88 8.01 18.71 148.79 0.00 0.00 0.00 0.00 0.00 0.00
CP	ET1	ELINE INI ET2	ET3	ET4	ET5	ET6	ET7	ET8	ET9	ET10-% TANG ERROR
12345678901234567890122345678901233456789012344567890	0.009 4.905 5.505 5.717 6.508 1.7.65	0.00 0.15 1.634 1.227 1.243 1.227 1.243 1.227 1.243 1.227 1.243 1.227 1.243 1.253 1.455 1.45	0.00 1.087 0.87 0.99 0.99 0.99 0.89 0.99 0.89 0.99 0.89 0.99 0.89 0.99 0.89 0.99 0.9	0.067 2.607 2.801 1.946 1.684 1.	0.5381 5.221.0968 1.822.5355 2.9355 1.822.535 1.822.535 1.822.535 1.823.699 1.82	0.030 5.945 5.54 3.651 42.097 42.096 42.651 42.097 42.096 43.480 44.480 44.480 44.480 44.480 44.480 44.480 44.480	0.03563 0.03563 1.432340 1.2321 1.2121 1.2076 1.2072 1.207	0.021 1.540 1.	09.00446228 4.4622437052437052436316396678 1.14.463153838 1.10.24370524363153836 1.10.24370524363153836 1.10.24370524363153836 1.10.24370524363153836 1.10.24370524363153836 1.10.24370524363153836 1.10.236363153836 1.10.2363636 1.10.2366 1.10.2366	0.00 9.27 7.86 6.78 15.65 6.95 19.50 5.17 5.66 11.25 10.55 8.09 10.61 4.16 7.02 10.56 3.17 2.5.98 10.61 4.16 7.02 3.89 5.54 3.17 2.5.98 1.56 1.56 1.32 2.89 5.54 1.56 1.50 0.00 0.00 0.00 0.00

CP	DASI ETI	ELINE INI ET2	LET, STAT	ION 7 ET4	ET5	ET6	ET7	ETB	ET9	ETID-X TAHG ERROR
12345678901234567890123456789012345678901234567890	0.00 0.00 4.34 5.57 6.79 1.15	0.87 0.87 1.32 1.82 1.32 1.81 0.81 0.81 0.77	0.00 0.00 3.92 2.24 2.10 2.10 2.10 2.10 2.10 2.10 2.10 2.10	0.00 0.72 5.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.36 6.37	0.00 0.00 3.44 5.75 2.65 2.65 1.50 1.77 1.47 1.32 1.32 1.32 1.32 1.32 1.32 1.32 1.32	0.00 0.00 4.83 5.83 5.22 2.26 1.85 1.70 1.33 1.35 1.35 1.37 1.31 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.17 1.08 1.08 1.08 1.09	0.00 0.00 6.45 3.05 3.29 3.25 4.05 2.38 2.38 2.38 2.38 2.38 2.38 2.38 2.38	0.00 0.01 13.16 10.11 10.11 4.35 4.45 4.52 2.45 4.45 4.28 2.13 2.23 2.13 2.23 2.13 2.23 2.13 2.23 2.2	0.00 0.03 4.49 4.78 3.49 4.78 3.49 22.22 2.85 11.58 11.76 12.37 12	0.00 0.00 7.70 5.94 6.03 5.85 7.59 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.177 6.187 6
СР	BASE!	ET2	ET, STATI	ON 8 ET4	ET5	ET6	ET7	ET8	ET9	ET10-2 TANG ERROR
12345678901234567890123456789012345678901234567890	25,7020000000000000000000000000000000000	794650444828159594945884330665926682284642213366592464438850369327659246444644645443334454423354375437543754375437543754375437543754	19.5722005 19.5222005 10.522	14.80 35.44 36.74 37.12 38.14 39.77 39.12 39	14 0 223 184 115 11 14 17 18 18 18 18 18 18 18 18 18 18 18 18 18	214.558211558582899677.10474488991111.2246426262333661755826269967.1111.22466263233366175532662632333665238266263233366523886265656652386626526526526526526526526526526526526526	14.651 9.651 10.653	10.4414 86.4498 66.555 57.224 57.227 10.465 10.4	12.79 8.86 8.85 5.82 5.82 6.73 5.82 6.73 6.73 7.73 7.73 7.73 7.73 7.73 7.73	4.53 4.54 4.54 4.54 4.54 5.29 2.20 2.33 2.20 2.33 2.33 2.33 2.33 2.33

	ENHAN	CED_INLE	T, STATI	ION 1						
CP 1 2	ET1 0.22 0.18	ET2 0.24 0.18	613 0.18 0.25	ET4 0.14 0.15	ET5 0.19 0.19	8.14 0.18	ET7 0.15 0.14	ET8 8.15 9.13	ET9 0.15 0.14	ET10-% TANG ERROR 0.65 0.62
3 4 5	0.24 0.32 0.38	0.17 0.23 0.21	0.17 0.18 0.16	0.20 0.17 0.17	0.20 0.13 0.24	0.18 0.20 0.20	0.17 0.22 0.18	0.17 0.10 0.13	0.23 0.20 0.19	0.21 0.34 0.49
67	0.14 0.17	0.19 0.19	0.16 0.19	0.18 0.19	0.18 0.17	0.17 0.20	0.17 0.15	0.15 0.12	0.18 0.22	0.30 0.44
10	0.15 0.21 0.18	0.20 0.15 0.22	0.19 0.24 0.21	0.19 0.16 0.22	0.19 0.17 0.21 0.22	0.19 0.23 0.25	0.18 0.15 0.15	0.13 0.17 0.14	0.20 0.16 0.16	0.47 0.45 0.40
11 12 13	0.33 0.12 6.28	0.17 0.20 0.17	0.20 0.19 0.17	0.19 0.24 0.17	0.21 0.21	0.17 0.20 0.21	0.20 0.20 0.21	0.13 0.15 0.15	0.23 0.16 0.19	6.26 6.31 0.37
14 15 16	0.12 0.20 0.12	0.21 0.19 0.22	0.26 0.24 0.18	0.18 0.21 0.18	0.22 0.23 0.18 0.20	0.22 0.19 0.18	0.18 0.18 0.19 0.20	0.15 0.13 0.15 0.17	0.16 0.17 0.17	0.46 0.73 9.67
17 18 19	0.21 0.17 0.16	0.19 0.19 0.20	0.27 0.20 0.22	0.22 0.22 0.20	0.23 0.22	0.27 0.20 0.22	0.21 0.17	0.15	0.22 0.18 0.19	0.61 0.68 0.29
20 21 22	0.37 0.20 0.21	0.25 0.20 0.17	0.22 0.22 0.21 0.19	0.18 0.23 0.19	0.22 0.26 0.19	0.24 0.21 0.18	0.23 0.16 0.18	0.15 0.17 0.14 0.17	8.20 0.19 0.17	0.41 0.43 0.36
23 24 25	0.14 0.24 0.22	0.19 0.20 0.16 0.20	0.23 0.19 0.21 0.24	0.16 0.14 0.26	0.21 0.22 0.21 0.19	0.21 0.20 0.21 0.22	0.22 0.15 0.15 0.23	0.14 0.16 0.12 0.16	0.18 0.17 0.17	0.36 0.37 0.41
26 27 28 29	0.25 8.31 0.16	D.18 0.19	0.19 0.22	0.11 0.15 0.18	0.22 0.19 0.21	0.17	0.19	0.14	0.15 0.19 0.14	0.29 0.65 0.38
30 31 32	0.24 0.28 0.21	0.22 0.17 0.16	0.15 0.16 0.16	0.17 0.15 0.18	0.19	0.19 0.18 0.17	0.21 0.16 0.16 0.18	0.13 0.15 0.13	0.15 0.17 0.14	0.57 0.31 0.29
33 34 35	0.11 0.17 0.16 0.23	0.17 0.19 0.18 0.15	0.17 0.17 0.14	0.21 0.19 0.16	0.20 0.23 0.17	0.17 0.18 0.24 0.19	0.17 0.15	0.14 0.12 0.12	0.19 0.18 0.14 0.13	0.46 0.55 0.40
36 37 38	0.23 0.19 0.18	0.19 0.16 0.18	0.16 0.20 0.20 0.13	0.15 0.14 0.15 0.12	0.22 0.25 0.19 0.26	0.19 0.19 0.18	0.16 0.20 0.11 0.14	0.09 0.13 0.16 0.11	0.12 0.14 0.16	0.38 0.35 0.28 0.46
39 40 41	0.17 0.23 0.13	0.15 0.24 0.20 0.25	0.14 0.12 0.17	0.18 0.17 0.21	0.17 0.18 0.20	0.17 0.16 0.19	0.14 0.16 0.16	0.13 0.11 0.12	0.16 0.14 0.15	0.25 0.21 0.11
42 43 44	0.20 0.13 0.18	0.25 0.20 0.18	0.16 0.11 0.18	0.12 0.14 0.15	0.20 0.22 0.18	0.17 0.19 0.14	0.12 0.16 0.10	0.08 0.12 0.12	0.10 0.15 0.17	0.55 0.21 0.26
45 46 47	0.15 0.17 0.18	0.19 0.19 0.16	0.12 0.19 0.14	0.14 0.19 0.16	0.16 0.12 0.15	0.15 0.15 0.19	0.16 0.16 0.15	0.09 0.10 0.11	0.14 0.10 0.13	0.61 0.23 0.26
48 49 50	0.15 0.20 0.20	0.16 0.15 0.12	0.12 0.21 0.15	0.16 0.13 0.14	0.19 0.18 0.13	0.13 0.19 0.15	0.12 0.12 0.07	0.13 0.11 0.09	0.16 0.16 0.09	0.45 0.40 0.64
СР	ENHAN ET1	ICED INLE ET2	T, STATI	ION 2 ET4	ET5	ET6	ET7	ET8	ET9	ET10-% TANG ERROR
1 2	0.24	0.27 0.31	0.53 0.26	1.09	1.34	1.37	1.41	1.78	1.68	4.20 4.94
3 4 5	0.44 0.78 0.53 0.60	0.43 0.58 0.70	0.05 0.15 0.26	0.19 0.57 0.59	0.27 0.67 0.80	0.44 0.39 0.72	0.69 0.11 0.63	1.47 0.26 0.64	1.60 1.77 0.05	4.97 9.41 5.50
6 7 8	0.74 0.48 0.48	0.78 0.73 0.70	0.44 0.46 0.45	0.84 0.80 1.05	1.24 1.23 0.92	1.11 1.32 1.06	0.99 1.42 1.25 1.12	0.85 1.15 1.12	1.42	2.68 3.74 6.12
9 10 11	0.49 0.35 0.49	0.72 0.61 0.59	0.63 0.32 0.48	0.91 1.08 0.91	1.17 1.28 1.10	1.15 1.22 1.08	1.12 1.15 1.06	1.24 1.43 1.29	1.94 1.28 1.74 1.56	3.89 3.77 3.98
12 13 14	0.45 0.22 0.37	0.45 0.46 0.34	0.59 0.57 0.46	0.90 1.01 1.06	1.20 0.94 0.74	1.05 0.93 0.77	0.93 0.92 0.80	0.99 1.14 1.09	1.33 1.40 1.31	8.16 4.48 9.67
15 16 17	0.13 0.09 0.05	0.39 0.31 0.13	0.40 0.40 0.26	1.01 0.57 0.87	0.65 0.65 0.60	0.59 0.66 0.61	0.55 0.67 0.62	0.96 0.89 0.76	1.61 1.35 0.98	5.86 5.41 3.86
18 19 20 21	0.05 0.05 0.06	0.23 9.22 0.26	0.27 0.24 0.14	0.65 0.61 0.47	0.45 0.44 0.38	0.52 0.44 0.39	0.60 0.43 0.40	0.60 0.50 0.42	1.18 1.37 0.70	3.34 4.74 6.21
22 23	0.12 0.11 0.15	0.24 0.25 0.22	0.12 0.08 0.05	0.44 0.33 0.24	0.29 0.26 0.12	0.30 0.27 0.16	0.31 0.28 0.21	0.44 9.38 0.25	0.53 0.66 0.29	2.57 5.36 3.01
25 26 27	0.20 0.18 0.19	0.21 0.26 0.31 0.24	0.05 0.07 0.10	0.18 0.15 0.09	0.07 0.08 0.11 0.17	0.09 0.06 0.06	0.12 0.05 6.01	0.23 0.89 0.06	0.30 0.19 0.07	2.65 3.08 2.12
28 29	0.22 0.24 0.27		0.14 0.14 0.22	0.10 0.17 0.17	0.19	0.10 0.15 0.16	0.03 0.10 0.16	0.02 0.09 0.16	0.10 0.21 0.34	2.39 4.59 1.99
30 31 32	0.27 0.29 0.33 0.20	0.35 0.35 0.31 0.24 0.37	0.18 0.23 0.22 0.39	0.22 0.31 0.22 0.33	0.28 0.34 0.27 0.39	0.28 0.31 0.24	0.28 0.27 0.20 0.33 0.32	0.19 0.29 0.21 0.50	0.32 0.37 0.30	3.90 2.77 2.24
33 34 35 36	0.35 0.39 0.41		0.32	0.46	0.41 0.53	0.36 0.55	U, 57	0.42 0.44	0.58 0.51	3.26 4.91 2.30
37 38 39	0.41 0.48 0.41 0.50	0.48 0.43 0.45 0.49	0.45 0.44 0.56	0.48 0.53 0.67	0.51 0.60 0.73 0.74 0.72	0.45 0.49 0.69	0.40 0.40 0.66	0.51 0.55 0.63	0.57 0.62 0.76	2.99 3.75 2.33 2.64
40 41 42	0.50 0.43 0.49	0.42 0.43 0.47	0.56 0.38 0.40 0.51 0.51	0.50 0.76 0.75	0.64	0.70 0.71 0.79	0.68 0.70 0.74 0.74	0.63 0.81 0.62	0.83 0.81 0.76	3.09 3.31
43 44 45	0.55 0.55 0.42 0.49	0.55 0.57 0.42	0.50 0.60 0.50	0.80 0.89 0.93 0.97	0.89 0.76 0.86 1.12	0.81 0.94 0.85 1.07	1.26 0.84 1.01	0.77 0.70 0.96 1.00	0.85 0.91 1.26 1.26	5.52 3.16 3.80 7.40
46 47 48	0.50 0.48	0.55 0.58 0.50	0.62 0.76 0.67	0.97 1.23 1.11	0.98 0.92	0.93 0.97 1.12	0.89 1.02	0.87 1.10 1.11	1.30 1.27 1.34	4.13 4.16 4.10
49 50	0.46 0.37 0.11	0.45	0.57	0.96 1.31	1.03 1.30 1.13	1.34	1.23 1.38 1.32	1.05	1.74	4.13 4.14

CP	ENHAN ET1	ICED INL ET2	ET, STAT ET3	ION 3 ET4	ET5	ET6	ET7	ET8	ET9	ET10-X TANG ERROR
123456789012345678901234567890123456789012345678901234567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	8.00 8.00 9.00	0.00 0.00	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0.000 0.000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	37. 35 0.00 0.00 0.00 0.00 0.50 1.33 0.00 0.00 0.00 0.00 0.00 0.00 0.0
CP	ENHAN ET1	CED INL	ET, STAT	ION 4 ET4	ETS	ET6	ET7	ET8	ET9	ET10-% TANG ERROR
12345678901234567890123456789012345678901234567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.00 0.00 23.25237 00.125537 00	0.00 0.00 0.00 57.92 85.20 0.23 3.22 2.24 1.33 1.27 1.16	0.00 0.00 236.90 20.99 7.75 3.70 1.956 1.067 0.69 0.62 0.74 0.74 0.74 0.77 0.89 1.064 0.62 0.70 0.74 0.74 0.74 0.77 0.90 0.00 0.00 0.00 0.00 0.00 0.00	0.00 102.45 8.95 8.95 1.11 1.18 1.70 1.39 1.18 1.43 1.19 1.18 1.43 1.19 1.127 1.39 1.16 1.20 1.20 1.24 1.32 1.27 1.44 0.00 0.00 0.00 0.00 0.00 0.00 0.00	6.28 1.40 2.840 2.840 2.840 2.771 2.460 2.777 2.777 2.	3.1249962011.1.34966611.1.34966611.1.34966611.1.34966611.1.34966611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466611.1.3466111.1.346611.1.346611.1.346611.1.346611.1.3466111.1.346611.1.346611.1.346611.1.3466111.1.3466111.1.3466111.1.3466111.1.3466111.	6.4957 1.957 0.699 0.791 0.889 0.887 0.889 0.887 0.889 1.100 1	15.625 15.798 64.105 11.3798 64.105 12.445 14.823 12.445 12.529 12.450 1	0.00 96.16 0.00 19.19 0.00 19.19 0.00 60.74 22.36 10.36 16.21 11.10 11.29 17.69 21.94 29.36 33.53 14.56 16.33 14.64 21.7.74 71.62 77.68 19.27 21.53 22.69 69.65 19.27 21.53 22.69 69.65 19.27 21.53 22.69 69.65 19.27 21.53 23.69 69.65 19.27 21.53 23.69 69.65 19.27 21.53 23.69 69.65 19.27 21.53 23.69 69.65 19.27 21.53 23.69 69.65 19.27 21.53 23.69 69.65 19.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0

CP ET1 ET2 ET3 ET4 ET5 ET6 ET7 ET8 ET9 ET10-X TANG ERROR

-	E11	2.12	L	617		2.0	E17	-10		2110 -	I AND ERROR
1234567890123456789012334567890123345678901234567890	0.00 0.00 0.00 41.162 48.98 53.416 70.89 134.480 123.70 1425.70 151.114 168.89 134.48 168.89 134.48 168.89 134.89 134.89 134.89 134.89 134.89 134.89 135.114 149.13 177.13 189.13	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	0.00 0.000 179.053 17.84 17.89 6.56 6.50 6.50 6.50 7.89 22.85 3.10 32.89 22.85 31.10 1.55 1.55 1.55 1.64 1.63 1.63 1.64 1.63 1.64 1.63 1.64 1.64 1.64 1.64 1.64 1.64 1.64 1.64	6.00 27.13 6.267 4.38 3.02 2.03 1.23 1.23 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.06 1.07 1.08	8.857.121 1.2.864.386.28.2 1.2.864.386.2 1.2.864.386.2 1.2.838	8.002 9.23 9.23 1.45 1.53 1.45 1.53	7.75538080808080808080808080808080808080808	288998907295536386441059577577191900117628866510000000000000000000000000000000000	848.7803900000000000000000000000000000000000	2.893.4944.293.8557.831.5633.801.5621.451.021.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.	
СР	ENHA ET1	NCED INL	ET, STAT	ION 6	ETS	ET6	ET7	ETB	ET9	ET10-7	TANG ERROR
123456789012345678901223456789012334567890123444444444444444444444444444444444444	0.000 0.000 0.000 471.154 1192.529.2253 1192	0.000 0.006 43.87 18.98 43.87 19.94 10.89 19.77 18.06 19.07 19.89 19.08	0.000 0.000 191.099 211.553 177.502 18.8877 3.6344 63.993 4.877 3.244 63.993 4.877 3.244 63.993 63.9	0.81 39.81 1149.95 59.64 35.79 23.63 12.97 6.98 2.90 2.90 2.55 2.55 2.55 2.68 2.77 2.68 2.77 2.68 2.77 2.68 2.77 2.77 2.77 2.77 2.77 2.77 2.77 2.7	0.080 14.98 17.43 4.67 4.66 43.05 22.09 11.53 11.22 11.22 11.22 11.22 11.22 11.22 11.23 11	0.00 12.35 13.20 5.70 3.07 2.71 3.06 12.23 11.84 12.63 11.66 12.69 11.69	0.06673 1.1.744	0.02 1.818 1.336 1.438 1.233 1.1.233 1	0.189 4.794 4.794 4.794 4.3.174 4.3.174 5.0.92 2.4.67 1.103 1.785		Opioni

CP	ENHA ET1	NCED INL	ET, STAT ET3	10N 7 ET4	ET5	ET4	ET7	ETO	ET9	ET10-X	TANG ERROR
12345678910112134 111213 11121 111213 11121 111213 11213 11213 11213 11213 11213 11213 11213 11213 11213 11213 11213 1	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	8.00 9.31 10.24 13.12 1.68 7.90 11.68 7.90 10.00 0.	0.00 0.00 21.89 25.40 0.00 4.07 2.95 2.40 1.75 1.75 1.75 1.75 1.75 1.54 1.31 1.31 1.31 1.31 1.31 1.31 1.31 1.3	8.00 9.00 23.564 23.564 27.578 9.727 2.327 2	0.00 0.00 12 1.23 50 1.43 23.178 21.190	8.00 8.00 22.29 17.02 16.01 7.75 6.18 3.27 1.38 1.18 1.28 1.28 1.19 1.00 1.00 1.00 1.00 1.00 1.00 0.94 1.00 0.85 0.80 0.96 0.85 0.97 1.08 0.97 0.00 0.85 0.97 0.00 0.97 0.00 0.97 0.00 0.0	8.80 8.80 64.69 32.83 32.22 37.43 11.42 12.33 11.42 11	0.00 0.00 4.98 2.51 2.148 2.55 1.58 1.40 1.40 1.40 1.121 1.130 1.121 1.130 1.1	0.008 4.249 5.573 6.544 5.573 6.544 6.	0.00 0.00	
СР	ENH/ ET1	NCED INL	ET, STAT	ION 8 ET4	ET5	ET6	ET7	ET8	ET9	ET10-X	TANG ERROR
123456789011234567890112322225312335567890112344567890	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	23.495 5.465 7.5646 7.5646 7.5646 7.5647 7.5	0.329 0.329 0.2148 0.1195 0.11	11.024 4.024 4.024 4.024 10.385 1	0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	1.56 1.61	0.75 0.78 0.81 0.76 0.77	1.27 1.39 1.26 1.40 1.00	3.39	2.11 4.41 2.22 2.06 3.00 4.53 1.78	

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	nesis in partial fulfillment of the r f Toledo, Toledo, Ohio. Respons			
efficiency. Laser Anemome unmodified baseline flow, as well upstream of the rotor. If flow condition are identified passage vortex due to the distions from a Navier-Stokes:	s were taken inside an isolated axeter measurements were made with and the second profile was distorted. A primary flow is defined in the relation of the inlet flow. A comparison between the two stortion of the inlet flow. A comparison between the two stortion of the inlet flow. A comparison profile has minimal effect on the	h two inlet velocity d by placing axisym otor and deviations flow deviations is n parison of experimen etween predicted and	profiles. One profile metric screens on the from this primary flow adde to assess the detail results with comit measured flows. A	e consisted of an the hub and shroud ow for each inlet evelopment of a putational predictional results
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